

§ 39.13 [Amended]

2. Section 39.13 is amended by removing amendment AD 39–10851 (63 FR 56542, October 22, 1998), and by adding a new airworthiness directive (AD), to read as follows:

Airbus: Docket 2002–NM–177–AD.

Supersedes AD 98–22–05, Amendment 39–10851.

Applicability: Model A320 series airplanes, certificated in any category, on which Airbus Modification 22418 (reference Airbus Service Bulletin A320–57–1043) has not been done.

Compliance: Required as indicated, unless accomplished previously.

To prevent fatigue cracking of the lower surface panel on the wing center box, which could result in reduced structural integrity of the airplane, accomplish the following:

Restatement of Requirements of AD 98–22–05

Repetitive Inspections

(a) Except as provided by paragraph (e) of this AD: Prior to the accumulation of 20,000 total flight cycles, or within 60 days after November 27, 1998 (the effective date of AD 98–22–05, amendment 39–10851), whichever occurs later, perform a high frequency eddy current inspection to detect fatigue cracking of the lower surface panel on the wing center box, in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997, or Revision 03, dated April 30, 2002. Repeat the eddy current inspection thereafter at intervals not to exceed 7,500 flight cycles until the actions required by paragraph (c) of this AD are accomplished.

Repair

(b) Except as provided by paragraph (d) of this AD: If any cracking is detected during any inspection required by paragraph (a) of this AD, prior to further flight, repair in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997, or Revision 03, dated April 30, 2002. Accomplishment of the repair constitutes terminating action for the repetitive inspections for the repaired area only.

Inspection/Modification/Repair

(c) Prior to the accumulation of 25,000 total flight cycles, or within 60 days after November 27, 1998, whichever occurs later: Perform a high frequency eddy current inspection to detect fatigue cracking of the lower surface panel on the wing center box, in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997, or Revision 03, dated April 30, 2002.

(1) If no cracking is detected: Prior to further flight, modify the lower surface panel on the wing center box, in accordance with Airbus Service Bulletin A320–57–1043, Revision 02, dated May 14, 1997, or Revision 05, dated April 30, 2002. Accomplishment of the modification constitutes terminating action for the requirements of paragraph (a) of this AD.

(2) Except as provided by paragraph (d) of this AD, if any cracking is detected: Prior to

further flight, repair in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997, or Revision 03, dated April 30, 2002; and modify any uncracked area in accordance with Airbus Service Bulletin A320–57–1043, Revision 02, dated May 14, 1997, or Revision 05, dated April 30, 2002. Accomplishment of the repair of cracked area(s) and modification of uncracked area(s) constitutes terminating action for the requirements of paragraph (a) of this AD.

(d) If any cracking is detected during any inspection required by paragraph (b) or (c)(2) of this AD, and the applicable service bulletin specifies to contact Airbus for an appropriate action: Prior to further flight, repair in accordance with a method approved by either the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate; or the Direction Générale de l'Aviation Civile (or its delegated agent).

(e) The actions required by paragraph (a) of this AD are not required to be accomplished if the requirements of paragraph (c) of this AD are accomplished at the time specified in paragraph (a) of this AD.

New Requirements of This AD

Initial Inspection

(f) For airplanes on which the inspection required by paragraph (a) of this AD has not been done before the effective date of this AD: Perform a high frequency eddy current inspection to detect fatigue cracking of the lower surface panel on the wing center box, in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997, or Revision 03, dated April 30, 2002; at the later of the times specified in paragraphs (f)(1) and (f)(2) of this AD.

Accomplishment of the inspections required by this paragraph terminates the requirements of paragraph (a) of this AD.

(1) Prior to the accumulation of 13,200 total flight cycles or 39,700 total flight hours, whichever is first.

(2) Prior to the accumulation of 20,000 total flight cycles, or within 3,500 flight cycles after the effective date of this AD, whichever is first.

Repetitive Inspections

(g) If no cracking is detected during the inspection required by paragraph (a) or (f) of this AD: Repeat the inspection at the applicable time specified in paragraph (g)(1) or (g)(2) of this AD.

(1) For airplanes on which the inspections required by paragraph (a) of this AD have been done before the effective date of this AD: Do the next inspection within 5,700 flight cycles after accomplishment of the last inspection, or within 1,800 flight cycles after the effective date of this AD, whichever is later. Repeat the inspection thereafter at intervals not to exceed 5,700 flight cycles.

(2) For airplanes on which no inspection required by paragraph (a) of this AD has been done before the effective date of this AD: Do the next inspection within 5,700 flight cycles after accomplishment of the inspection required by paragraph (f) of this AD. Repeat the inspection thereafter at intervals not to exceed 5,700 flight cycles.

Repair/Modification

(h) If any cracking is detected during any inspection required by paragraph (f) or (g) of this AD, prior to further flight, repair in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997; or Revision 03, dated April 30, 2002; and modify any uncracked area in accordance with Airbus Service Bulletin A320–57–1043, Revision 02, dated May 14, 1997; or Revision 05, dated April 30, 2002. Where Service Bulletin A320–57–1082 specifies to contact Airbus for an appropriate repair action: Prior to further flight, repair in accordance with a method approved by either the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate; or the Direction Générale de l'Aviation Civile (DGAC) (or its delegated agent). Accomplishment of the repair of cracked area(s) and modification of uncracked area(s) constitutes terminating action for the requirements of this AD.

Actions Done per Previous Issues of Service Bulletins

(i) Accomplishment of inspections and repairs before the effective date of this AD in accordance with Airbus Service Bulletin A320–57–1082, Revision 02, dated July 26, 1999; and accomplishment of the modification before the effective date of this AD in accordance with Airbus Service Bulletin Airbus Service Bulletin A320–57–1043, dated February 16, 1993; Revision 01, dated June 14, 1996; Revision 03, dated October 24, 1997; or Revision 04, dated March 15, 1999; are considered acceptable for compliance with the applicable actions specified in this AD.

Alternative Methods of Compliance

(j) In accordance with 14 CFR 39.19, the Manager, International Branch, ANM–116, is authorized to approve alternative methods of compliance for this AD.

Note 1: The subject of this AD is addressed in French airworthiness directive 2002–342(B), dated June 26, 2002.

Issued in Renton, Washington, on January 29, 2004.

Kalene C. Yanamura,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 04–2481 Filed 2–5–04; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2003–NM–25–AD]

RIN 2120–AA64

Airworthiness Directives; Saab Model SAAB SF340A and SAAB 340B Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Saab Model SAAB SF340A and SAAB 340B series airplanes. This proposal would require replacement of certain assistor springs and bearings with certain new assistor springs and bearings. This action is necessary to prevent possible collapse of a main landing gear upon landing and consequent reduced controllability of the airplane. This action is intended to address the identified unsafe condition.

DATES: Comments must be received by March 8, 2004.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2003-NM-25-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: 9-anm-nprmcomment@faa.gov. Comments sent via fax or the Internet must contain "Docket No. 2003-NM-25-AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 or 2000 or ASCII text.

The service information referenced in the proposed rule may be obtained from Saab Aircraft AB, SAAB Aircraft Product Support, S-581.88, Linköping, Sweden. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington.

FOR FURTHER INFORMATION CONTACT: Todd Thompson, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-1175; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date

for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this action may be changed in light of the comments received.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.
- For each issue, state what specific change to the proposed AD is being requested.
- Include justification (*e.g.*, reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2003-NM-25-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2003-NM-25-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

Discussion

Luftfartsverket (LFV), which is the airworthiness authority for Sweden, notified the FAA that an unsafe condition may exist on certain Saab Model SAAB SF340A and SAAB 340B series airplanes. LFV advises that certain assistor springs of the main landing gear with the wire-formed end loop design have failed in service. Further, one known incident has been reported where the assistor spring cable (wire-rope) prevented the main landing gear from locking in fully down position. Such assistor spring failure, if not corrected, could result in collapse of a main landing gear upon landing and consequent reduced controllability of the airplane.

Explanation of Relevant Service Information

Saab has issued Service Bulletin 340-32-130, dated April 28, 2003, which describes procedures for replacement of assistor springs and bearings of the main landing gear with new assistor springs and bearings, respectively. Accomplishment of the actions specified in the service bulletin is intended to adequately address the identified unsafe condition. LFV classified this service bulletin as mandatory and issued Swedish airworthiness directive No 1-191, dated April 28, 2003 to ensure the continued airworthiness of these airplanes in Sweden.

LFV refers to Saab Service Bulletin 340-32-126, dated December 18, 2002, as a factor in determining the appropriate compliance time for accomplishment of the actions required by this AD.

FAA's Conclusions

These airplane models are manufactured in Sweden and are type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, LFV has kept the FAA informed of the situation described above. The FAA has examined the findings of LFV, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, the proposed AD would require accomplishment of the actions specified in the service bulletin described previously.

Cost Impact

The FAA estimates that 288 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 2 work hours per airplane to accomplish the proposed actions, and that the average labor rate is \$65 per work hour. Required parts would cost approximately \$750 per airplane. Based on these figures, the cost impact of the proposed AD on U.S. operators is estimated to be \$253,440, or \$880 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

TABLE 1.—COMPLIANCE TIMES

Within—	If—
6 months after the effective date of this AD	Saab SB 340–32–126, dated December 18, 2002 has been performed.
3 months after the effective date of this AD	Saab SB 340–32–126, dated December 18, 2002 has not been performed.

(1) Replace the assistor springs of the main landing gear with new assistor springs, per Part 1 of the service bulletin.

(2) Replace the bearings of the main landing gear with new bearings, per Part 2 of the service bulletin.

Parts Installation

(b) As of the effective date of this AD, no person may install an assistor spring, part number AIR125132 or AIR131330, on any airplane.

Alternative Methods of Compliance

(c) In accordance with 14 CFR 39.19, the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate, is authorized to approve alternative methods of compliance for this AD.

Note 1: The subject of this AD is addressed in Swedish airworthiness directive No 1–191, dated April 28, 2003.

Issued in Renton, Washington, on January 29, 2004.

Kalene C. Yanamura,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.
[FR Doc. 04–2482 Filed 2–5–04; 8:45 am]

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PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Saab Aircraft AB: Docket 2003–NM–25–AD.

Applicability: Model SAAB SF340A series airplanes with serial numbers 004 through 159 inclusive, and Model SAAB 340B series airplanes with serial numbers 160 through 459 inclusive; certificated in any category.

Compliance: Required as indicated, unless accomplished previously.

To prevent possible collapse of a main landing gear upon landing and consequent reduced controllability of the airplane, accomplish the following:

Replacements

(a) Within the Compliance Times listed in Table 1 of this AD, perform the actions specified in paragraphs (a)(1) and (a)(2) of this AD per the Accomplishment Instructions of Saab Service Bulletin (SB) 340–32–130, dated April 28, 2003.

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002–NM–278–AD]

RIN 2120–AA64

Airworthiness Directives; Airbus Model A319 and A320 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Airbus Model A319 and A320 series airplanes. This proposal would require modifying the electrical bonding of the fuel return line in each wing between ribs 7 and 8. This action is necessary to reduce the potential for electrical arcing within the fuel tank due to insufficient electrical bonding, which could result in a fire or explosion in the fuel tank. This action is intended

to address the identified unsafe condition.

DATES: Comments must be received by March 8, 2004.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–114, Attention: Rules Docket No. 2002–NM–278–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227–1232. Comments may also be sent via the Internet using the following address: 9-anm-nprmcomment@faa.gov. Comments sent via fax or the Internet must contain "Docket No. 2002–NM–278–AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 or 2000 or ASCII text.

The service information referenced in the proposed rule may be obtained from Airbus, 1 Rond Point Maurice Bellonte,