

described in Special Condition No. 4(a) may be used to meet this requirement.

(f) For each smaller section within the main overhead crew rest compartment created by the installation of a partition with a door, the following requirements of these special conditions must be met with the door open or closed:

(1) No smoking placards (Special Condition No. 1);

(2) Emergency illumination (Special Condition No. 5);

(3) Two-way voice communication (Special Condition No. 6);

(4) Emergency alarm system (Special Condition No. 7);

(5) Seat belt fasten signal or return to seat signal as applicable (Special Condition No. 8);

(6) Emergency firefighting and protective equipment (Special Condition No. 9); and

(7) Smoke or fire detection system (Special Condition No. 10).

15. The requirements of two-way voice communication with the flight deck and provisions for emergency firefighting and protective equipment are not applicable to lavatories or other small areas that are not intended to be occupied for extended periods of time.

16. Where a waste disposal receptacle is fitted, it must be equipped with an automatic fire extinguisher that meets the performance requirements of § 25.854(b).

17. Materials (including finishes or decorative surfaces applied to the materials) must comply with the flammability requirements of § 25.853(a) as amended by Amendment 25–83. Mattresses must comply with the flammability requirements of § 25.853(c), as amended by Amendment 25–83.

18. The addition of a lavatory within the overhead crew rest compartment would require the lavatory to meet the

same requirements as those for a lavatory installed on the main deck except with regard to Special Condition No. 10 for smoke detection.

19. All enclosed stowage compartments within the overhead crew rest compartment that are not limited to stowage of emergency equipment or airplane supplied equipment (*i.e.*, bedding) must meet the design criteria given in the table below. Enclosed stowage compartments greater than 200 ft<sup>3</sup> in interior volume are not addressed by this special condition. The in flight accessibility of very large enclosed stowage compartments and the subsequent impact on the crewmembers' ability to effectively reach any part of the compartment with the contents of a hand fire extinguisher will require additional fire protection considerations similar to those required for inaccessible compartments such as Class C cargo compartments.

| Fire protection features                     | Stowage compartment interior volumes |                                |                                 |
|----------------------------------------------|--------------------------------------|--------------------------------|---------------------------------|
|                                              | Less than 25 cubic feet              | 25 cubic feet to 57 cubic feet | 57 cubic feet to 200 cubic feet |
| Materials of Construction <sup>1</sup> ..... | Yes .....                            | Yes .....                      | Yes.                            |
| Detectors <sup>2</sup> .....                 | No .....                             | Yes .....                      | Yes.                            |
| Liner <sup>3</sup> .....                     | No .....                             | Yes .....                      | Yes.                            |
| Locating Device <sup>4</sup> .....           | No .....                             | Yes .....                      | Yes.                            |

<sup>1</sup> *Material:* The material used to construct each enclosed stowage compartment must at least be fire resistant and must meet the flammability standards established for interior components (*i.e.*, 14 CFR part 25 Appendix F, parts I, IV, and V) per the requirements of § 25.853. For compartments less than 25 ft<sup>3</sup> in interior volume, the design must ensure the ability to contain a fire likely to occur within the compartment under normal use.

<sup>2</sup> *Detectors:* Enclosed stowage compartments equal to or exceeding 25 ft<sup>3</sup> in interior volume must be provided with a smoke or fire detection system to ensure that a fire can be detected within a one-minute detection time. Flight tests must be conducted to show compliance with this requirement. Each system (or systems) must provide: (a) A visual indication in the flight deck within one minute after the start of a fire, (b) An aural warning in the overhead crew rest compartment, and (c) A warning in the main passenger cabin. This warning must be readily detectable by a flight attendant, taking into consideration the positioning of flight attendants throughout the main passenger compartment during various phases of flight.

<sup>3</sup> *Liner:* If it can be shown that the material used to construct the stowage compartment meets the flammability requirements of a liner for a Class B cargo compartment (*i.e.*, § 25.855 at Amendment 25–93, and Appendix F, part I, paragraph (a)(2)(ii)), then no liner would be required for enclosed stowage compartments equal to or greater than 25 ft<sup>3</sup> in interior volume but less than 57 ft<sup>3</sup> in interior volume. For all enclosed stowage compartments equal to or greater than 57 ft<sup>3</sup> in interior volume but less than or equal to 200 ft<sup>3</sup>, a liner must be provided that meets the requirements of § 25.855 for a Class B cargo compartment.

<sup>4</sup> *Location Detector:* Overhead crew rest compartment which contain enclosed stowage compartments exceeding 25 ft<sup>3</sup> interior volume and which are located away from one central location such as the entry to the overhead crew rest compartment or a common area within the overhead crew rest compartment would require additional fire protection features and/or devices to assist the firefighter in determining the location of a fire.

Issued in Renton, Washington on April 9, 2003.

Ali Bahrami,

Acting Manager, Transport Airplane

Directorate, Aircraft Certification Service.

[FR Doc. 03–9505 Filed 4–16–03; 8:45 am]

BILLING CODE 4910–13–P

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2001–NM–99–AD; Amendment 39–13114; AD 2003–08–03]

RIN 2120–AA64

**Airworthiness Directives; McDonnell Douglas Model DC–10–10, DC–10–10F, DC–10–15, DC–10–30, DC–10–30F, DC–10–30F (KC10A and KDC–10), DC–10–40, DC–10–40F, MD–10–10F, and MD–10–30F Airplanes**

AGENCY: Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to certain McDonnell Douglas airplane models. This amendment requires repetitive inspections for chafing or potential chafing of the wiring for the throttle control module (TCM) on the center pedestal in the flight deck compartment, corrective actions if necessary, an inspection of the TCM to determine its part number and configuration, and modification of the TCM. Doing this modification terminates the repetitive inspections. The actions specified by this AD are intended to prevent chafing

of wiring inside the TCM, fuel shutoff lever lights, and/or aft pedestal lightplates due to degradation of protective sleeving, which could result in electrical arcing and failure of the auto throttle/speed control system and consequent smoke and/or fire in the cockpit.

**DATES:** Effective May 22, 2003.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of May 22, 2003.

**ADDRESSES:** The service information referenced in this AD may be obtained from Boeing Commercial Aircraft Group, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024). This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; at the FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Natalie Phan-Tran, Aerospace Engineer, Systems and Equipment Branch, ANM-130L, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712-4137; telephone (562) 627-5343; fax (562) 627-5210.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain McDonnell Douglas Model DC-10-10, DC-10-10F, DC-10-15, DC-10-30, DC-10-30F, DC-10-30F (KC10A and KDC-10), DC-10-40, DC-10-40F, MD-10-10F, and MD-10-30F airplanes, was published as a supplemental notice of proposed rulemaking (NPRM) in the **Federal Register** on January 3, 2003 (68 FR 305). That action proposed to require an inspection of the throttle control module (TCM) on the center pedestal in the flight deck compartment to determine its part number and configuration, modification of the TCM, repetitive inspections for chafing or potential chafing of the TCM wiring, and corrective actions if necessary.

#### Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due

consideration has been given to the comments received.

#### Support for the Proposal

The Air Transport Association of America (ATA) states that its members support the intent of the proposal. In a comment attached to the ATA's comment, one ATA member states that it appreciates the FAA's decision in the supplemental NPRM to extend the compliance time for the proposed actions to 5 years.

#### Conclusion

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

#### Cost Impact

There are approximately 401 Model DC-10-10, DC-10-10F, DC-10-15, DC-10-30, DC-10-30F, DC-10-30F (KC10A and KDC-10), DC-10-40, DC-10-40F, MD-10-10F, and MD-10-30F airplanes of the affected design in the worldwide fleet. The FAA estimates that 321 airplanes of U.S. registry will be affected by this AD.

We estimate that it will take approximately 2 work hours per airplane to perform the required inspections, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the inspections required by this AD on U.S. operators is estimated to be \$38,520, or \$120 per airplane, per inspection cycle.

It will take approximately 15 work hours per airplane to accomplish the required modification at an average labor rate of \$60 per work hour. Required parts will cost approximately \$1,712 per airplane. Based on these figures, the cost impact of the modification required by this AD on U.S. operators is estimated to be \$838,452, or \$2,612 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

#### Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. Section 39.13 is amended by adding the following new airworthiness directive:

#### 2003-08-03 McDonnell Douglas:

Amendment 39-13114. Docket 2001-NM-99-AD.

**Applicability:** Model DC-10-10, DC-10-10F, DC-10-15, DC-10-30, DC-10-30F, DC-10-30F (KC10A and KDC-10), DC-10-40, DC-10-40F, MD-10-10F, and MD-10-30F airplanes; as listed in Boeing Alert Service Bulletin DC10-76A048, Revision 01, dated January 29, 2002; certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or

repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent chafing of wiring inside the throttle control module, fuel shutoff lever lights, and/or aft pedestal lightplates due to degradation of protective sleeving, which could result in electrical arcing and failure of the auto throttle/speed control system and consequent smoke and/or fire in the cockpit, accomplish the following:

#### Repetitive Inspections for Chafing

(a) Within 18 months after the effective date of this AD, perform a general visual inspection for chafing or potential chafing of the wiring of the throttle control module located on the center pedestal in the flight compartment, per Boeing Alert Service Bulletin (ASB) DC10-76A049, excluding the Appendix and Evaluation Form, dated January 29, 2002. Thereafter, repeat the inspection at intervals not to exceed 18 months, until the actions specified in paragraph (c) of this AD are accomplished.

**Note 2:** For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made from within touching distance unless otherwise specified. A mirror may be necessary to enhance visual access to all exposed surfaces in the inspection area. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or droplight and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

#### Corrective Actions for Chafing or Potential Chafing

(b) If any evidence of chafing or potential chafing is found during any inspection required by paragraph (a) of this AD, before further flight, repair the chafed wires or reposition wires, as applicable, per Boeing ASB DC10-76A049, excluding the Appendix and Evaluation Form, dated January 29, 2002.

#### Inspection and Modification

(c) Within 5 years after the effective date of this AD, do the actions specified in paragraphs (c)(1) and (c)(2) of this AD, per Boeing ASB DC10-76A048, excluding the Evaluation Form, dated August 6, 2001; or Revision 01, excluding the Evaluation Form, dated January 29, 2002.

(1) Do an inspection of the throttle control module on the center pedestal in the flight deck compartment to determine its part number and configuration, which will identify the group applicability information.

(2) Modify the throttle control module on the center pedestal in the flight deck

compartment per the applicable figure in the service bulletin. Accomplishment of the modification constitutes terminating action for the requirements of paragraph (a) of this AD.

#### Alternative Methods of Compliance

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Los Angeles Aircraft Certification Office (ACO), FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Los Angeles ACO.

**Note 3:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Los Angeles ACO.

#### Special Flight Permits

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

#### Incorporation by Reference

(f) The actions shall be done in accordance with Boeing Alert Service Bulletin DC10-76A048, excluding the Evaluation Form, dated August 6, 2001, or Boeing Alert Service Bulletin DC10-76A048, Revision 01, excluding the Evaluation Form, dated January 29, 2002; and Boeing Alert Service Bulletin DC10-76A049, excluding the Appendix and Evaluation Form, dated January 29, 2002; as applicable. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Boeing Commercial Aircraft Group, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-0024). Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; at the FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California; or at the Office of the Federal Register, 800 North Capitol Street, NW., Suite 700, Washington, DC.

#### Effective Date

(g) This amendment becomes effective on May 22, 2003.

Issued in Renton, Washington, on April 7, 2003.

**Ali Bahrami,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 03-8894 Filed 4-16-03; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2002-SW-37-AD; Amendment 39-13117; AD 2003-08-06]

**RIN 2120-AA64**

#### Airworthiness Directives; Eurocopter France Model AS350B, B1, B2, BA, and D Helicopters

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD) for the specified Eurocopter model helicopters that requires fireproofing the engine oil tank breather pipe (breather pipe) where it passes through the firewall from the engine compartment to the main gearbox compartment. This amendment is prompted by the discovery of a design deficiency that permitted the installation of a non-fireproof breather pipe. The actions specified by this AD are intended to prevent the spread of fire between two designated fire zones of the helicopter, additional structural damage, and a decrease in the time available to execute an emergency landing.

**DATES:** Effective May 22, 2003.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of May 22, 2003.

**ADDRESSES:** The service information referenced in this AD may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005, telephone (972) 641-3460, fax (972) 641-3527. This information may be examined at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Ed Cuevas, Aviation Safety Engineer, FAA, Rotorcraft Directorate, Regulations Group, Fort Worth, Texas 76193-0111, telephone (817) 222-5355, fax (817) 222-5961.

**SUPPLEMENTARY INFORMATION:** A proposal to amend 14 CFR part 39 to include an AD for Eurocopter France (Eurocopter) Model AS350B, B1, B2, BA, and D helicopters was published in the **Federal Register** on November 4, 2002 (67 FR 67131). That action