

information. The street address for the Docket Operations office (telephone: 800-647-5527) is in the **ADDRESSES** section.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2016-02-03 Airbus: Amendment 39-18382. Docket No. FAA-2015-1429; Directorate Identifier 2014-NM-246-AD.

(a) Effective Date

This AD becomes effective March 1, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to the airplanes identified in paragraphs (c)(1) and (c)(2) of this AD, certificated in any category, all manufacturer serial numbers.

(1) Airbus Model A319-113 and -114 airplanes.

(2) Airbus Model A320-211 and -212 airplanes.

(d) Subject

Air Transport Association (ATA) of America Code 71, Powerplant.

(e) Reason

This AD was prompted by a report that the aft mount pylon bolts of the CFM56-5 engines may have been installed using the wrong torque values. We are issuing this AD to detect and correct improper torque of the aft mount pylon bolts, which, if combined with any maintenance damage, could lead to aft engine mount failure, possibly resulting in engine detachment and consequent reduced control of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Inspection for Incorrect Torque Values

Within 6 months or 1,500 flight cycles, whichever occurs first after the effective date of this AD, inspect to determine the method used to install the engines, in accordance with the Accomplishment Instructions of

Airbus Service Bulletin A320-71-1063, including Appendix 01, dated August 13, 2014. A review of airplane maintenance records is acceptable in lieu of this inspection if the method used to install the engines can be conclusively determined from that review. For any engine replaced as specified in the Airbus A318/A319/A320/A321 Aircraft Maintenance Manual (AMM), Task 71-00-00-400-040-A01, "Installation of the Power Plant with Engine Positioner TWW 75E," dated May 2013: Within 6 months or 1,500 flight cycles, whichever occurs first after the effective date of this AD, re-torque the 4 aft mount pylon bolts using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA).

Note 1 to paragraph (g) of this AD:

Additional guidance for the re-torque can be found in Airbus A318/A319/A320/A321 AMM Task 71-00-00-400-040-A01, "Installation of the Power Plant with Engine Positioner TWW 75E," dated May 2014.

(h) Parts Installation Limitation

As of the effective date of this AD, no person may install a CFM56-5 engine, on any airplane, unless the inspection, and, as applicable, the re-torque, is done as specified in paragraph (g) of this AD.

(i) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Sanjay Ralhan, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone: 425-227-1405; fax: 425-227-1149. Information may be emailed to: 9-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) *Contacting the Manufacturer:* For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the EASA; or Airbus's EASA DOA. If approved by the DOA, the approval must include the DOA-authorized signature.

(j) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA AD 2014-0258, dated November 28, 2014, for related information. This MCAI may be

found in the AD docket on the Internet at <http://www.regulations.gov/#!documentDetail;D=FAA-2015-1429-0002>.

(2) Service information identified in this AD that is not incorporated by reference is available at the addresses specified in paragraphs (k)(3) and (k)(4) of this AD.

(k) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(i) Airbus Service Bulletin A320-71-1063, including Appendix 01, dated August 13, 2014.

(ii) Reserved.

(3) For service information identified in this AD, contact Airbus SAS, Airworthiness Office—EAW, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone: +33 5 61 93 36 96; fax: +33 5 61 93 44 51; email: account.airworth-eas@airbus.com; Internet <http://www.airbus.com>.

(4) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on January 11, 2016.

Michael Kaszycki,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016-01108 Filed 1-25-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-1991; Directorate Identifier 2014-NM-251-AD; Amendment 39-18381; AD 2016-02-02]

RIN 2120-AA64

Airworthiness Directives; Airbus Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for all Airbus Model A318-111 and -112 airplanes; Model A319-111, -112, and -115 airplanes; Model A320-214

airplanes; and Model A321-111, -112, -211, -212, and -213 airplanes. This AD was prompted by reports of cracked cadmium-plated lock nuts that attach the hinge to the fan cowl door. This AD requires inspecting to determine the serial number of each engine fan cowl door, inspecting for cracking of the hinge lock nuts of any affected door, and replacing the lock nuts if necessary. We are issuing this AD to detect and correct cracking of the hinge lock nuts, which could result in separation of the hinge from the fan cowl door, in-flight loss of the door, and consequent damage to the airplane.

DATES: This AD becomes effective March 1, 2016.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of March 1, 2016.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov/#!docketDetail;D=FAA-2015-1991>; or in person at the Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC.

For service information identified in this final rule, contact the following:

For Airbus service information contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone: +33 5 61 93 36 96; fax: +33 5 61 93 44 51; email: account.airworth-eas@airbus.com; Internet <http://www.airbus.com>.

For Goodrich service information contact Goodrich Aerostructures, 850 Lagoon Drive, Chula Vista, California, 91910-2098; telephone: 619-691-2719; email: jan.lewis@goodrich.com; Internet: <http://www.goodrich.com/TechPubs>.

You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-1991.

FOR FURTHER INFORMATION CONTACT: Sanjay Ralhan, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-1405; fax 425-227-1149.

SUPPLEMENTARY INFORMATION

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to all Airbus Model A318-111 and -112 airplanes; Model A319-111, -112, and -115 airplanes; Model A320-214 airplanes; and Model A321-111, -112, -211, -212, and -213 airplanes. The NPRM published in the **Federal Register** on July 2, 2015 (80 FR 38036).

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2014-0276, dated December 19, 2014 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition for all Airbus Model A318-111 and -112 airplanes; Model A319-111, -112, and -115 airplanes; Model A320-214 airplanes; and Model A321-111, -112, -211, -212, and -213 airplanes. The MCAI states:

In-service findings have been reported of cracked cadmium plated lock nuts. This cracking occurs shortly after installation. Investigation results attribute the cause to an improper manufacturing procedure of the nuts. It was determined that the affected batch of lock nuts was used on the fan cowl to attach hinges to the cowl doors on CFM56-5B engines only.

This condition, if not corrected, could lead to separation of the hinge from the fan cowl door, possibly resulting in in-flight loss of a fan cowl door, with consequent damage to the aeroplane and/or injury to persons on the ground.

For the reasons describes above, this [EASA] AD required identification of the affected fan cowl doors, a one-time inspection of the fan cowl door hinge nuts and, depending on findings, replacement of the affected nuts.

You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov/#!documentDetail;D=FAA-2015-1991-0003>.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM (80 FR 38036, July 2, 2015) or on the determination of the cost to the public.

Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting this AD as proposed, except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM (80 FR 38036, July 2, 2015) for correcting the unsafe condition; and

- Do not add any additional burden upon the public than was already proposed in the NPRM (80 FR 38036, July 2, 2015).

Related Service Information under 1 CFR part 51

Airbus has issued Service Bulletin A320-71-1062, dated July 28, 2014. Goodrich Aerostructures has issued Service Bulletin RA32071-151, dated June 11, 2014. The service information describes procedures for inspection and replacement of the hinge nuts of the fan cowl door. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

Costs of Compliance

We estimate that this AD affects 437 airplanes of U.S. registry.

We also estimate that it takes about 2 work-hours per product to comply with the basic requirements of this AD. The average labor rate is \$85 per work-hour. Required parts will cost about \$0 per product. Based on these figures, we estimate the cost of this AD on U.S. operators to be \$74,290, or \$170 per product.

We have received no definitive data that would enable us to provide a cost estimate for the on-condition actions specified in this AD.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between

the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov/#!docketDetail;D=FAA-2015-1991>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone 800-647-5527) is in the **ADDRESSES** section.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2016-02-02 Airbus: Amendment 39-18381. Docket No. FAA-2015-1991; Directorate Identifier 2014-NM-251-AD.

(a) Effective Date

This AD becomes effective March 1, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to the airplanes, certificated in any category, identified in paragraphs (c)(1), (c)(2), (c)(3), and (c)(4) of this AD, all manufacturer serial numbers.

(1) Airbus Model A318-111 and -112 airplanes.

(2) Airbus Model A319-111, -112, and -115 airplanes.

(3) Airbus Model A320-214 airplanes.

(4) Airbus Model A321-111, -112, -211, -212, and -213 airplanes.

(d) Subject

Air Transport Association (ATA) of America Code 71, Powerplant.

(e) Reason

This AD was prompted by reports of cracked cadmium-plated lock nuts that attach the hinge to the fan cowl door. We are issuing this AD to detect and correct cracking of the hinge lock nuts, which could result in separation of the hinge from the fan cowl door, the in-flight loss of the door, and consequent damage to the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Inspect to Determine Serial Number

Within 24 months after the effective date of this AD: Inspect to determine if any fan cowl door has a serial number 10029001 through 11092003 inclusive, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-71-1062, dated July 28, 2014; or Goodrich Aerostructures Service Bulletin RA32071-151, dated June 11, 2014. A review of airplane maintenance records is acceptable in lieu of the inspection required by this paragraph, provided those records can be relied upon for that purpose and the serial number can be positively identified by that review.

(h) Inspection and Replacement

For any fan cowl door having any serial number identified in paragraph (g) of this AD: Within 24 months after the effective date of this AD, do a detailed inspection for cracking of the hinge lock nuts of the door, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-71-1062, dated July 28, 2014; or Goodrich Aerostructures Service Bulletin RA32071-151, dated June 11, 2014. If any crack is found, before further flight, replace each cracked hinge lock nut, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-71-1062, dated July 28, 2014; or Goodrich Aerostructures Service Bulletin RA32071-151, dated June 11, 2014.

(i) Special Flight Permits

Special flight permits, as described in Section 21.197 and Section 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199), are not allowed.

(j) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) **Alternative Methods of Compliance (AMOCs):** The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19.

In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Sanjay Ralhan, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone: 425-227-1405; fax: 425-227-1149. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) **Contacting the Manufacturer:** For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). If approved by the DOA, the approval must include the DOA-authorized signature.

(k) Related Information

Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA Airworthiness Directive 2014-0276, dated December 19, 2014, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov/#!documentDetail;D=FAA-2015-1991-0003>.

(l) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(i) Airbus Service Bulletin A320-71-1062, dated July 28, 2014.

(ii) Goodrich Aerostructures Service Bulletin RA32071-151, dated June 11, 2014.

(3) For Airbus service information identified in this AD, contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone: +33 5 61 93 36 96; fax: +33 5 61 93 44 51; email: account.airworth-eas@airbus.com; Internet <http://www.airbus.com>.

(4) For Goodrich service information identified in this AD, contact Goodrich Aerostructures, 850 Lagoon Drive, Chula Vista, California, 91910-2098; telephone: 619-691-2719; email: jan.lewis@goodrich.com; Internet: <http://www.goodrich.com/TechPubs>.

(5) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(6) You may view this service information that is incorporated by reference at the

National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on January 9, 2016.

Michael Kaszycki,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016-00952 Filed 1-25-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-2983; Directorate Identifier 2015-NE-20-AD; Amendment 39-18383; AD 2016-02-04]

RIN 2120-AA64

Airworthiness Directives; CFM International S.A. Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain CFM International S.A. (CFM) CFM56-5B series turbofan engines. This AD was prompted by a corrected lifing analysis by the engine manufacturer that shows the need to identify an initial and repetitive inspection threshold for certain part number (P/N) turbine rear frames (TRFs). This AD requires initial and repetitive inspections of certain P/N TRFs on the low-pressure turbine (LPT) frame assembly. We are issuing this AD to prevent failure of the TRF on the LPT frame assembly, which could lead to engine separation, damage to the engine, and damage to the airplane.

DATES: This AD is effective March 1, 2016.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of March 1, 2016.

ADDRESSES: For service information identified in this AD, contact CFM International Inc., Aviation Operations Center, 1 Neumann Way, M/D Room 285, Cincinnati, OH 45125; phone: 877-432-3272; fax: 877-432-3329; email: aviation.fleetsupport@ge.com. You may view this service information at the FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125. It is also available on the Internet at <http://www.regulations.gov> by

searching for and locating Docket No. FAA-2015-2983.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-2983; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Kyle Gustafson, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781-238-7183; fax: 781-238-7199; email: kyle.gustafson@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to certain CFM CFM56-5B series turbofan engines. The NPRM published in the **Federal Register** on October 2, 2015 (80 FR 59672). The NPRM was prompted by a corrected lifing analysis by the engine manufacturer that shows the need to identify an initial and repetitive inspection threshold for certain P/N TRFs. The NPRM proposed to require initial and repetitive inspections of certain P/N TRFs on the LPT frame assembly. We are issuing this AD to correct the unsafe condition on these products.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM (80 FR 59672, October 2, 2015) or on the determination of the cost to the public.

Clarification to the Repetitive Inspection Requirements

We have revised the Compliance, paragraph (e) of this AD, to clarify the repetitive inspection requirements for when the initial inspection is done prior to the initial inspection threshold.

Conclusion

We reviewed the relevant data and determined that air safety and the public interest require adopting this AD

as proposed except for the changes described above. We have determined that the changes described above are minor changes, as they:

- Are consistent with the intent that was proposed in the NPRM (80 FR 59672, October 2, 2015) for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM (80 FR 59672, October 2, 2015).

We also determined that these changes will not increase the economic burden on any operator or increase the scope of this AD.

Related Service Information Under 14 CFR Part 51

We reviewed CFM Service Bulletin (SB) No. CFM56-5B S/B 72-0850, dated December 19, 2012, which describes procedures for inspecting the TRF. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section of this final rule.

Other Related Service Information

We also reviewed CFM SB No. CFM56-5B S/B 72-0308. Operators subject to this AD are required to follow different initial and repetitive inspection intervals depending on whether CFM SB No. CFM56-5B S/B 72-0308 has been applied.

Costs of Compliance

We estimate that this AD affects about 94 engines installed on airplanes of U.S. registry. We also estimate that it will take about 3 hours per engine to do the inspection. The average labor rate is \$85 per hour. Based on these figures, we estimate the cost of this AD on U.S. operators to be \$23,970.

Authority for this Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority