

actions required by this AD, unless the AD specifies otherwise.

TABLE 2—ALL MATERIAL INCORPORATED BY REFERENCE

Document	Revision	Date
Airbus A318/A319/A320/A321 ALS Part 1—Safe Life Airworthiness Limitation Items .....	Revision 00 .....	February 28, 2006.
Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96.	Issue 7 .....	December 2005.
Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96.	Issue 08 .....	March 2006.
Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96.	Issue 09 .....	November 2006.
Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96.	Issue 10 .....	October 2009.
Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96.	Issue 11 .....	September 2010.

The issue level of Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96, Issue 10, dated October 2009; and Issue 11, dated September 2010; is indicated only on the title page and in the Record of Revisions of these documents.

(1) The Director of the Federal Register approved the incorporation by reference of Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96, Issue 10, dated October 2009; and Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96, Issue 11,

dated September 2010; under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The Director of the Federal Register previously approved the incorporation by reference of the service information contained in table 3 of this AD on November 7, 2007 (72 FR 56262, October 3, 2007).

TABLE 3—MATERIAL PREVIOUSLY INCORPORATED BY REFERENCE

Document	Revision	Date
Airbus A318/A319/A320/A321 ALS Part 1—Safe Life Airworthiness Limitation Items .....	Revision 00 .....	February 28, 2006.
Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96.	Issue 7 .....	December 2005.
Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96.	Issue 08 .....	March 2006.
Airbus A318/A319/A320/A321 Airworthiness Limitation Items, Document AI/SE-M4/95A.0252/96.	Issue 09 .....	November 2006.

(3) For service information identified in this AD, contact Airbus, Airworthiness Office—EAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; e-mail [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet <http://www.airbus.com>.

(4) You may review copies of the service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

Issued in Renton, Washington, on June 24, 2011.

**Ali Bahrami,**

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2011-16559 Filed 7-15-11; 8:45 am]

BILLING CODE 4910-13-P

**DEPARTMENT OF TRANSPORTATION**

**Federal Aviation Administration**

**14 CFR Part 39**

**[Docket No. FAA-2011-0309; Directorate Identifier 2010-NM-255-AD; Amendment 39-16755; AD 2011-15-08]**

**RIN 2120-AA64**

**Airworthiness Directives; Airbus Model A300 B4-600, B4-600R, and F4-600R Series Airplanes, and Model A300 C4-605R Variant F Airplanes (Collectively Called A300-600 Series Airplanes); and Model A310 Series Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation

product. The MCAI describes the unsafe condition as:

A specific failure case of the THSA [trimmable horizontal stabilizer actuator] upper primary attachment, which may result in a loading of the upper secondary attachment, has been identified by analysis.

Primary load path failure can be caused by bearing migration from the upper attachment gimbal by failure or loss of a retention bolt.

In case of failure of the THSA upper primary attachment, the THSA upper secondary attachment would engage. Because the upper attachment secondary load path can only withstand the loads for a limited period of time, the condition where it would be engaged could lead, if not detected, to the failure of the secondary load path, which would likely result in loss of control of the aeroplane.

\* \* \* \* \*

We are issuing this AD to require actions to correct the unsafe condition on these products.

**DATES:** This AD becomes effective August 22, 2011.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of August 22, 2011.

**ADDRESSES:** You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, Washington 98057-3356; telephone (425) 227-2125; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:**

**Discussion**

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on April 8, 2011 (76 FR 19724). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

A specific failure case of the THSA [trimmable horizontal stabilizer actuator] primary attachment, which may result in a loading of the upper secondary attachment, has been identified by analysis.

Primary load path failure can be caused by bearing migration from the upper attachment gimbal by failure or loss of a retention bolt.

In case of failure of the THSA upper primary attachment, the THSA upper secondary attachment would engage. Because the upper attachment secondary load path can only withstand the loads for a limited period of time, the condition where it would be engaged could lead, if not detected, to the failure of the secondary load path, which would likely result in loss of control of the aeroplane.

For the reasons explained above, this [EASA] AD requires installation of three secondary retention plates for the gimbal bearings on the THSA upper primary attachment.

You may obtain further information by examining the MCAI in the AD docket.

**Comments**

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

**Conclusion**

We reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

**Differences Between This AD and the MCAI or Service Information**

We have reviewed the MCAI and related service information and, in

general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have required different actions in this AD from those in the MCAI in order to follow our FAA policies. Any such differences are highlighted in a Note within the AD.

**Costs of Compliance**

We estimate that this AD will affect 215 products of U.S. registry. We also estimate that it will take about 4 work-hours per product to comply with the basic requirements of this AD. The average labor rate is \$85 per work-hour. Required parts will cost about \$3,021 per product. Where the service information lists required parts costs that are covered under warranty, we have assumed that there will be no charge for these parts. As we do not control warranty coverage for affected parties, some parties may incur costs higher than estimated here. Based on these figures, we estimate the cost of this AD to the U.S. operators to be \$722,615, or \$3,361 per product.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

**Regulatory Findings**

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and

responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD docket.

**Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains the NPRM, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone (800) 647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

- 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

- 2. The FAA amends § 39.13 by adding the following new AD:

**2011-15-08 Airbus:** Amendment 39-16755. Docket No. FAA-2011-0309; Directorate Identifier 2010-NM-255-AD.

**Effective Date**

- (a) This airworthiness directive (AD) becomes effective August 22, 2011.

**Affected ADs**

- (b) None.

**Applicability**

- (c) This AD applies to all Airbus Model A300 B4-601, B4-603, B4-620, and B4-622 airplanes, Model A300 B4-605R and B4-622R airplanes, Model A300 F4-605R and F4-622R airplanes, and Model A300 C4-

605R Variant F airplanes; and Model A310–203, –204, –221, –222, –304, –322, –324, and –325 airplanes; certificated in any category.

#### Subject

(d) Air Transport Association (ATA) of America Code 27: Flight controls.

#### Reason

(e) The mandatory continuing airworthiness information (MCAI) states:

A specific failure case of the THSA [trimmable horizontal stabilizer actuator] upper primary attachment, which may result in a loading of the upper secondary attachment, has been identified by analysis.

Primary load path failure can be caused by bearing migration from the upper attachment gimbal by failure or loss of a retention bolt.

In case of failure of the THSA upper primary attachment, the THSA upper secondary attachment would engage. Because the upper attachment secondary load path can only withstand the loads for a limited period of time, the condition where it would be engaged could lead, if not detected, to the failure of the secondary load path, which would likely result in loss of control of the aeroplane.

\* \* \* \* \*

#### Compliance

(f) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

#### Installation

(g) Within 30 months after the effective date of this AD, install three retention plates on the THSA upper primary attachment, in accordance with the Accomplishment Instructions of Airbus Mandatory Service Bulletin A300–27–6066 (for Model A300–600 series airplanes) or Airbus Mandatory Service Bulletin A310–27–2103 (for Model A310 series airplanes), both dated June 10, 2010.

#### FAA AD Differences

**Note 1:** This AD differs from the MCAI and/or service information as follows: No differences.

#### Other FAA AD Provisions

(h) The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Dan Rodina, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, Washington 98057–3356; telephone (425) 227–2125; fax (425) 227–1149. Information may be e-mailed to: 9–ANM–116–AMOC–REQUESTS@faa.gov. Before using any approved AMOC, notify

your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) Airworthy Product: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

#### Related Information

(i) Refer to MCAI European Aviation Safety Agency (EASA) Airworthiness Directive 2010–0224, dated November 4, 2010; and Airbus Mandatory Service Bulletins A300–27–6066 and A310–27–2103, both dated June 10, 2010.

#### Material Incorporated by Reference

(j) You must use Airbus Mandatory Service Bulletin A300–27–6066, dated June 10, 2010; or Airbus Mandatory Service Bulletin A310–27–2103, dated June 10, 2010; as applicable; to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Airbus SAS—EAW (Airworthiness Office), 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; e-mail [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet <http://www.airbus.com>.

(3) You may review copies of the service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington. For information on the availability of this material at the FAA, call 425–227–1221.

(4) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

Issued in Renton, Washington, on July 6, 2011.

**Kalene C. Yanamura,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2011–17698 Filed 7–15–11; 8:45 am]

**BILLING CODE 4910–13–P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA–2011–0308; Directorate Identifier 2010–NM–233–AD; Amendment 39–16754; AD 2011–15–07]

RIN 2120–AA64

#### Airworthiness Directives; 328 Support Services GmbH (Type Certificate Previously Held by AvCraft Aerospace GmbH; Fairchild Dornier GmbH; Dornier Luftfahrt GmbH) Model 328–100 and –300 Airplanes

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

During maintenance, it has been discovered that at the installation of the fixation brackets for rudder spring tabs and trim tabs an incorrect installation of the fixation brackets may have occurred. \* \* \*

If the orientation of the fixation bracket is reversed or upside down the screws may not reach into the helicoil thread to a sufficient depth.

An incorrect installation, if not detected and corrected, could lead to an in-flight failure of the fixation brackets for rudder spring tabs and trim tabs resulting in and reduced control of the aeroplane.

\* \* \* \* \*

We are issuing this AD to require actions to correct the unsafe condition on these products.

**DATES:** This AD becomes effective August 22, 2011.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of August 22, 2011.

**ADDRESSES:** You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue, SE., Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Tom Rodriguez, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA,