

into consideration the positioning of flight attendants throughout the main passenger compartment during various phases of flight.

11. The crew rest compartment must be designed such that fires within the compartment can be controlled without a crewmember having to enter the compartment, or the design of the access provisions must allow crewmembers equipped for firefighting to have unrestricted access to the compartment. The time for a crewmember on the main deck to react to the fire alarm, to don the fire fighting equipment, and to gain access must not exceed the time for the compartment to become smoke-filled, making it difficult to locate the fire source.

12. There must be a means provided to exclude hazardous quantities of smoke or extinguishing agent originating in the crew rest compartment from entering any other compartment occupied by crewmembers or passengers. The means must include the time periods during the evacuation of the crew rest compartment and, if applicable, when accessing the crew rest compartment to manually fight a fire. Smoke entering any other compartment occupied by crewmembers or passengers must dissipate within 5 minutes after closing the access to the crew rest compartment. Flight tests must be conducted to show compliance with this requirement.

13. There must be a supplemental oxygen system equivalent to that provided for main deck passengers for each seat and berth in the crew rest compartment. The system must provide:

(a) An aural and visual warning to the occupants of the crew rest compartment to don oxygen masks in the event of decompression; and

(b) A decompression warning that activates before the cabin pressure altitude exceeds 15,000 feet. The warning must sound continuously until a reset pushbutton in the crew rest compartment is depressed.

14. The following requirements apply to a crew rest compartment that is divided into several sections by the installation of curtains or partitions:

(a) To compensate for sleeping occupants, there must be an aural alert that can be heard in each section of the crew rest compartment that accompanies automatic presentation of supplemental oxygen masks. Two supplemental oxygen masks are required in each section whether or not seats or berths are installed in each section. There must also be a means by which the oxygen masks can be manually deployed from the flight deck.

(b) A placard is required adjacent to each curtain that visually divides or separates, for privacy purposes, the overhead crew rest compartment into small sections. The placard must require that the curtain(s) remain open when the private section it creates is unoccupied. The vestibule section adjacent to the stairway is not considered a private area and, therefore, does not require a placard.

(c) For each crew rest section created by the installation of a curtain, the following requirements of these special conditions must be met with the curtain open or closed:

(1) No smoking placard (special condition no. 1),

(2) Emergency illumination (special condition no. 5),

(3) Emergency alarm system (special condition no. 7),

(4) Seat belt fasten signal (special condition no. 8), and

(5) The smoke or fire detection system (special conditions no.'s 10, 11, and 12).

(d) Overhead crew rest compartments visually divided to the extent that evacuation could be affected must have exit signs that direct occupants to the primary stairway exit. The exit signs must be provided in each separate section of the crew rest compartment, and must meet the requirements of § 25.812(b)(1)(i).

(e) For sections within an overhead crew rest compartment that are created by the installation of a rigid partition with a door physically separating the sections, the following requirements of these special conditions must be met with the door open or closed:

(1) There must be a secondary evacuation route from each section to the main deck, or alternatively, it must be shown that any door between the sections has been designed to preclude anyone from being trapped inside the compartment.

(2) Any door between the sections must be shown to be openable when crowded against, even when crowding occurs at each side of the door.

(3) There may be no more than one door between any seat or berth and the primary stairway exit.

(4) There must be exit signs in each section meeting the requirements of § 25.812(b)(1)(i) that direct occupants to the primary stairway exit.

(f) For each smaller section within the main crew rest compartment created by the installation of a partition with a door, the following requirements of these special conditions must be met with the door open or closed:

(1) No smoking placards (special condition no. 1),

(2) Emergency illumination (special condition no. 5),

(3) Two-way voice communication (special condition no. 6),

(4) Emergency alarm system (special condition no. 7),

(5) Seat belt fasten signal (special condition no. 8),

(6) Emergency fire fighting and protective equipment (special condition no. 9), and

(7) Smoke or fire detection system (special conditions no.'s 10, 11, and 12).

15. The requirements of two-way voice communication with the flight deck and provisions for emergency firefighting and protective equipment are not applicable to lavatories or other small areas that are not intended to be occupied for extended periods of time.

16. Where a waste disposal receptacle is fitted, it must be equipped with an automatic fire extinguisher that meets the performance requirements of § 25.854(b).

17. Materials (including finishes or decorative surfaces applied to the materials) must comply with the flammability requirements of § 25.853(a), as amended by Amendment 25-83. Mattresses must comply with the flammability requirements of § 25.853(c), as amended by Amendment 25-83.

Issued in Renton, Washington on December 1, 2000.

**Donald L. Riggins,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service, ANM-100.*

[FR Doc. 00-31478 Filed 12-8-00; 8:45 am]

**BILLING CODE 4910-13-U**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2000-NM-03-AD; Amendment 39-12032; AD 2000-24-25]

RIN 2120-AA64

#### **Airworthiness Directives; Raytheon Model Hawker 800A (U-125A) and Hawker 800XP Series Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to certain Raytheon Model Hawker 800A (U-125A) and Hawker 800XP series airplanes, that requires inspecting the roller clearance in the nose landing gear drag stay and making

any necessary adjustments. This amendment is prompted by reports indicating multiple findings of roller clearances that are in excess of specifications. The actions specified by this AD are intended to prevent the inability to extend the nose landing gear, which could result in damage to the airplane upon landing.

**DATES:** Effective January 16, 2001.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of January 16, 2001.

**ADDRESSES:** The service information referenced in this AD may be obtained from Raytheon Aircraft Company, 9709 East Central, Wichita, Kansas 67206. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Mid-Continent Airport, Wichita, Kansas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Paul C. DeVore, Aerospace Engineer, Systems and Propulsion Branch, ACE-116W, FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Mid-Continent Airport, Wichita, Kansas 67209; telephone (316) 946-4142; fax (316) 946-4407.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Raytheon Model Hawker 800A (U-125A) and Hawker 800XP series airplanes was published in the **Federal Register** on August 8, 2000 (65 FR 48402). That action proposed to require inspecting the roller clearance in the nose landing gear drag stay and making any necessary adjustments.

#### Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

#### Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

#### Cost Impact

There are approximately 85 airplanes of the affected design in the worldwide

fleet. The FAA estimates that 50 airplanes of U.S. registry will be affected by this AD, that it will take approximately 7 work hours per airplane to accomplish the required inspection and any necessary adjustments, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$21,000, or \$420 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

#### Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

## PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

### 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

#### 2000-24-25 Raytheon Aircraft Company:

Amendment 39-12032. Docket 2000-NM-03-AD.

*Applicability:* Model Hawker 800XP and Hawker 800A (U-125A) series airplanes, as specified in Raytheon Aircraft Service Bulletin SB 32-3274, dated August 1999; certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

*Compliance:* Required as indicated, unless accomplished previously.

To prevent the inability to extend the nose landing gear due to excessive clearance of the roller in the drag stay rigging, which could result in damage to the airplane upon landing, accomplish the following:

#### Inspection and Adjustment

(a) Within 50 hours time-in-service after the effective date of this AD: Remove the paint from the drag stay arm of the nose landing gear at its point of contact with the stop bolt, do a check of the roller clearances, and make any necessary adjustments, in accordance with the Accomplishment Instructions of Raytheon Aircraft Service Bulletin SB 32-3274, dated August 1999.

(b) Airplanes which have had the 600-hour inspection specified in the Aircraft Maintenance Manual before the effective date of this AD or which will have the 600-hour inspection within 50 hours time-in-service after the effective date of this AD are considered to be in compliance with paragraph (a) of this AD.

#### Alternative Methods of Compliance

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Wichita Aircraft Certification Office (ACO), FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Wichita ACO.

**Note 2:** Information concerning the existence of approved alternative methods of

compliance with this AD, if any, may be obtained from the Wichita ACO.

**Special Flight Permits**

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

**Incorporation by Reference**

(e) The actions shall be done in accordance with Raytheon Aircraft Service Bulletin SB 32-3274, dated August 1999. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Raytheon Aircraft Company, 9709 East Central, Wichita, Kansas 67206. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Mid-Continent Airport, Wichita, Kansas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Effective Date**

(f) This amendment becomes effective on January 16, 2001.

Issued in Renton, Washington, on November 29, 2000.

**Donald L. Riggins,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*  
[FR Doc. 00-30946 Filed 12-8-00; 8:45 am]  
**BILLING CODE 4910-13-U**

**DEPARTMENT OF TRANSPORTATION**

**Federal Aviation Administration**

**14 CFR Part 39**

[Docket No. 2000-CE-49-AD; Amendment 39-12030; AD 2000-24-23]

**RIN 2120-AA64**

**Airworthiness Directives; S.N. CENTRAIR 101 Series Gliders**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD) that applies to certain S.N. CENTRAIR 101 series gliders. This AD requires you to

inspect the airbrake control system for cracks; and if cracks are detected, replace the airbrake control system. This AD is the result of mandatory continuing airworthiness information (MCAI) issued by the airworthiness authority for France. The actions specified in this AD are intended to detect cracks in the airbrake control system and replace cracked parts with parts of improved design. A crack in the airbrake control system could prevent the pilot from using the airbrake system.

**DATES:** This AD becomes effective on January 27, 2001.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the regulations as of January 27, 2001.

**ADDRESSES:** You may get the service information referenced in this AD from S.N. CENTRAIR, Aerodome—36300 Le Blanc, France; telephone: 02.54.37.07.96; facsimile: 02.54.37.48.64. You may examine this information at the Federal Aviation Administration (FAA), Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2000-CE-49-AD, 901 Locust, Room 506, Kansas City, Missouri 64106; or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Mike Kiesov, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4144; facsimile: (816) 329-4090.

**SUPPLEMENTARY INFORMATION:**

**Discussion**

*What Events Have Caused This AD?*

The Direction Generale de l'Aviation Civile (DGAC), which is the airworthiness authority for France, recently told the FAA that an unsafe condition may exist on certain S.N. CENTRAIR 101 series gliders. The DGAC reports that a failure analysis of the welded parts of airbrake arms revealed that cracks could occur in these parts.

*What Happens If You Do Not Correct the Condition?*

This condition, if not corrected, could result in undetected cracks.

Consequently, a crack in the airbrake control system could prevent the pilot from using the airbrake system.

*Has FAA Taken Any Action to This Point?*

We issued a proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD that would apply to certain S.N. CENTRAIR 101 series gliders. This proposal was published in the **Federal Register** as a notice of proposed rulemaking (NPRM) on September 14, 2000 (65 FR 55466). The NPRM proposed to require you to inspect the airbrake control system for cracks and if cracks are detected, replace the airbrake control system.

*Was the Public Invited To Comment?*

Interested persons were afforded an opportunity to participate in the making of this amendment. No comments were received on the proposed rule or the FAA's determination of the cost to the public.

**The FAA's Determination**

*What Is FAA's Final Determination on This Issue?*

After careful review of all available information related to the subject presented above, we have determined that air safety and the public interest require the adoption of the rule as proposed except for minor editorial corrections. We determined that these minor corrections:

- Will not change the meaning of the AD; and
- Will not add any additional burden upon the public than was already proposed.

**Cost Impact**

*How Many Gliders Does This AD Impact?*

We estimate that this AD affects 41 gliders in the U.S. registry.

*What Is the Cost Impact of This AD on Owners/Operators of the Affected Gliders?*

We estimate the following costs to accomplish the inspection:

Labor cost	Parts cost	Total cost per glider	Total cost on U.S. operators
2 workhours × \$60 per hour = \$120.	No parts required for the inspection.	\$120 per glider .....	\$120 × 41 = \$4,920.

We estimate the following costs to accomplish the replacement of the airbrake control system if necessary: