

the SB specifically list the fleets affected, and which steps are applicable based on several different configurations of various aircraft. Therefore, the commenter concluded there is no need to include the word “applicable” to exclude those products for which the requirements clearly do not apply.

ARSA commented that under part 39 the FAA cannot make anything “applicable” that is not directly related to the unsafe condition and specified actions must be limited to those that directly address the unsafe condition. In its view, the FAA’s interpretation mandates accomplishing all actions, whether or not necessary to correcting the unsafe condition, which is contrary to part 39.

FAA’s Response to Comments

The FAA intends “applicable” to have the same meaning in both places in paragraph (f) of *AD 2007–07–02*. The first usage limits the required actions to those that apply to a particular aircraft under the specific conditions found; it does not permit an operator to decide which actions are necessary and which are unnecessary to correct the unsafe condition.

The second usage references Table 1 in the AD that identifies the model of aircraft to which each service bulletin applies. The “applicable service bulletin” means the service bulletin that applies to each corresponding aircraft model, as indicated in Table 1 of the AD. Similarly, “all the applicable actions” specified in each applicable service bulletin are those actions that are identified as applying to a particular aircraft. “Applicable” is a necessary qualifier in this context for two reasons: (1) In many ADs, the referenced service bulletins specify different actions for different aircraft configurations, typically identified as “Group 1,” “Group 2,” etc.; (2) in many ADs, the referenced service bulletins specify different actions depending on conditions found during performance of previous steps in the instructions (e.g., if a crack is smaller than a specified size, repair in accordance with the Structural Repair Manual; if larger, repair in accordance with a method approved by the Aircraft Certification Office). The term “applicable” limits the AD’s requirements to only those that are specified in the service bulletin for the configuration and conditions of a particular aircraft. In this case, the word “all” means that every applicable action must be accomplished.

Although this response applies specifically to AD 2007–07–02, this general principle also applies to uses of the term “applicable” in other ADs. The

FAA promulgates ADs with specific standards to directly address the identified unsafe condition. As exemplified by AD 2007–07–02, ADs often require many different actions for various models and aircraft configurations. Because of those complexities, mandating AD actions without incorporating by reference the manufacturer’s service bulletin that may contain “normal” part 43 maintenance actions becomes impracticable or may interject unnecessary complexities or inconsistencies that adversely affect performance of the necessary corrective actions.

Question 4: What is the extent of an aircraft operator’s responsibilities when an AD requires an action that cannot be accomplished on a particular aircraft?

Answer: Sections 39.15 and 39.17 require ADs to apply to a specific product, even if the product has been changed through component removal or replacement or other modification. An operator who cannot comply with the specific requirements of an AD must request approval of an AMOC from the FAA. The operator must obtain an AMOC approval even if the affected component has been removed from the aircraft, rendering compliance with the specific requirements of the AD impossible. The AMOC process allows the FAA to determine whether the unsafe condition has been eliminated when an operator removes a component addressed in an AD and replaces it with a different component.

Summary of Comments

Some commenters stated the FAA’s interpretation is either wrong because when the AD pertains to a specific part or component that has since been legally removed or pertains to a part or such that is not installed on the aircraft, the AD no longer applies, or represents a change from past practice or guidance.

FAA Response to Comments

If a change to a product makes it impossible to comply with the requirements of an AD, then the operator must request an AMOC approval from the FAA. Sections 39.15 and 39.17 directly answer this issue. Section 39.15 provides that an AD applies to each product identified in the AD, even if an individual product has been changed by modifying, altering, or repairing it in the area addressed by the AD. Section 39.17 requires that if a change in a product affects an operator’s ability to accomplish the actions required by the AD in any way, the operator must request FAA approval of an AMOC. Together these sections

require an operator who cannot comply with the specific requirements of an AD to request FAA approval of an AMOC. The operator must obtain an AMOC approval even if the affected product has been removed from the aircraft, rendering compliance with the specific requirements of the AD impossible. The AMOC process allows the FAA to determine whether the unsafe condition has been eliminated when an operator removes a component to which an AD applies and replaces it with a different component.

This approach was clearly specified in the FAA’s part 39 rulemaking in 2002. See *Airworthiness Directives*, 67 FR 47998 (“Specifically, FAA is adding to part 39 the language explaining that ADs apply even if products have been modified, altered, or repaired in the area addressed by the directive.”). The 2002 rulemaking did not introduce any new regulatory requirements; rather, the FAA simply codified in part 39 provisions currently found in ADs. *Id.* at 47999. If a change in a product affects one’s ability to comply with the AD, the person operating the aircraft or using the product must ask the FAA’s permission to use an AMOC, and the request must either show that the change eliminated the unsafe condition or include the specific actions proposed. *Id.* at 48000.

This response was coordinated with the Aircraft Maintenance Division of the Flight Standards Service and the Design, Manufacturing and Airworthiness Division of the Aircraft Certification Service.

Issued in Washington, DC, on April 19, 2016.

Lorelei Peter,

Assistant Chief Counsel for Regulations.

[FR Doc. 2016–09667 Filed 4–26–16; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. **FAA–2015–4474**; Directorate Identifier **2015–NE–34–AD**; Amendment **39–18485**; AD **2016–08–09**]

RIN 2120–AA64

Airworthiness Directives; Pratt & Whitney Division Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain

Pratt & Whitney Division (PW) PW4000–94 inch and PW4000–100 inch model turbofan engines. This AD was prompted by a report of a crack found in the high-pressure compressor (HPC) 10th stage disk. This AD requires performing an ultrasonic inspection (USI) or an eddy current inspection (ECI) of the HPC 10th stage disk. We are issuing this AD to prevent failure of the HPC 10th stage disk, uncontained disk release, damage to the engine, and damage to the airplane.

DATES: This AD is effective June 1, 2016.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of June 1, 2016.

ADDRESSES: For service information identified in this final rule, contact Pratt & Whitney Division, 400 Main St., East Hartford, CT 06108; phone: 860 565–8770; fax: 860 565–4503. You may view this service information at the FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call (781) 238–7125. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2015–4474.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2015–4474; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800–647–5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Katheryn Malatek, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781–238–7747; fax: 781–238–7199; email: katheryn.malatek@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to certain PW PW4000–94 inch turbofan engines with HPC 10th stage disk, part number (P/N) 51H710 or

53H976–06, installed and certain PW4000–100 inch turbofan engines with HPC 10th stage disk, P/N 53H976–06, installed. The NPRM published in the **Federal Register** on December 9, 2015 (80 FR 76400). The NPRM was prompted by a report of a crack found in the HPC 10th stage disk. The root cause of the crack was a manual polishing procedure, previously used during manufacture, that caused surface scratches on the disk. The NPRM proposed to require a USI or ECI of the HPC 10th stage disk. We are issuing this AD to prevent failure of the HPC 10th stage disk, which could lead to an uncontained disk release, damage to the engine, and damage to the airplane.

Comments

We gave the public the opportunity to participate in developing this AD. The following presents the comments received on the NPRM (80 FR 76400, December 9, 2015) and the FAA's response to each comment.

Support for the NPRM

The Boeing Company and United Airlines expressed support for the NPRM (80 FR 76400, December 9, 2015).

Request to Use ECI as Follow-on to USI

American Airlines requested that we revise Compliance paragraph (e) of this AD to add a statement that the ECI can be used to confirm the presence of a crack if a USI is initially performed and the ECI is the final authority on whether or not a crack is present on the disk.

We agree. We revised Compliance paragraph (e) of this AD to allow a follow-on ECI.

Request to Allow Disk Replacement Repairs

Atlas Air requested that we revise the Compliance paragraph (e) of this AD to allow use of disk replacement repairs per the PW PW4000–94/100 Clean, Inspect, Repair (CIR) Manual Part No. 51A357, Section 72–35–10, Repair 07.

We disagree. This AD requires removal of the 10th stage disk if it fails inspection. There are no FAA-approved repairs allowed on the 10th stage disk. The previously approved PW4000–94/100 CIR Manual Part No. 51A357, Section 72–35–07, Repair 04 to the drum rotor, replaces the disk, resulting in a part eligible for installation. We did not change this AD.

Request to Allow ECI at Overhaul

Air India Limited requested that Compliance paragraph (e) of this AD allow an ECI when the HPC is “overhauled” rather than when it is “removed from the engine.” Air India

Limited indicated that “overhauled” is clearer than “removed from the engine”.

We disagree. The intent of this AD is to inspect the 10th stage disk at exposure. The phrase, “Whenever the HPC front drum rotor is removed from the engine . . .” clearly describes the appropriate level of exposure for performing the ECI. We did not change this AD.

Request To Waive Repeat USI

Air India Limited requested that we revise Compliance paragraph (e) of this AD to indicate that a repeat USI should be waived to reduce the maintenance burden if the low-pressure turbine (LPT) is removed in less than 100 hours since the last USI.

We disagree. Our safety risk assessment assumed that a USI is performed whenever the high-pressure turbine (HPT) or LPT is removed from the engine and an ECI is performed whenever the HPC front drum rotor disk assembly is removed from the engine. We determined the inspection interval in the Compliance paragraph (e) of this AD provides an acceptable level of safety. We did not change this AD.

Request To Remove Compliance Statement

FedEx requested that we revise Compliance paragraph (e) of this AD to remove the statement, “Comply with this AD within the compliance times specified, unless already done.” FedEx stated that there are no compliance times specified and the compliance requires a repetitive inspection, so the statement does not apply.

We disagree. The statement “. . . unless already done” allows an operator who has performed an initial inspection before the effective date of the AD, but has not yet returned the part to service, to take credit for that action. While there is no calendar or cyclic time given, the requirements of this AD must be met when the HPT, LPT, or HPC front drum rotor disk assembly is removed from the engine. We did not change this AD.

Conclusion

We reviewed the relevant data, considered the comments received, and determined that air safety and the public interest require adopting this AD with the changes described previously. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM (80 FR 76400, December 9, 2015) for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already

proposed in the NPRM (80 FR 76400, December 9, 2015).

We also determined that these changes will not increase the economic burden on any operator or increase the scope of this AD.

Related Service Information Under 1 CFR Part 51

We reviewed PW Alert Service Bulletin (ASB) PW4G-100-A72-255, dated August 31, 2015 and PW ASB PW4ENG A72-833, dated August 20, 2015. The ASBs provide lists of affected HPC disks and describe procedures for USI and ECI of the HPC 10th stage disk. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section.

Costs of Compliance

We estimate that this AD affects 763 engines installed on airplanes of U.S. registry. We also estimate that it would take about 12 hours per engine to do the inspection. The average labor rate is \$85 per hour. Based on these figures, we estimate the cost of this AD on U.S. operators to be \$778,260.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2016-08-09 Pratt & Whitney Division:

Amendment 39-18485; FAA-2015-4474; Directorate Identifier 2015-NE-34-AD.

(a) Effective Date

This AD is effective June 1, 2016.

(b) Affected ADs

None.

(c) Applicability

(1) This AD applies to all Pratt & Whitney Division (PW) PW4050, PW4052, PW4056, PW4060, PW4060A, PW4060C, PW4062, PW4062A, PW4152, PW4156, PW4156A, PW4158, PW4160, PW4460, PW4462, and PW4650 turbofan engines, including models with a "-3" suffix, with one of the following installed:

(i) High-pressure compressor (HPC) 10th stage disk, part number (P/N) 51H710, with a serial number (S/N) listed in Table 1 of PW Alert Service Bulletin (ASB) PW4ENG A72-833, dated August 20, 2015; or

(ii) HPC 10th stage disk, P/N 53H976-06, with an S/N listed in Table 2 of PW ASB PW4ENG A72-833, dated August 20, 2015.

(2) This AD also applies to all PW PW4164, PW4168, PW4168A, PW4164C, PW4164C/B, PW4170, PW4168A-1D, PW4168-1D, PW4164-1D, PW4164C-1D, and PW4164C/B-1D turbofan engines with an HPC 10th stage disk, P/N 53H976-06, with an S/N listed Table 1 of PW ASB PW4G-100-A72-255, dated August 31, 2015, installed.

(d) Unsafe Condition

This AD was prompted by a report of a crack found in the HPC 10th stage disk. We are issuing this AD to prevent failure of the HPC 10th stage disk, uncontained disk release, damage to the engine, and damage to the airplane.

(e) Compliance

Comply with this AD within the compliance times specified, unless already done.

(1) After the effective date of this AD, whenever the high-pressure turbine (HPT) or low-pressure turbine (LPT) is removed from the engine, perform an ultrasonic inspection (USI) of the HPC 10th stage disk for cracks. If the HPC 10th stage disk fails the USI, perform a follow-on eddy current inspection (ECI) or remove the disk from service and replace with a part eligible for installation.

(2) After the effective date of this AD, whenever the HPC front drum rotor disk assembly is removed from the engine, perform an ECI of the HPC 10th stage disk for cracks. Remove from service any HPC 10th stage disk that fails inspection and replace with a part eligible for installation. A USI as required by paragraph (e)(1) of this AD is not required if an ECI is performed.

(f) Alternative Methods of Compliance (AMOCs)

The Manager, Engine Certification Office, FAA, may approve AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to make your request. You may email your request to: ANE-AD-AMOC@faa.gov.

(g) Related Information

For more information about this AD, contact Katheryn Malatek, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781-238-7747; fax: 781-238-7199; email: katheryn.malatek@faa.gov.

(h) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Pratt & Whitney (PW) Alert Service Bulletin (ASB) PW4G-100-A72-255, dated August 31, 2015.

(ii) PW ASB PW4ENG A72-833, dated August 20, 2015.

(3) For PW service information identified in this AD, contact Pratt & Whitney Division, 400 Main St., East Hartford, CT 06108; phone: 860-565-8770; fax: 860-565-4503.

(4) You may view this service information at FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call

202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Burlington, Massachusetts, on April 7, 2016.

Colleen M. D'Alessandro,

Manager, Engine & Propeller Directorate, Aircraft Certification Service.

[FR Doc. 2016-09687 Filed 4-26-16; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-4344; Directorate Identifier 2015-NE-32-AD; Amendment 39-18486; AD 2016-08-10]

RIN 2120-AA64

Airworthiness Directives; General Electric Company Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for all General Electric Company (GE) CF6-80C2A1, CF6-80C2A2, CF6-80C2A3, CF6-80C2A5, CF6-80C2A5F, CF6-80C2A8, CF6-80C2B1, CF6-80C2B1F, CF6-80C2B1F1, CF6-80C2B1F2, CF6-80C2B2, CF6-80C2B2F, CF6-80C2B3F, CF6-80C2B4, CF6-80C2B4F, CF6-80C2B5F, CF6-80C2B6, CF6-80C2B6F, CF6-80C2B6FA, CF6-80C2B7F, CF6-80C2B8F, CF6-80C2D1F, CF6-80C2L1F, CF6-80C2K1F and CF6-80E1A1, CF6-80E1A2, CF6-80E1A3, CF6-80E1A4, and CF6-80E1A4/B turbofan engines. This AD was prompted by reports of a burn-through of the accessory heat shield during an engine fire, propagating the fire into the accessory compartment and igniting additional flammable fuel source. This AD requires replacing the accessory heat shield assembly. We are issuing this AD to prevent fires from propagating into the accessory compartment, resulting in an uncontrolled engine fire, and damage to the airplane.

DATES: This AD is effective June 1, 2016.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of June 1, 2016.

ADDRESSES: For service information identified in this final rule, contact General Electric Company, GE Aviation, Room 285, 1 Neumann Way, Cincinnati, OH 45215; phone: 513-552-3272; email: aviation.fleetsupport@ge.com. You may view this service information at the

FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-4344.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-4344; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT:

Herman Mak, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781-238-7147; fax: 781-238-7199; email: herman.mak@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to GE CF6-80C2A1, CF6-80C2A2, CF6-80C2A3, CF6-80C2A5, CF6-80C2A5F, CF6-80C2A8, CF6-80C2B1, CF6-80C2B1F, CF6-80C2B1F1, CF6-80C2B1F2, CF6-80C2B2, CF6-80C2B2F, CF6-80C2B3F, CF6-80C2B4, CF6-80C2B4F, CF6-80C2B5F, CF6-80C2B6, CF6-80C2B6F, CF6-80C2B6FA, CF6-80C2B7F, CF6-80C2B8F, CF6-80C2D1F, CF6-80C2L1F, and CF6-80C2K1F turbofan engines. This AD that would also apply to CF6-80E1A1, CF6-80E1A2, CF6-80E1A3, CF6-80E1A4, and CF6-80E1A4/B turbofan engines. The NPRM published in the **Federal Register** on December 7, 2015 (80 FR 75952). The NPRM was prompted by reports of a burn-through of the accessory heat shield during an engine fire leading to an accessory compartment fire. A fire burns through the accessory heat shield and ignites the integrated drive generator (IDG) and main fuel pump, which supports further combustion. The existing accessory heat shield assembly leaves a large area above the sensitive accessories, such as

the IDG and the main fuel pump, without adequate protection. A total of three burn-through events have occurred. The NPRM proposed to require replacing the accessory heat shield assembly. We are issuing this AD to prevent an uncontrolled engine fire, and damage to the airplane.

Comments

We gave the public the opportunity to participate in developing this AD. The following presents the comments received on the NPRM (80 FR 75952, December 7, 2015) and the FAA's response to each comment.

Support for the NPRM (80 FR 75952, December 7, 2015)

The Boeing Company and the National Transportation Safety Board expressed support for the NPRM (80 FR 75952, December 7, 2015).

Revision to Service Information

We revised the Discussion section and Applicability paragraph (e) of this AD to include all the GE CF6-80C2 and CF6-80E1 turbofan engine models.

Request To Change Summary

GE requested that we revise the Summary paragraph of this AD to correct the number of events and clarify the event description.

We agree. Only three of the originally specified five events resulted in heat shield burn-throughs. We revised the Summary paragraph of this AD to correct the number of events and clarify the event description.

Request To Revise the Other Related Service Information Paragraph

GE requested that we revise the Other Related Service Information paragraph of this AD to remove GE Service Bulletin (SB) CF6-80C2 S/B 72-1523, dated September 22, 2015. This SB only applies to the military variant of the engine.

We disagree. The military variant of the engine is also certified by the FAA. We did not change this AD.

Request To Revise the Costs of Compliance

GE, KLM Royal Dutch Airlines (KLM), All Nippon Airways (ANA), and Federal Express (FedEx) requested that we revise the Costs of Compliance paragraph of this AD to correct the parts cost used in the calculations.

We agree. We considered the costs of all the parts needed to comply with this AD and revised the costs per engine to \$14,207 and the total cost to U.S. operators to \$13,680,920.