

or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD docket. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

##### § 39.13 [Amended]

- 2. The Federal Aviation Administration (FAA) amends § 39.13 by adding the following new airworthiness directive (AD):

##### 2007–24–01 Hawker Beechcraft

**Corporation:** Amendment 39–15267.  
Docket No. FAA–2007–28883;  
Directorate Identifier 2007–NM–106–AD.

#### Effective Date

- (a) This AD becomes effective January 2, 2008.

#### Affected ADs

- (b) None.

#### Applicability

- (c) This AD applies to Hawker Beechcraft Model 400A series airplanes, certificated in any category; as identified in Raytheon Service Bulletin SB 25–3758, dated June 2006.

#### Unsafe Condition

- (d) This AD results from reports of undersized, and consequently unprotected, wire in the galley cabinets. We are issuing this AD to prevent overheating of wire insulation and consequent fire or smoke in the airplane cabin.

#### Compliance

- (e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

#### Inspection and Related Investigative/Corrective Actions

- (f) Within 200 flight hours or 12 months after the effective date of this AD, whichever occurs first, inspect the galley cabinets to determine if Precision Pattern galley cabinet, part number (P/N) 20917, 20918, or 20921 is installed, or if 8 American Wire Gauge (AWG) wire already exists; and, within 20 flight hours or 30 days after the inspection, whichever occurs later, do all applicable related investigative and corrective actions. The actions must be done in accordance with the Accomplishment Instructions of Raytheon Service Bulletin SB 25–3758, dated June 2006.

**Note 1:** Raytheon Service Bulletin SB 25–3758, dated June 2006, refers to Raytheon Kit 128–3068–0001, Revision 3, dated April 18, 2006, as an additional source of service information for replacing the undersized 10AWG wire with 8AWG wire in the gallery power circuit.

#### Alternative Methods of Compliance (AMOCs)

- (g)(1) The Manager, Wichita Aircraft Certification Office, FAA, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

- (2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

#### Material Incorporated by Reference

- (h) You must use Raytheon Service Bulletin SB 25–3758, dated June 2006, to perform the actions that are required by this AD, unless the AD specifies otherwise. The Director of the Federal Register approved the incorporation by reference of this document in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Contact Hawker Beechcraft Corporation, 9709 East Central, Wichita, Kansas 67206, for a copy of this service information. You may review copies at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on November 8, 2007.

#### Ali Bahrami,

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. E7–22545 Filed 11–27–07; 8:45 am]

**BILLING CODE 4910–13–P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA–2007–0193; Directorate Identifier 2007–NE–43–AD; Amendment 39–15273; AD 2007–24–07]

RIN 2120–AA64

#### Airworthiness Directives; General Electric Company (GE) CF6–80C2B1 Turbofan Engine

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule; request for comments.

**SUMMARY:** The FAA is adopting a new airworthiness directive (AD) for a GE CF6–80C2B1 turbofan engine, serial number (SN) 690203, with fan disk, part number (P/N) 1703M78P11, SN RPDMA662, installed. This AD requires stripping of thermal spray coating, inspection of dovetail slots, and reapplication of thermal spray coating on certain stage 1 fan disks. This AD results from a report that a repair shop did not meet the process requirements when applying copper-nickel-indium (Cu-Ni-In) thermal coating to certain stage 1 fan disks. We are issuing this AD to prevent possible uncontained release of multiple fan blades, resulting in damage to the airplane.

**DATES:** This AD becomes effective January 2, 2008. The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulations as of January 2, 2008.

We must receive any comments on this AD by January 28, 2008.

**ADDRESSES:** Use one of the following addresses to comment on this AD:

- **Federal eRulemaking Portal:** Go to <http://www.regulations.gov> and follow the instructions for sending your comments electronically.

- **Mail:** U.S. Docket Management Facility, Department of Transportation, 1200 New Jersey Avenue, SE., West Building Ground Floor, Room W12–140, Washington, DC 20590–0001.

- **Hand Delivery:** Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

- **Fax:** (202) 493–2251.

Contact General Electric Company via Lockheed Martin Technology Services, 10525 Chester Road, Suite C, Cincinnati, Ohio 45215, telephone (513) 672–8400, fax (513) 672–8422, for the service information identified in this AD.

**FOR FURTHER INFORMATION CONTACT:**

James Lawrence, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone: (781) 238-7176, fax: (781) 238-7199.

**SUPPLEMENTARY INFORMATION:**

GE reported that while recertifying a non-GE repair shop, they found the shop's process for coating dovetail slots with a Cu-Ni-In thermal coating did not meet the standard requirements of GE Substantiation Requirements Repair Document No. RSS-012-S2. Further investigation by GE identified twelve affected stage 1 fan disks. We discussed the issue with GE and the repair shop, and we agreed to allow the repair shop to strip, inspect, and recoat the 12 disks instead of issuing an AD to require the actions. The repair shop has stripped, inspected, and recoated 11 of the 12 disks. They have not been able to get the final stage 1 disk, P/N 1703M78P11, SN RPDMA662, to strip, inspect, and recoat that disk and it remains in service. This condition, if not corrected, could result in an uncontained release of multiple fan blades, and possible damage to the airplane.

**Relevant Service Information**

We have reviewed and approved the technical contents of GE Service Bulletin (SB) No. CF6-80C2 S/B 72-1121, dated January 23, 2004, that references procedures for stripping, inspecting, and recoating the affected stage 1 fan disks.

**Differences Between This AD and the Service Information**

GE SB No. CF6-80C2 S/B 72-1121, dated January 23, 2004, recommends performing the actions within 3,500 cycles-since-last Cu-Ni-In thermal spray coating of the dovetail slots. This AD requires performing the actions within 3,500 cycles-since-last Cu-Ni-In thermal spray coating of the dovetail slots, but before March 31, 2008.

**FAA's Determination and Requirements of This AD**

Although no airplanes that are registered in the United States use these GE CF6-80-C2B1 turbofan engines, the possibility exists that the engines could be used on airplanes that are registered in the United States in the future. The unsafe condition described previously is likely to exist or develop on other GE CF6-80C2B1 turbofan engines of the same type design. We are issuing this AD to prevent possible uncontained release of multiple fan blades, resulting in damage to the airplane. You must use the service information described

previously to perform the actions required by this AD.

**FAA's Determination of the Effective Date**

Since there are currently no domestic operators of this engine model, notice and opportunity for public comment before issuing this AD are unnecessary. A situation exists that allows the immediate adoption of this regulation.

**Comments Invited**

This AD is a final rule that involves requirements affecting flight safety and was not preceded by notice and an opportunity for public comment. However, we invite you to send us any written relevant data, views, or arguments regarding this AD. Send your comments to an address listed under **ADDRESSES**. Include "AD Docket No. FAA-2007-0193; Directorate Identifier 2007-NE-43-AD" in the subject line of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify it.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact with FAA personnel concerning this AD. Using the search function of the DMS Web site, anyone can find and read the comments in any of our dockets, including the name of the individual who sent the comment (or signed the comment on behalf of an association, business, labor union, etc.). You may review the DOT's complete Privacy Act Statement in the **Federal Register** published on April 11, 2000 (65 FR 19477-78).

**Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone (800) 647-5527) is the same as the Mail address provided in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII,

Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

**Regulatory Findings**

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a summary of the costs to comply with this AD and placed it in the AD Docket. You may get a copy of this summary at the address listed under **ADDRESSES**.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

■ Under the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive:

**2007-24-07 General Electric Company:**  
Amendment 39-15273. Docket No.  
FAA-2007-0193; Directorate Identifier  
2007-NE-43-AD.

#### Effective Date

(a) This airworthiness directive (AD) becomes effective January 2, 2008.

#### Affected ADs

(b) None.

#### Applicability

(c) This AD applies to GE CF6-80C2B1 turbofan engine, serial number 690203, with fan disk, part number (P/N) 1703M78P11, SN RMDA662, installed. This engine is installed on, but not limited to, a Boeing 747-300 airplane.

#### Unsafe Condition

(d) This AD results from a report that a repair shop did not meet the process requirements when applying copper-nickel-indium (Cu-Ni-In) thermal coating to certain stage 1 fan disks. We are issuing this AD to prevent possible uncontained release of multiple fan blades, resulting in damage to the airplane.

#### Compliance

(e) You are responsible for having the actions required by this AD performed within 3,500 cycles-since-last Cu-Ni-In thermal spray coating of the dovetail slots, but no later than March 31, 2008, unless the actions have already been done.

#### Stripping, Inspecting and Recoating the Stage 1 Fan Disk

(f) Strip the Cu-Ni-In thermal coating from the pressure faces and slot bottoms of the stage 1 fan disk, and perform a microstructure evaluation. Use 3.A.(2)(a) through 3.A.(2)(b) of GE Service Bulletin (SB) No. CF6-80C2 S/B 72-1121, dated January 23, 2004, to strip the thermal coating and perform the microstructure evaluation.

(g) Ultrasonic inspect, fluorescent penetrant inspect, and eddy current inspect stage 1 fan disk. Use 3.A.(2)(c) of GE SB No. CF6-80C2 S/B 72-1121, dated January 23, 2004, to inspect the disk.

(h) Apply Cu-Ni-In thermal coating to the pressure faces and slot bottoms of the stage 1 fan disks, using 3.A.(2)(d) of GE SB No. CF6-80C2 S/B 72-1121, dated January 23, 2004.

#### Alternative Methods of Compliance

(i) The Manager, Engine Certification Office, has the authority to approve alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19.

#### Related Information

(j) Contact James Lawrence, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone: (781) 238-7176, fax: (781) 238-7199, for more information about this AD.

#### Material Incorporated by Reference

(k) You must use GE Service Bulletin No. CF6-80C2 S/B 72-1121, dated January 23,

2004, to perform the actions required by this AD. The Director of the Federal Register approved the incorporation by reference of this service bulletin in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Contact General Electric Company via Lockheed Martin Technology Services, 10525 Chester Road, Suite C, Cincinnati, Ohio 45215, telephone (513) 672-8400, fax (513) 672-8422, for a copy of this service information. You may review copies at the FAA, New England Region, 12 New England Executive Park, Burlington, MA; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Burlington, Massachusetts, on November 15, 2007.

**Peter A. White,**

*Assistant Manager, Engine and Propeller Directorate, Aircraft Certification Service.*

[FR Doc. E7-22922 Filed 11-27-07; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2007-26966; Directorate Identifier 99-NE-01-AD; Amendment 39-15271; AD 2007-24-05]

**RIN 2120-AA64**

#### Airworthiness Directives; Rolls-Royce Corporation AE 3007A and AE 3007C Series Turbofan Engines

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** The FAA is superseding an existing airworthiness directive (AD) for Rolls-Royce Corporation (RRC) AE 3007A and AE 3007C series turbofan engines. That AD currently prohibits any flight following a ground engine start where the engine oil temperature is below 32 °F (0 °C), unless certain preflight operational procedures are followed. This AD also requires those actions and would also require a terminating action. This AD results from design improvements to components in the accessory gearbox air turbine starter mounting pad. We are issuing this AD to prevent an in-flight engine shutdown due to loss of engine oil from the engine accessory gearbox starter pad shaft seal drain and possible loss of the airplane.

**DATES:** This AD becomes effective January 2, 2008. The Director of the Federal Register approved the incorporation by reference of certain

publications listed in the regulations as of January 2, 2008.

**ADDRESSES:** You can get the service information identified in this AD from Rolls-Royce Corporation, P.O. Box 420, Indianapolis, IN 46206; telephone (317) 230-3774; fax (317) 230-8084; e-mail: [indy.pubs.services@rolls-royce.com](mailto:indy.pubs.services@rolls-royce.com).

The Docket Operations office is located at Docket Management Facility, U.S. Department of Transportation, 1200 New Jersey Avenue, SE., West Building Ground Floor, Room W12-140, Washington, DC 20590-0001.

**FOR FURTHER INFORMATION CONTACT:** Kyri Zaroyiannis, Aerospace Engineer, Chicago Aircraft Certification Office, Small Airplane Directorate, FAA, 2300 E. Devon Ave., Des Plaines, IL 60018; telephone (847) 294-7836; fax (847) 294-7834.

**SUPPLEMENTARY INFORMATION:** The FAA proposed to amend 14 CFR part 39 by superseding AD 99-02-51, Amendment 39-11108 (64 FR 16339, April 5, 1999), with a proposed AD. The proposed AD applies to RRC AE 3007A and AE 3007C series turbofan engines. We published the proposed AD in the **Federal Register** on March 29, 2007 (72 FR 14724). That action proposed to:

- Prohibit before further flight, any flight following a ground engine start where the engine oil temperature is below 32 °F (0 °C), unless certain preflight operational procedures are followed to ensure that there is no excessive loss of oil from leakage at the air turbine starter shaft; and
- Require terminating action to the prohibition requirements of the existing AD, by removing from service certain seal P/Ns from the accessory gearbox air turbine starter mounting pad and installing an improved seal; and
- Require removing certain P/N drain caps, drain adapters, and orifice inserts, and installing an open adapter on the starter pad drain.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone (800) 647-5527) is provided in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

#### Comments

We provided the public the opportunity to participate in the