

considered, and this rule may be amended in light of the comments received. Factual information that supports the commenter's ideas and suggestions is extremely helpful in evaluating the effectiveness of the AD action and determining whether additional rulemaking action would be needed.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify the rule. All comments submitted will be available in the Rules Docket for examination by interested persons. A report that summarizes each FAA-public contact concerned with the substance of this AD will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their mailed comments submitted in response to this rule must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket No. 2001-SW-08-AD." The postcard will be date stamped and returned to the commenter.

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

The FAA has determined that this regulation is an emergency regulation that must be issued immediately to correct an unsafe condition in aircraft, and that it is not a "significant regulatory action" under Executive Order 12866. It has been determined further that this action involves an emergency regulation under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979). If it is determined that this emergency regulation otherwise would be significant under DOT Regulatory Policies and Procedures, a final regulatory evaluation will be prepared and placed in the Rules Docket. A copy of it, if filed, may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration

amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. FAA amends § 39.13 by adding the following new airworthiness directive:

2001-13-04 Eurocopter France:

Amendment 39-12284. Docket No. 2001-SW-08-AD.

Applicability: Model EC 155B helicopters, certificated in any category.

Note 1: This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required before each flight (takeoff and landing) with an open cabin sliding door, unless accomplished previously.

To prevent in-flight loss of a cabin sliding door, impact with the main rotor or fenestron, and subsequent loss of control of the helicopter, accomplish the following:

(a) Visually check each cabin sliding door in the open position to ensure that each roller is in its rail. If any roller is outside its rail, before further flight, replace the roller inside the rail.

Note 2: Maintenance Manual Task 52-12-00-061 pertains to the subject of this AD.

(b) An owner/operator (pilot) may perform the visual check required by this AD and must record compliance with the visual check required by paragraph (a) of this AD in accordance with 14 CFR 43.11 and 91.417(a)(2)(v)). This AD allows a pilot to perform this check because it involves only a visual check of each cabin sliding door to detect any roller outside its rail and can be performed equally well by a pilot or a mechanic.

(c) This AD revises the Limitations section of the Rotorcraft Flight Manual (RFM) by either inserting statements prohibiting the opening and closing of a cabin sliding door in flight and requiring, before each flight with an open cabin sliding door, visually checking the open door to ensure each door roller is inside its rail, or by inserting a copy of this AD into the Limitations section of the RFM.

(d) An alternative method of compliance or adjustment of the compliance time that

provides an acceptable level of safety may be used if approved by the Manager, Regulations Group, Rotorcraft Directorate, FAA.

Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment, and then send it to the Manager, Rotorcraft Directorate.

(e) A special flight permit is prohibited.

(f) This amendment becomes effective on July 23, 2001.

Note 3: The subject of this AD is addressed in Direction Générale De L'Aviation Civile (France) AD No. T2001-058-001(A), dated February 9, 2001.

Issued Fort Worth, Texas, on June 12, 2001.

Eric Bries,

Acting Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 01-16045 Filed 6-26-01; 8:45 am]

BILLING CODE 4910-13-U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2000-SW-48-AD; Amendment 39-12281; AD 2001-13-01]

RIN 2120-AA64

Airworthiness Directives; Bell Helicopter Textron, Inc. Model 205A-1, 205B, 212, 412, 412EP, and 412CF Helicopters

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), for Bell Helicopter Textron, Inc. (BHTI) Model 205A-1, 205B, 212, 412, 412EP, and 412CF helicopters. This AD requires removing each existing tail rotor counterweight bellcrank (bellcrank) retention nut (retention nut), replacing each retention nut with a zero hours time-in-service (TIS) retention nut; and follow-up inspections of installed retention nuts. This AD is prompted by an in-flight loss of a bellcrank due to failure of the retention nut. The actions specified by this AD are intended to prevent failure of the retention nut, which could result in the bellcrank migrating off the crosshead spindle, loss of tail rotor control, and subsequent loss of control of the helicopter.

DATES: Effective August 1, 2001.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of August 1, 2001.

ADDRESSES: The service information referenced in this AD may be obtained

from Bell Helicopter Textron, Inc., P.O. Box 482, Fort Worth, Texas 76101, telephone (817) 280-3391, fax (817) 280-6466. This information may be examined at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Michael Kohner, Aviation Safety Engineer, FAA, Rotorcraft Directorate, Rotorcraft Certification Office, Fort Worth, Texas 76193-0170, telephone (817) 222-5447, fax (817) 222-5783.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD for BHTI Model 205A-1, 205B, 212, 412, 412EP, and 412CF helicopters was published in the **Federal Register** on March 8, 2001 (66 FR 13858). That action proposed to require the following:

- Removing the two existing retention nuts within 100 hours TIS or 90 days, whichever occurs first;
- Installing a retention nut, part number MS14145L6 or MS17826-6, which is limited to a one-time installation;
- Inspecting the corrosion preventive compound (CPC) coating of the retention nut for deficiencies;
- Inspecting the retention nut for corrosion, mechanical damage, a crack, or looseness; and
- Replacing each retention nut, when necessary.

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were received on the proposal or the FAA's determination of the cost to the public. The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

The FAA estimates that 423 helicopters of U.S. registry will be affected by this AD. It will take approximately 2.5 work hours per helicopter to replace each retention nut and 0.5 work hour to inspect each retention nut once. The average labor rate is \$60 per work hour. Required parts will cost approximately \$7 per helicopter. Based on these figures, the total cost impact of this AD on U.S. operators is estimated to be \$155,241 to replace the retention nuts and inspect them once.

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and

responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding a new airworthiness directive to read as follows:

2001-13-01 Bell Helicopter Textron, Inc.:
Amendment 39-12281. Docket No. 2000-SW-48-AD.

Applicability: Model 205A-1, 205B, 212, 412, 412EP, and 412CF helicopters, certificated in any category.

Note 1: This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (f) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent failure of the tail rotor counterweight bellcrank (bellcrank) retention nut, which could result in the bellcrank migrating off the crosshead spindle, loss of tail rotor control, and subsequent loss of control of the helicopter, accomplish the following:

(a) For Model 205A-1 helicopters:

(1) Within 100 hours time-in-service (TIS) or 90 days after the effective date of this AD, whichever occurs first, remove the two existing retention nuts retaining the bellcranks, part number (P/N) 212-010-709-001 or 212-011-705-001, and install zero hours TIS retention nuts, P/N MS14145L6 or MS17826-6, in accordance with paragraphs (1) through (5) of the Accomplishment Instructions in Bell Helicopter Textron, Inc. Alert Service Bulletin (ASB) 205-00-77, Revision A, September 13, 2000 (205A-1 ASB). A used nut may not be installed.

(2) At intervals not to exceed 100 hours TIS after accomplishing paragraph (a)(1) of this AD, inspect the retention nuts and corrosion preventive compound (CPC) coating in accordance with paragraph (6) of the Accomplishment Instructions of the 205A-1 ASB. Reapply the CPC coating if deficiencies are found in the coverage and protection of the area. Replace any retention nut with any corrosion, mechanical damage, a crack, or looseness with an airworthy new retention nut before further flight.

(b) For Model 205B helicopters:

(1) Within 100 hours TIS or 90 days after the effective date of this AD, whichever occurs first, remove the two existing retention nuts retaining the bellcranks, P/N 212-011-705-001, and install retention nuts, P/N MS14145L6 or MS17826-6, in accordance with paragraphs (1) through (5) of the Accomplishment Instructions in Bell Helicopter Textron, Inc. ASB 205B-00-31, Revision A, dated September 13, 2000 (205B ASB). A used nut may not be installed.

(2) At intervals not to exceed 100 hours TIS after accomplishing paragraph (b)(1) of this AD, inspect the retention nuts and CPC coating in accordance with paragraph (6) of the Accomplishment Instructions in the 205B ASB. Reapply the CPC coating if deficiencies are found in the coverage and protection of the area. Replace any retention nut with any corrosion, mechanical damage, a crack, or looseness with an airworthy new retention nut before further flight.

(c) For Model 212 helicopters:

(1) Within 100 hours TIS or 90 days after the effective date of this AD, whichever occurs first, remove the two existing retention nuts retaining the bellcranks, P/N 212-010-709-001 or 212-011-705-001, and install retention nuts, P/N MS14145L6 or MS17826-6, in accordance with paragraphs (1) through (5) of the Accomplishment Instructions in Bell Helicopter Textron, Inc. Alert Service Bulletin 212-00-107, Revision A, dated September 13, 2000 (212 ASB). A used retention nut may not be installed.

(2) At intervals not to exceed 100 hours TIS after accomplishing paragraph (c)(1) of this AD, inspect the retention nuts and CPC coating in accordance with paragraph (6) of the Accomplishment Instructions in the 212

ASB. Reapply the CPC coating if deficiencies are found in the coverage and protection of the area. Replace any retention nut with any corrosion, mechanical damage, a crack, or looseness with an airworthy new nut before further flight.

(d) *For Model 412 or 412EP helicopters:*

(1) Within 100 hours TIS or 90 days after the effective date of this AD, whichever occurs first, remove the two existing retention nuts retaining the bellcranks, P/N 212-011-705-001, and install retention nuts, P/N MS14145L6 or MS17826-6, in accordance with paragraphs (1) through (5) of the Accomplishment Instructions in Bell Helicopter Textron, Inc. ASB 412-00-102, Revision A, dated September 13, 2000 (412 ASB). A used nut may not be installed.

(2) At intervals not to exceed 100 hours TIS after accomplishing paragraph (d)(1) of this AD, inspect the retention nuts and CPC coating in accordance with paragraph (6) of the Accomplishment Instructions in the 412 ASB. Reapply the CPC coating if deficiencies are found in the coverage and protection of the area. Replace any retention nut with any corrosion, mechanical damage, a crack, or looseness with an airworthy new retention nut before further flight.

(e) *For Model 412CF helicopters:*

(1) Within 100 hours TIS or 90 days after the effective date of this AD, whichever occurs first, remove the two existing retention nuts retaining the bellcranks, P/N 212-011-705-001, and install retention nuts, P/N MS14145L6 or MS17826-6, in accordance with paragraphs (1) through (5) of the Accomplishment Instructions in Bell Helicopter Textron, Inc. ASB 412CF-00-10, Revision A, September 13, 2000 (412CF ASB). A used nut may not be installed.

(2) At intervals not to exceed 100 hours TIS after accomplishing paragraph (e)(1) of this AD, inspect the retention nuts and CPC coating in accordance with paragraph (6) of the Accomplishment Instructions in the 412CF ASB. Reapply the CPC coating if deficiencies are found in the coverage and protection of the area. Replace any retention nut with any corrosion, mechanical damage, a crack, or looseness with an airworthy new nut before further flight.

(f) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Rotorcraft Certification Office, Rotorcraft Directorate, FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Rotorcraft Certification Office.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Rotorcraft Certification Office.

(g) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the helicopter to a location where the requirements of this AD can be accomplished.

(h) The modifications and inspections shall be done in accordance with paragraphs (1) through (6) of the Accomplishment

Instructions in the following Bell Helicopter Textron, Inc. Alert Service Bulletins: No. 205-00-77, Revision A, dated September 13, 2000; No. 205B-00-31, Revision A, dated September 13, 2000; No. 212-00-107, Revision A, dated September 13, 2000; No. 412-00-102, Revision A, dated September 13, 2000; or No. 412CF-00-10, Revision A, September 13, 2000, as applicable. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Bell Helicopter Textron, Inc., P.O. Box 482, Fort Worth, Texas 76101, telephone (817) 280-3391, fax (817) 280-6466. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(i) This amendment becomes effective on August 1, 2001.

Issued in Fort Worth, Texas, on June 13, 2001.

Eric Bries,

*Acting Manager, Rotorcraft Directorate,
Aircraft Certification Service.*

[FR Doc. 01-15794 Filed 6-26-01; 8:45 am]

BILLING CODE 4910-13-U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-SW-04-AD; Amendment 39-12271; AD 2001-12-16]

RIN 2120-AA64

Airworthiness Directives; Eurocopter France Model AS332L2 Helicopters

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment adopts a new airworthiness directive (AD) for Eurocopter France Model AS332L2 helicopters. This AD requires, at specified time intervals, visually inspecting the main rotor blade sleeve yoke (sleeve) for cracks, corrosion, fretting, or bonding separation; the bearing surface of the metal bushing (bushing) for fretting or cracks; and the sleeve-to-damper attachment bolt (bolt) for corrosion and deterioration of the fluorimid varnish coating. Replacing any cracked or nonairworthy sleeve, bushing, or bolt is also required before further flight. This AD is prompted by the discovery of extensive deterioration of the fluorimid varnish coating on the bolt; cracks in the bushing; and fretting and corrosion of the sleeve. The actions specified in this AD are intended to

detect corrosion and cracks in the yoke, which could result in separation of the blade damper assembly and subsequent loss of control of the helicopter.

DATES: Effective July 12, 2001.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of July 12, 2001.

Comments for inclusion in the Rules Docket must be received on or before August 27, 2001.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Office of the Regional Counsel, Southwest Region, Attention: Rules Docket No. 2001-SW-04-AD, 2601 Meacham Blvd., Room 663, Fort Worth, Texas 76137. You may also send comments electronically to the Rules Docket at the following address: 9-asw-adcomments@faa.gov.

The service information referenced in this AD may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005, telephone (972) 641-3460, fax (972) 641-3527. This information may be examined at the FAA, Office of the Regional Counsel, Southwest Region, Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Jim Grigg, Aviation Safety Engineer, FAA, Rotorcraft Directorate, Regulations Group, Fort Worth, Texas 76193-0111, telephone (817) 222-5490, fax (817) 222-5961.

SUPPLEMENTARY INFORMATION: The Direction Générale de L'Aviation Civile (DGAC), which is the airworthiness authority for France, notified the FAA that an unsafe condition may exist on Eurocopter France Model AS332L2 helicopters. The DGAC advises that cracks in the yokes of the damper attachment sleeves may result in loss of the damper attachment and the occurrence of vibrations, leading to loss of control of the helicopter.

Eurocopter issued Eurocopter Service Bulletin No. 05.00.53, Revision 1, dated July 6, 1999, which specifies checking the sleeve yoke for cracks and the damper attachment for damage. The DGAC classified this service bulletin as mandatory and issued AD No. 1999-260-014(A) R1, dated July 13, 1999, to ensure the continued airworthiness of these helicopters in France.

This helicopter model is manufactured in France and is type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation