

model series of airplanes. It is not a rule of general applicability.

The substance of these special conditions has been subjected to the notice and comment period in several prior instances and has been derived without substantive change from those previously issued. It is unlikely that prior public comment would result in a significant change from the substance contained herein. Therefore, the FAA has determined that prior public notice and comment are unnecessary, and good cause exists for adopting these special conditions upon publication in the **Federal Register**. The FAA is requesting comments to allow interested persons to submit views that may not have been submitted in response to the prior opportunities for comment described above.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

■ The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

The Special Conditions

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for Bombardier Model BD-700-2A12 and BD-700-2A13 airplanes.

1. In addition to meeting the requirements of § 25.773(b)(1), the airplane must have a means to maintain a clear portion of the windshield, during precipitation conditions, such that both pilots to have a sufficiently extensive view along the ground or flight path in taxi and flight. This means must be designed to function, without continuous attention on the part of the crew, in conditions from light misting precipitation to heavy rain, at speeds from fully stopped in still air to 1.5 V_{SR1} with lift and drag devices retracted, and in icing conditions specified in § 25.1419 if certification for flight in icing conditions is requested.

2. The precipitation removal system must comply with § 25.773.

3. Instructions to maintain the precipitation-removal system must comply with § 25.1529.

4. The materials used in the precipitation removal system must comply with § 25.603.

Issued in Renton, Washington, on August 23, 2016.

John P. Piccola, Jr.,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016-21079 Filed 8-31-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2016-7048; Directorate Identifier 2016-CE-014-AD; Amendment 39-18635; AD 2016-18-05]

RIN 2120-AA64

Airworthiness Directives; PILATUS AIRCRAFT LTD. Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for PILATUS AIRCRAFT LTD. Models PC-12, PC-12/45, PC-12/47, and PC-12/47E airplanes installed with an affected engine mounting frame assembly. This AD results from mandatory continuing airworthiness information (MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as longitudinal material separation on the internal surface of the engine mounting frame assembly tubes. We are issuing this AD to detect and correct this condition, which could lead to partial or complete failure of the structural joint and possibly result in in-flight detachment of the engine with consequent loss of control.

DATES: This AD is effective October 6, 2016.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the AD as of October 6, 2016.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-7048; or in person at Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

For service information identified in this AD, contact Pilatus Aircraft Ltd., Customer Support PC-12, CH-6371 Stans, Switzerland; phone: +41 41 619

33 33; fax: +41 41 619 73 11; email: SupportPC12@pilatus-aircraft.com; Internet: www.pilatus-aircraft.com. For information on the availability of this material at the FAA, call (816) 329-4148. It is also available on the Internet at <http://www.regulations.gov> by searching for Docket No. FAA-2016-7048.

FOR FURTHER INFORMATION CONTACT:

Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; fax: (816) 329-4090; email: doug.rudolph@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to PILATUS AIRCRAFT LTD. Models PC-12, PC-12/45, PC-12/47, and PC-12/47E airplanes installed with an affected engine mounting frame assembly. The NPRM was published in the **Federal Register** on June 13, 2016 (81 FR 38115). The NPRM proposed to correct an unsafe condition for the specified products and was based on mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country. The MCAI states:

The PC-12 Engine Mounting Frame Assembly (hereafter referred to as "EMF" in this AD), Part Number (P/N) 571.20.12.036, is a welded structure including three special tubes, P/N 571.20.12.073, P/N 571.20.12.074 and P/N 571.20.12.107, the ends of which are subject to a special swaging process during manufacturing. Longitudinal material separation on the internal surface of the special tubes was detected on few EMFs on new production aeroplanes. Investigations identified the root cause to be an incorrect accomplishment of the swaging process.

This condition, if not detected and corrected, could lead to growth of the material separation and subsequent partial or complete failure of the structural joint, possibly resulting in in-flight detachment of the engine and consequent reduced control, or loss of control, of the aeroplane.

To address this potential unsafe condition, Pilatus issued Service Bulletin (SB) No. 71-009, now at Revision 2 (hereafter referred to as "the SB" in this AD), to provide inspection instructions for the affected EMF to detect indications of material separation.

For the reason described above, this AD requires identification and inspection of the affected EMF and, depending on the findings, their replacement with serviceable EMF.

The MCAI can be found in the AD docket on the Internet at: <https://www.regulations.gov/document?D=FAA-2016-7048-0002>.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM (81 FR 38115, June 13, 2016) or on the determination of the cost to the public.

Conclusion

We reviewed the relevant data and determined that air safety and the public interest require adopting the AD as proposed except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM (81 FR 38115, June 13, 2016) for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM (81 FR 38115, June 13, 2016).

Related Service Information Under 1 CFR Part 51

We reviewed PILATUS AIRCRAFT LTD. PILATUS PC-12 Service Bulletin No: 71-009, Reference No: 345, Modification No: EC-15-0632, Revision 2, dated March 18, 2016; Pilatus Powerplant Mounting Frame, Removal/Installation, Date module/Technical publication 12-A-71-00-05-00A-920A-A, dated February 26, 2010, found in Pilatus Model type-PC-12, PC-12/45, PC-12/47 MSN-101-888 Aircraft Maintenance Manual (AMM), Document No. 02049, 12-A-AM-00-00-00-I; and Pilatus Powerplant Mounting Frame, Removal/Installation, Date module/Technical publication 12-B-71-00-05-00A-920A-A, dated October 4, 2010, found in Pilatus Model type-PC-12/47E MSN-1001-UP Aircraft Maintenance Manual (AMM), Document No. 02300, 12-B-AM-00-00-00-I. The service information describes procedures for determining if an affected engine mounting frame assembly (EMF) is installed, inspecting the EMF, and replacing the EMF with a serviceable EMF. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section of this AD.

Costs of Compliance

We estimate that this AD will affect 888 products of U.S. registry. We also estimate that it would take about 3 work-hours per product to comply with the basic requirements of this AD. The average labor rate is \$85 per work-hour.

Based on these figures, we estimate the cost of this AD on U.S. operators to be \$226,440, or \$255 per product.

In addition, we estimate that any necessary follow-on actions would cost the following amounts. We have no way of determining the number of products that may need these actions.

The visual and eddy current inspections would take about 3 work-hours for a cost of \$255 per product.

The replacement of the EMF would take about 90 work-hours and require parts costing \$33,336, for a cost of \$40,986 per product.

According to the manufacturer, some of the costs of this AD may be covered under warranty, thereby reducing the cost impact on affected individuals. We do not control warranty coverage for affected individuals. As a result, we have included all costs in our cost estimate.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities

under the criteria of the Regulatory Flexibility Act.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-7048; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains the NPRM, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone (800) 647-5527) is in the ADDRESSES section. Comments will be available in the AD docket shortly after receipt.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new AD:

2016-18-05 PILATUS AIRCRAFT LTD.:
Amendment 39-18635; Docket No. FAA-2016-7048; Directorate Identifier 2016-CE-014-AD.

(a) Effective Date

This airworthiness directive (AD) becomes effective October 6, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to PILATUS AIRCRAFT LTD. Models PC-12, PC-12/45, PC-12/47, and PC-12/47E airplanes, all serial numbers, that are:

- (1) Installed with an affected serial number engine mounting frame assembly (EMF), part number (P/N) 571.20.12.036, listed in figure 1 of paragraph (c)(1) of this AD; and

FIGURE 1 TO PARAGRAPH (c)(1) OF THIS AD: EMF P/N 571.20.12.036, AFFECTED SERIAL NUMBERS

0001 through 1200 inclusive.
1202 through 1272 inclusive.
1275 through 1323 inclusive.
1325 through 1328 inclusive.

FIGURE 1 TO PARAGRAPH (c)(1) OF THIS AD: EMF P/N 571.20.12.036, AFFECTED SERIAL NUMBERS—Continued

1334 through 1338 inclusive.
1340 and 1342.
1344 through 1346 inclusive.
1348 and 1349.
1358, 1361, and 1365.

(2) Certificated in any category.

(d) Subject

Air Transport Association of America (ATA) Code 71: Power Plant.

(e) Reason

This AD was prompted by mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as longitudinal material separation on the internal surface of the engine mounting frame assembly tubes (EMF). We are issuing this AD to detect and correct material separation on the internal surface of the engine mounting frame assembly tubes, which could lead to partial or complete failure of the structural joint and possibly result in in-flight detachment of the engine with consequent loss of control.

(f) Actions and Compliance

Do the actions in paragraphs (f)(1) through (7) of this AD. If paragraphs (f)(1) through (6) of this AD have already been done before October 6, 2016 (the effective date of this AD), then only paragraph (f)(7) of this AD applies.

(1) Within the compliance time identified in figure 2 of paragraph (f)(1) of this AD, do an ultrasonic inspection of the swaged engine mounting tube ends of the affected EMF following the instructions of paragraph 3.B.(1) of PILATUS AIRCRAFT LTD PILATUS PC-12 Service Bulletin No: 71-009, Reference No: 345, Modification No: EC-15-0632, Revision 2, dated March 18, 2016.

FIGURE 2 TO PARAGRAPH (f)(1) OF THIS AD: INITIAL COMPLIANCE TIME

A or B, Whichever Occurs Later

A	Before the EMF exceeds 11,000 hours time-in-service (TIS) or 13,500 flight cycles (FC), whichever occurs first since first installation of the EMF on an airplane.
B	Within 1,000 hours TIS or 1,000 FC or 6 months, whichever occurs first after October 6, 2016 (the effective date of this AD).

(2) If an indication with an echo of less than 40 percent full screen height is detected on an EMF during the ultrasonic inspection required in paragraph (f)(1) of this AD, except for paragraph (f)(7), no further actions are required for this AD. Document compliance with this AD in the maintenance records.

(3) If an indication with an echo of 40 percent full screen height or more is detected on an EMF during the ultrasonic inspection required in paragraph (f)(1) of this AD, do the actions in paragraphs (f)(3)(i) through (iii) of this AD, as applicable.

(i) Before further flight and repetitively thereafter at intervals not to exceed 600 hours TIS or 12 months, whichever occurs first, do a visual inspection of the welding and do an eddy current inspection of the tubes at the indication point detected during the ultrasonic inspection. Use the instructions of paragraphs 3.B.(2) and 3.B.(3) of PILATUS AIRCRAFT LTD PILATUS PC-12 Service Bulletin No: 71-009, Reference No: 345, Modification No: EC-15-0632, Revision 2, dated March 18, 2016.

(ii) If any cracks are found during any of the visual inspections or if an indication with a signal of 20 percent or more is detected during any of the eddy current inspections required in paragraph (f)(3)(i) of this AD, before further flight, replace the EMF with a serviceable EMF following the instructions in the service information listed in paragraph (f)(5) of this AD, including all subparagraphs as applicable.

(iii) Unless already done as required by paragraph (f)(3)(ii) of this AD, with ≤ 1,800 hours TIS or 36 months after the initial visual and eddy current inspections of the affected EMF required by paragraph (f)(3)(i) of this AD, whichever occurs first, replace the EMF with a serviceable EMF following the instructions in the service information listed in paragraph (f)(5) of this AD, including all subparagraphs as applicable.

(4) For the purpose of this AD, a serviceable EMF is defined as any EMF that is not listed in figure 1 of paragraph (c)(1) of this AD or an affected EMF that is listed in figure 1 of paragraph (c)(1) of this AD but has had the ultrasonic inspection required in paragraph (f)(1) of this AD and had an indication with an echo of less than 40 percent full screen height.

(5) For replacement of the EMF, follow the instructions listed in paragraphs (f)(5)(i) and (ii), as applicable.

(i) *For Models PC-12, PC-12/45, and PC-12/47, manufacturer serial numbers (MSN) 101-888: Pilatus Powerplant Mounting Frame, Removal/Installation, Date module/Technical publication 12-A-71-00-05-00A-920A-A, dated February 26, 2010, found in Pilatus Model type-PC-12, PC-12/45, PC-12/47 MSN-101-888 Aircraft Maintenance Manual (AMM), Document No. 02049, 12-A-AM-00-00-00-I.*

(ii) *For Model PC-12/47E, MSN 1001 and up: Pilatus Powerplant Mounting Frame, Removal/Installation, Date module/Technical publication 12-B-71-00-05-00A-920A-A, dated October 4, 2010, found in Pilatus Model type-PC-12/47E MSN-1001-UP Aircraft Maintenance Manual (AMM), Document No. 02300, 12-B-AM-00-00-00-I.*

(6) If an EMF has an indication with an echo of 40 percent or more during the ultrasonic inspection required in paragraph (f)(1) of this AD, you may replace the EMF with a serviceable EMF in lieu of the visual or eddy current inspections required in paragraph (f)(3)(i) of this AD. For

replacement of the EMF, follow the instructions in the service information listed in paragraph (f)(5) of this AD, including all subparagraphs as applicable.

(7) As of October 6, 2016 (the effective date of this AD), do not install an EMF P/N 571.20.12.036 unless it has been determined to be a serviceable EMF as specified in paragraph (f)(4) of this AD.

(8) Airplanes with an MSN of 1556 or higher are not affected by this AD provided that the EMF has not been replaced since its manufacture. A review of the maintenance records, Airworthiness Approval Tag (FAA Form 8130-3), or other positive form of parts identification such as a shipping ticket, invoice, or direct ship authority letter, can be used to determine the serial number of the EMF.

(g) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, Standards Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Doug Rudolph, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4059; fax: (816) 329-4090; email: doug.rudolph@faa.gov. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) *Airworthy Product:* For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(h) Related Information

(1) Refer to MCAI European Aviation Safety Agency (EASA) AD No. 2016-0081, dated April 25, 2016, for related information pertaining to this AD. The MCAI can be found in the AD docket on the Internet at: <https://www.regulations.gov/document?D=FAA-2016-7048-0002>.

(i) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) PILATUS PC-12 Service Bulletin No: 71-009, Reference No: 345, Modification No: EC-15-0632, Revision 2, dated March 18, 2016;

(ii) Pilatus Powerplant Mounting Frame, Removal/Installation, Date module/Technical publication 12-A-71-00-05-00A-920A-A, dated February 26, 2010, found in Pilatus Model type-PC-12, PC-12/45, PC-12/47 MSN-101-888 Aircraft Maintenance Manual

(AMM), Document No. 02049, 12-A-AM-00-00-00-I; and

(iii) Pilatus Powerplant Mounting Frame, Removal/Installation, Date module/Technical publication 12-B-71-00-05-00A-920A-A, dated October 4, 2010, found in Pilatus Model type- PC-12/47E MSN-1001-UP Aircraft Maintenance Manual (AMM), Document No. 02300, 12-B-AM-00-00-00-I.

(3) For service information identified in this AD, contact Pilatus Aircraft Ltd., Customer Support PC-12, CH-6371 Stans, Switzerland; phone: +41 41 619 33 33; fax: +41 41 619 73 11; email: SupportPC12@pilatus-aircraft.com; Internet: www.pilatus-aircraft.com.

(4) You may view this service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148. In addition, you can access this service information on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-7048.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Kansas City, Missouri, on August 23, 2016.

David R. Showers,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016-20833 Filed 8-31-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2016-5460; Directorate Identifier 2015-NM-188-AD; Amendment 39-18599; AD 2016-16-01]

RIN 2120-AA64

Airworthiness Directives; Airbus Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule; correction.

SUMMARY: The FAA is correcting an airworthiness directive (AD) that published in the **Federal Register**. That AD applies to certain Airbus Model A330-200 Freighter, -200, and -300 series airplanes. Paragraphs (i) and (l) of the regulatory text contain typographical errors in the service bulletin number. This document corrects those errors. In all other respects, the original document remains the same.

DATES: This final rule is effective September 8, 2016.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of September 8, 2016 (81 FR 51325, August 4, 2016).

ADDRESSES: For service information identified in this final rule, contact Airbus SAS, Airworthiness Office-EAL, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone: +33 5 61 93 36 96; fax: +33 5 61 93 45 80; email: airworthiness.A330-A340@airbus.com; Internet: <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-5460.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT:

Vladimir Ulyanov, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone: 425-227-1138; fax: 425-227-1149.

SUPPLEMENTARY INFORMATION:

Airworthiness Directive 2016-16-01, Amendment 39-18599 (81 FR 51325, August 4, 2016) ("AD 2016-16-01"), currently requires an inspection of affected structural parts in the cargo and cabin compartments to determine if proper heat treatment has been done, and replacement if necessary, for certain Airbus Model A330-200 Freighter, -200, and -300 series airplanes.

Need for the Correction

As published, paragraphs (i) and (l) of the regulatory text identify the service information by the wrong service bulletin number. Where paragraphs (i) and (l) incorrectly specify Airbus

Service Bulletins "A320-53-3227" and "A320-53-3228," the correct service bulletin numbers are "A330-53-3227" and "A330-53-3228," respectively.

Related Service Information Under 14 CFR Part 51

Airbus has issued the following service information:

- Airbus Service Bulletin A330-53-3227, dated August 18, 2015. The service information describes procedures to inspect affected structural parts in the cargo compartment to determine if proper heat treatment has been done, and replacement of parts; and
- Airbus Service Bulletin A330-53-3228, dated August 18, 2015. The service information describes procedures to inspect affected structural parts in the cabin compartment to determine if proper heat treatment has been done, and replacement of parts.

This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

Correction of Publication

This document corrects two errors and correctly adds the AD as an amendment to 14 CFR 39.13. Although no other part of the preamble or regulatory information has been corrected, we are publishing the entire rule in the **Federal Register**.

The effective date of this AD remains September 8, 2016.

Since this action only corrects a typographical error in two locations, it has no adverse economic impact and imposes no additional burden on any person. Therefore, we have determined that notice and public procedures are unnecessary.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Correction

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.