

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2018–02–16 Bombardier, Inc.: Amendment 39–19169; Docket No. FAA–2017–0621; Product Identifier 2017–NM–049–AD.

(a) Effective Date

This AD is effective March 5, 2018.

(b) Affected ADs

None.

(c) Applicability

This AD applies to Bombardier, Inc., Model DHC–8–400, –401, and –402 airplanes, certificated in any category, serial numbers 4001 and 4003 through 4488 inclusive, except those incorporating Bombardier ModSum IS4Q5200050.

(d) Subject

Air Transport Association (ATA) of America Code 52, Doors.

(e) Reason

This AD was prompted by reports of interrupted operation of translating fuselage doors caused by corrosion in the door lift and latch shaft roller bearings. We are issuing this AD to detect and correct bearing corrosion and prevent door operation interruptions that could inhibit safe evacuation of the airplane in an emergency.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Inspection and Replacement of Bearings

Within 6,000 flight hours or 36 months after the effective date of this AD, whichever occurs earlier, do a detailed visual inspection of all translating fuselage door lift and latch shaft roller bearings for signs of corrosion, damaged seals, and loss of lubricant; replace any corroded bearings; and apply corrosion-inhibiting compound (CIC); in accordance with paragraph 3.B., “Procedure,” of the Accomplishment Instructions of Bombardier Service Bulletin 84–52–88, dated April 14, 2016.

(h) Credit for Previous Actions

This paragraph provides credit for actions required by paragraph (g) of this AD, if those actions were performed before the effective date of this AD using Bombardier Service Bulletin 84–52–85, dated September 23, 2015; or Bombardier Service Bulletin 84–52–85, Revision A, dated January 22, 2016.

(i) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, New York ACO Branch, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the certification office, send it to ATTN: Program Manager, Continuing Operational Safety, FAA, New York ACO Branch, 1600 Stewart Avenue, Suite 410, Westbury, NY 11590; telephone 516–228–7300; fax 516–794–5531. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(2) *Contacting the Manufacturer:* For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, New York ACO Branch, FAA; or Transport Canada Civil Aviation (TCCA); or Bombardier, Inc.’s TCCA Design Approval Organization (DAO). If approved by the DAO, the approval must include the DAO-authorized signature.

(j) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) Canadian Airworthiness Directive CF–2016–18, dated June 6, 2016, for related information. This MCAI may be found in the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2017–0621.

(2) For more information about this AD, contact Aziz Ahmed, Aerospace Engineer, Airframe and Mechanical Systems Section, FAA, New York ACO Branch, 1600 Stewart Avenue, Suite 410, Westbury, NY 11590; telephone 516–228–7329; fax 516–794–5531.

(3) Service information identified in this AD that is not incorporated by reference is available at the addresses specified in paragraphs (k)(3) and (k)(4) of this AD.

(k) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(i) Bombardier Service Bulletin 84–52–88, dated April 14, 2016.

(ii) Reserved.

(3) For service information identified in this AD, contact Bombardier, Inc., Q-Series Technical Help Desk, 123 Garratt Boulevard, Toronto, Ontario M3K 1Y5, Canada; telephone 416–375–4000; fax 416–375–4539; email thd.qseries@aero.bombardier.com; internet <http://www.bombardier.com>.

(4) You may view this service information at the FAA, Transport Standards Branch, 1601 Lind Avenue SW, Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on January 17, 2018.

Michael Kaszycki,

Acting Director, System Oversight Division, Aircraft Certification Service.

[FR Doc. 2018–01317 Filed 1–26–18; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2017–0896; Product Identifier 2017–SW–034–AD; Amendment 39–19166; AD 2018–02–13]

RIN 2120–AA64

Airworthiness Directives; Sikorsky Aircraft Corporation Helicopters

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are superseding Airworthiness Directive (AD) 2017–07–02 for Sikorsky Aircraft Corporation (Sikorsky) Model 269D and Model 269D Configuration A helicopters. AD 2017–07–02 required reducing the life limit of and inspecting certain drive shafts. This new AD retains the requirements of AD 2017–07–02 and requires repeating the inspections. The actions of this AD are intended to detect and prevent an unsafe condition on these products.

DATES: This AD is effective March 5, 2018.

ADDRESSES: For service information identified in this final rule, contact Sikorsky Aircraft Corporation, Customer Service Engineering, 124 Quarry Road, Trumbull, CT 06611; telephone 1-800-Winged-S or 203-416-4299; email wcs_cust_service_eng.gr-sik@lmco.com. You may view this referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy, Room 6N-321, Fort Worth, TX 76177.

Examining the AD Docket

You may examine the AD docket on the internet at <http://www.regulations.gov> in Docket No. FAA-2017-0896; or in person at Docket Operations between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the economic evaluation, any comments received, and other information. The address for Docket Operations (phone: 800-647-5527) is Document Operations, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE, Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT:

Michael Schwetz, Aviation Safety Engineer, Boston ACO Branch, Compliance and Airworthiness Division, FAA, 1200 District Avenue, Burlington, Massachusetts 01803; telephone (781) 238-7761; email michael.schwetz@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to remove AD 2017-07-02, Amendment 39-18840 (82 FR 15120, March 27, 2017) and add a new AD. AD 2017-07-02 applied to Sikorsky Model 269D and Model 269D Configuration A helicopters with a Kaflex engine side drive shaft part number (P/N) SKCP2738-7 and Kaflex pulley side drive shaft P/N SKCP2738-5 installed. AD 2017-07-02 required reducing the life limit of the drive shafts and performing several inspections of the drive shafts within 25 hours time-in-service (TIS). AD 2017-07-02 also specified replacing the drive shaft assemblies as an optional terminating action for the requirements of the AD. AD 2017-07-02 was prompted by four incidents involving failure of the engine side drive shaft. The actions required by AD 2017-07-02 were intended to prevent failure of the drive shaft, loss of rotor drive, and subsequent loss of control of the helicopter.

The NPRM published in the **Federal Register** on September 22, 2017 (82 FR

44353) to provide the public an opportunity to comment on proposed requirements with longer intervals. The NPRM proposed to retain the requirements of AD 2017-07-02 and repeat some of the inspections every 100 hours TIS or 400 hours TIS. Repeating these inspections is necessary to detect and prevent the unsafe condition.

Comments

We gave the public the opportunity to participate in developing this AD, but we did not receive any comments on the NPRM.

FAA's Determination

We have reviewed the relevant information and determined that an unsafe condition exists and is likely to exist or develop on other helicopters of these same type designs and that air safety and the public interest require adopting the AD requirements as proposed.

Related Service Information

We reviewed Appendix B to Sikorsky S-330 Model 269D Helicopter Basic Handbook of Maintenance Instructions No. CSP-D-2, dated February 1, 1993, and revised October 15, 2014; and Appendix B to Sikorsky S-333 Model 269D Config. "A" Helicopter Basic Handbook of Maintenance Instructions No. CSP-D-9, dated July 20, 2001, and revised October 15, 2014. This service information specifies repetitive inspection procedures, overhaul and retirement schedules, and weight and balance procedures. The Airworthiness Limitations section, which is included in this service information, contains the life limits for drive shaft assembly P/Ns SKCP2738-5 and SKCP2738-7.

We also reviewed Sikorsky 269D Helicopter Alert Service Bulletin DB-052, Basic Issue, dated January 16, 2014, for Sikorsky Model 269D and Model 269D Configuration A helicopters. This service information distributes the service life reduction information and implements a new 1,200-hour overhaul inspection for drive shaft assembly P/Ns SKCP2738-3, SKCP2738-5, and SKCP2738-7.

Differences Between This AD and the Service Information

The Sikorsky service information specifies a drive shaft assembly service life of 3,000 hours TIS with a 1,200 hour overhaul inspection for Model 269D Configuration A helicopters, while this AD specifies a service life of 1,200 hours TIS.

The Sikorsky service information specifies different inspection procedures if there is spline engagement

interference or resistance while inspecting the drive shaft alignment. This AD specifies replacing both the engine side and pulley side drive shafts if there is any spline engagement interference or resistance.

The Sikorsky service information specifies inspecting the working fastener condition without any specific succeeding action regarding the inspection. This AD specifies replacing both the engine side and pulley side drive shafts if there is any joint movement.

The Sikorsky service information specifies returning the drive shaft assembly to Sikorsky if there is fretting dust or red metallic residue at a joint. This AD specifies replacing both the engine side and pulley side drive shafts if there is any fretting corrosion.

Costs of Compliance

We estimate that this AD will affect 18 helicopters of U.S. Registry. We estimate that operators may incur the following costs in order to comply with this AD.

Removing the engine side and pulley side drive shafts that have reached the new life limit will take about 4 work-hours for a cost of \$340 per helicopter. Inspecting the lower pulley to engine alignment using the belt alignment tool will take about 0.5 work-hour for an estimated cost of \$43 per helicopter and \$774 for the U.S. fleet per inspection cycle. Adjusting the engine elevation alignment will take about 0.5 work-hour for an estimated cost of \$43 per helicopter. Inspecting the drive shaft alignment by checking spline engagement will take about 1 work-hour for a cost of \$85 per helicopter and \$1,530 for the U.S. fleet per inspection cycle. Inspecting the drive shafts for damage will take about 1 work-hour for an estimated cost of \$85 per helicopter and \$1,530 for the U.S. fleet per inspection cycle. Inspecting the joints will take about 1 work-hour for an estimated cost of \$85 per helicopter and \$1,530 for the U.S. fleet per inspection cycle. Replacing the engine side and pulley side drive shafts, if required, will take about 8 work-hours and parts will cost about \$20,000, for an estimated cost of \$20,680 per helicopter.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska to the extent that a regulatory distinction is required, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD) 2017-07-02, Amendment 39-18840 (82 FR 15120, March 27, 2017), and adding the following new AD:

2018-02-13 Sikorsky Aircraft Corporation (Sikorsky): Amendment 39-19166; Docket No. FAA-2017-0896; Product Identifier 2017-SW-034-AD.

(a) Applicability

This AD applies to Sikorsky Model 269D and Model 269D Configuration A helicopters with a KAflex engine side drive shaft part number (P/N) SKCP2738-7 and KAflex pulley side drive shaft P/N SKCP2738-5 installed, certificated in any category.

(b) Unsafe Condition

This AD defines the unsafe condition as failure of a drive shaft. This condition could result in loss of rotor drive and subsequent loss of control of the helicopter.

(c) Affected ADs

This AD supersedes AD 2017-07-02, Amendment 39-18840 (82 FR 15120, March 27, 2017).

(d) Effective Date

This AD becomes effective March 5, 2018.

(e) Compliance

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

(f) Required Actions

- (1) Before further flight:
 - (i) For Model 269D helicopters, remove from service any KAflex engine side drive shaft P/N SKCP2738-7 and any KAflex pulley side drive shaft P/N SKCP2738-5 that has 6,000 or more hours time-in-service (TIS). Thereafter, remove from service any KAflex engine side drive shaft P/N SKCP2738-7 and any KAflex pulley side drive shaft P/N SKCP2738-5 before accumulating 6,000 hours TIS.
 - (ii) For Model 269D Configuration A helicopters, remove from service any KAflex engine side drive shaft P/N SKCP2738-7 and any KAflex pulley side drive shaft P/N SKCP2738-5 that has 1,200 or more hours TIS. Thereafter, remove from service any KAflex engine side drive shaft P/N SKCP2738-7 and any KAflex pulley side drive shaft P/N SKCP2738-5 before accumulating 1,200 hours TIS.
 - (iii) If interchanged between Model 269D and Model 269D Configuration A helicopters, remove from service any KAflex engine side drive shaft P/N SKCP2738-7 and any KAflex pulley side drive shaft P/N SKCP2738-5 that has 1,200 or more hours TIS. Thereafter, if interchanged between Model 269D and Model 269D Configuration A helicopters, remove from service any KAflex engine side drive shaft P/N SKCP2738-7 and any KAflex pulley side drive shaft P/N SKCP2738-5 before accumulating 1,200 hours TIS.
- (2) Within 25 hours TIS, and thereafter at intervals not to exceed 25 hours TIS, using a belt drive alignment tool 269T3303-003, inspect the lower pulley to engine alignment by engaging the tool on the drive shaft and inserting in the lower pulley bore. Rotate the tool 360° around the drive shaft and inspect for interference. If there is any interference with the rotation of the tool, before further flight, adjust the engine elevation alignment to eliminate the interference.
- (3) Within 25 hours TIS, and thereafter at intervals not to exceed 100 hours TIS:
 - (i) Remove the drive shaft to adapter bolt and inspect the drive shaft alignment. Engage

and disengage the splines a minimum of 3 times by sliding the engine power output shaft in and out of the engine. Inspect the alignment at each 90° interval by rotating the lower pulley with the power shaft disengaged. Determine whether the adapter slides on and off the drive shaft splines without spline engagement interference or resistance along the entire length of movement. If there is any spline engagement interference or resistance, before further flight, replace both the engine side and pulley side drive shafts.

(ii) Inspect each drive shaft for a crack, any corrosion or pitting, a nick, a dent, and a scratch. If there is a crack, any corrosion or pitting, a nick, a dent, or a scratch that exceeds allowable limits, before further flight, replace both the engine side and pulley side drive shafts.

(4) Within 25 hours TIS, and thereafter at intervals not to exceed 400 hours TIS, remove the engine side drive shaft and pulley side drive shaft and perform the following:

(i) Inspect each flex frame (frame) bolted joint (joint) for movement by hand. If there is any movement, before further flight, replace both the engine side and pulley side drive shafts.

(ii) Visually inspect each joint for fretting corrosion (which might be indicated by metallic particles) and each frame and mount bolt torque stripe for movement. If there is any fretting corrosion or torque stripe movement, before further flight, replace both the engine side and pulley side drive shafts.

(iii) Using a 10x or higher power magnifying glass, visually inspect each joint for fretting and for a crack around the bolt head and washer side, and around the nut and washer side. Also inspect both sides of each frame for a crack on the inside and outside corner radii and radii edge (four). If there is any fretting, a crack at any point over the full circumference (360°) of the bolt head and washer side or the nut and washer side, or a crack in any of the corner radii edges, before further flight, replace both the engine side and pulley side drive shafts.

(5) As an optional terminating action to the repetitive inspections in this AD, you may install KAflex engine side drive shaft P/N SKCP2738-9 and KAflex pulley side drive shaft P/N SKCP2738-101.

(g) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Boston ACO Branch, FAA, may approve AMOCs for this AD. Send your proposal to: Michael Schwetz, Aviation Safety Engineer, Boston ACO Branch, Compliance and Airworthiness Division, FAA, 1200 District Avenue, Burlington, Massachusetts 01803; telephone (781) 238-7761; email michael.schwetz@faa.gov.

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office, before operating any aircraft complying with this AD through an AMOC.

(h) Additional Information

Appendix B of Sikorsky S-330 Model 269D Helicopter Basic Handbook of Maintenance Instructions, No. CSP-D-2, dated February 1, 1993, and revised October 15, 2014; Appendix B of Sikorsky S-330 Model 269D Config. "A" Helicopter Basic Handbook of Maintenance Instructions, No. CSP-D-9, dated July 20, 2001, and revised October 15, 2014; and Sikorsky 269D Helicopter Alert Service Bulletin DB-052, Basic Issue, dated January 16, 2014, which are not incorporated by reference, contain additional information about the subject of this AD. For service information identified in this AD, contact Sikorsky Aircraft Corporation, Customer Service Engineering, 124 Quarry Road, Trumbull, CT 06611; telephone 1-800-Winged-S or 203-416-4299; email wcs_cust_service_eng.gr-sik@lmco.com. You may review a copy of the service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy, Room 6N-321, Fort Worth, TX 76177.

(i) Subject

Joint Aircraft Service Component (JASC)
Code: 6310, Engine/Transmission Coupling.

Issued in Fort Worth, Texas, on January 17, 2018.

Lance T. Gant,

*Director, Compliance & Airworthiness
Division, Aircraft Certification Service.*

[FR Doc. 2018-01572 Filed 1-26-18; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF JUSTICE**28 CFR Part 85**

[Docket No. OAG 159; AG Order No. 4093-2018]

Civil Monetary Penalties Inflation Adjustment

AGENCY: Department of Justice.

ACTION: Final rule.

SUMMARY: The Department of Justice is adjusting for inflation the civil monetary penalties assessed or enforced by components of the Department, in accordance with the provisions of the Bipartisan Budget Act of 2015, for penalties assessed after January 29, 2018, with respect to violations occurring after November 2, 2015.

DATES: This rule is effective January 29, 2018.

FOR FURTHER INFORMATION CONTACT: Robert Hinchman, Senior Counsel, Office of Legal Policy, U.S. Department of Justice, Room 4252 RFK Building, 950 Pennsylvania Avenue NW, Washington, DC 20530, telephone (202) 514-8059 (not a toll-free number).

SUPPLEMENTARY INFORMATION:**I. Statutory Process for Implementing Annual Inflation Adjustments**

Section 701 of the Bipartisan Budget Act of 2015, Public Law 114-74 (Nov. 2, 2015) ("BBA"), 28 U.S.C. 2461 note, substantially revised the prior provisions of the Federal Civil Monetary Penalties Inflation Adjustment Act of 1990, Public Law 101-410 (the "Inflation Adjustment Act"), and substituted a different statutory formula for calculating inflation adjustments on an annual basis.

In accordance with the provisions of the BBA, on June 30, 2016 (81 FR 42491), the Department of Justice published an interim rule to adjust for inflation the civil monetary penalties assessed or enforced by components of the Department after August 1, 2016, with respect to violations occurring after November 2, 2015, the date of enactment of the BBA. Readers may refer to the Supplementary Information (also known as the preamble) of the Department's 2016 interim rule for additional background information regarding the statutory authority for adjustments of civil monetary penalty amounts to take account of inflation and the Department's past implementation of inflation adjustments.

The BBA also provides for agencies to adjust their civil penalties on January 15 of each year to account for inflation during the preceding year, rounded to the nearest dollar. Accordingly, on February 3, 2017 (82 FR 9131), the Department published a final rule to adjust for inflation the civil monetary penalties assessed or enforced by components of the Department after that date, with respect to violations occurring after November 2, 2015.

II. Inflation Adjustments Made by This Rule

As required, the Department is publishing this final rule to adjust the civil penalties that were most recently adjusted as of February 3, 2017. Under the statutory formula, the adjustments made by this rule are based on the Bureau of Labor Statistics' Consumer Price Index for October 2017. The OMB Memorandum for the Heads of Executive Departments and Agencies M-18-03 (Dec. 15, 2017), <https://www.whitehouse.gov/wp-content/uploads/2017/11/M-18-03.pdf> (last visited Jan. 1, 2018), instructs that the applicable inflation factor for this adjustment is 1.02041. Accordingly, this rule adjusts the civil penalty amounts in 28 CFR 85.5 by applying this inflation factor mechanically to each of the civil penalty amounts listed (rounded to the nearest dollar).

Example:

- In 2016, the Program Fraud Civil Remedies Act penalty was increased to \$10,781 in accordance with the adjustment requirements of the BBA.
- For 2017, where the applicable inflation factor was 1.01636, the existing penalty of \$10,781 was multiplied by 1.01636 and revised to \$10,957 (rounded to the nearest dollar).
- For this final rule in 2018, where the applicable inflation factor is 1.02041, the existing penalty of \$10,957 is multiplied by 1.02041 and revised to \$11,181 (rounded to the nearest dollar).

This rule adjusts for inflation civil monetary penalties within the jurisdiction of the Department of Justice for purposes of the Inflation Adjustment Act, as amended. Other agencies are responsible for the inflation adjustments of certain other civil monetary penalties that the Department's litigating components bring suit to collect. The reader should consult the regulations of those other agencies for inflation adjustments to those penalties.

III. Effective Date of Adjusted Civil Penalty Amounts

Under this rule, the adjusted civil penalty amounts are applicable only to civil penalties assessed after January 29, 2018, with respect to violations occurring after November 2, 2015, the date of enactment of the BBA.

The penalty amounts set forth in the existing table in 28 CFR 85.5 are applicable to civil penalties assessed after August 1, 2016, and on or before the effective date of this rule, with respect to violations occurring after November 2, 2015. Civil penalties for violations occurring on or before November 2, 2015, and assessments made on or before August 1, 2016, will continue to be subject to the civil monetary penalty amounts set forth in the Department's regulations in 28 CFR parts 20, 22, 36, 68, 71, 76, and 85 as such regulations were in effect prior to August 1, 2016 (or as set forth by statute if the amount had not yet been adjusted by regulation prior to August 1, 2016).

Statutory and Regulatory Analyses*Administrative Procedure Act*

The BBA provides that, for each annual adjustment made after the initial adjustments of civil penalties in 2016, the head of an agency shall adjust the civil monetary penalties each year notwithstanding 5 U.S.C. 553. Accordingly, this rule is being issued as a final rule without prior notice and public comment, and without a delayed effective date.