(f) *Can I comply with this AD in any other way?* (1) You may use an alternative method of compliance or adjust the compliance time if:

(i) Your alternative method of compliance provides an equivalent level of safety; and

(ii) The Manager, Wichita Aircraft Certification Office (ACO), approves your alternative. Submit your request through an FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Wichita ACO.

(2) Alternative methods of compliance approved in accordance with AD 98–13–02, which is superseded by this AD, are approved as alternative methods of compliance for the corresponding portion of this AD.

Note 2: This AD applies to each airplane identified in paragraph (a) of this AD, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (f) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if you have not eliminated the unsafe condition, specific actions you propose to address it.

(g) *Where can I get information about any already-approved alternative methods of compliance?* Contact Mr. T.N. Baktha, Aerospace Engineer, FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Wichita, Kansas 67209; telephone: (316) 946–4155; facsimile: (316) 946–4407.

(h) *What if I need to fly the airplane to another location to comply with this AD?* The FAA can issue a special flight permit under sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate your airplane to a location where you can accomplish the requirements of this AD.

(i) *Are any service bulletins incorporated into this AD by reference?* Actions required by this AD must be done in accordance with Raytheon Service Bulletin SB 27–3358, Issued: February, 2000. The Director of the Federal Register approved this incorporation by reference under 5 U.S.C. 552(a) and 1 CFR part 51. You can get copies from Raytheon Aircraft Company, PO Box 85, Wichita, Kansas 67201–0085. You can look at copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

(j) *Does this AD action affect any existing AD actions?* This amendment supersedes AD 98–13–02, Amendment 39–10590.

(k) *When does this amendment become effective?* This amendment becomes effective on December 10, 2002.

Appendix to AD 2002–21–13

Weight and Balance Accuracy Method No. 1

1. Review existing weight and balance documentation to assure completeness and accuracy of the documentation from the most recent FAA-approved weighing or from factory delivery to date of compliance with this AD.

2. Compare the actual configuration of the airplane to the configuration described in the weight and balance documentation.

3. If equipment additions or deletions are not reflected in the documentation or if modifications affecting the location of the center of gravity (*e.g.*, paint or structural repairs) are not documented, determine the accuracy of the airplane weight and balance data in accordance with Method No. 2.

Weight and Balance Information Accuracy Method No. 2

1. Determine the basic empty weight and center of gravity (CG) of the empty airplane using the Weighing Instructions in the Weight and Balance section of the airplane flight manual/pilot's operating handbook (AFM/POH).

2. Record the results in the airplane records, and use these new values as the basis for computing the weight and CG information as specified in the Weight and Balances section of the AFM/POH.

Issued in Kansas City, Missouri, on October 15, 2002.

Dorenda D. Baker,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 02–26661 Filed 10–23–02; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002–NM–216–AD; Amendment 39–12912; AD 2002–21–06]

RIN 2120–AA64

Airworthiness Directives; McDonnell Douglas Model DC–9–81 (MD–81), DC–9–82 (MD–82), DC–9–83 (MD–83), DC–9–87 (MD–87), and MD–88 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment supersedes an existing airworthiness directive (AD), applicable to all McDonnell Douglas Model DC–9–81 (MD–81), DC–9–82 (MD–82), DC–9–83 (MD–83), DC–9–87 (MD–87), and MD–88 airplanes, that currently requires revisions to the Airplane Flight Manual; installation of inspection aids on the wing upper surfaces; and, among other actions, installation of an overwing heater blanket system or primary upper wing ice detection system, and installation of a heater protection panel or an equipment protection device on certain overwing heater blanket systems. This amendment retains those requirements and adds a requirement to disable the anti-ice systems for the upper wing surface on certain airplanes. The actions specified in this AD are intended to prevent ingestion of ice into one or both engines and consequent loss of thrust from one or both engines; and damage to the upper wing skin surface and its structure, due to prolonged short-circuit electrical arcing of certain anti-ice systems.

DATES: Effective November 8, 2002.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of November 8, 2002.

The incorporation by reference of certain publications, as listed in the regulations, was approved previously by the Director of the Federal Register as January 17, 1992 (57 FR 2014, January 17, 1992).

The incorporation by reference of certain other publications, as listed in the regulations, was approved previously by the Director of the Federal Register as of May 7, 2001 (66 FR 17499, April 2, 2001).

Comments for inclusion in the Rules Docket must be received on or before December 23, 2002.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–114, Attention: Rules Docket No. 2002–NM–216–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227–1232. Comments may also be sent via the Internet using the following address: *9-anm-iarcomment@faa.gov*. Comments sent via fax or the Internet must contain ''Docket No. 2002–NM–216–AD'' in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 for Windows or ASCII text.

The service information referenced in this AD may be obtained from Boeing Commercial Aircraft Group, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1–L5A (D800–0024). This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Los Angeles Aircraft

Certification Office, 3960 Paramount Boulevard, Lakewood, California; or at the Office of the Federal Register, 800 North Capitol Street, NW., Suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: *Technical Information:* Daniel Bui, Aerospace Engineer, Systems and Equipment Branch, ANM–130L, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712–4137; telephone (562) 627–5339; fax (562) 627–5210.

Other Information: Judy Golder, Airworthiness Directive Technical Editor/Writer; telephone (425) 687–4241, fax (425) 227–1232. Questions or comments may also be sent via the Internet using the following address: *judy.golder@faa.gov.* Questions or comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 for Windows or ASCII text.

SUPPLEMENTARY INFORMATION: On May 30, 2001, the FAA issued AD 2001–06–16 COR, amendment 39–12163 (66 FR 31121, June 11, 2001), applicable to all McDonnell Douglas Model DC–9–81 (MD–81), DC–9–82 (MD–82), DC–9–83 (MD–83), DC–9–87 (MD–87), and MD–88 airplanes. That amendment corrected an incorrect paragraph reference in AD 2001–06–16, amendment 39–12163 (66 FR 17499, April 2, 2001). AD 2001–06–16 was prompted by incidents in which ice accumulation on the wing upper surfaces shed into the engines during takeoff. AD 2001–06–16 COR requires revisions to the Airplane Flight Manual (AFM); installation of inspection aids on the wing upper surfaces; and, among other actions, installation of an overwing heater blanket system or primary upper wing ice detection system, and installation of a heater protection panel or an equipment protection device on certain overwing heater blanket systems. The actions required by AD 2002–06–16 COR are intended to prevent ice accumulation on the wing upper surfaces, which could result in ingestion of ice into one or both engines and consequent loss of thrust from one or both engines.

Actions Since Issuance of Previous Rule

Since the issuance of AD 2001–06–16 COR, the FAA has received reports of short-circuit electrical arcing at the upper wing interface unit and on the wing upper surface of the Honeywell Anti-Ice System, which was installed per Supplemental Type Certificate (STC) SA6061NM. During the investigation of one incident, a burn-through hole resulting from a high energy electrical short was observed on the cover plate of the interface unit. Investigation of another incident revealed a small pitting hole on the upper wing surface near the heater. Wire chafing appears to have caused the over-wing heater blanket system to short circuit. This condition, if not corrected, could cause prolonged short-circuit electrical arcing of the anti-ice system, which could result in damage to the upper wing skin surface and its structure.

Issuance of New Service Information

Honeywell has issued Alert Service Bulletin, 109XXXX–30–38, dated August 8, 2002, which describes procedures to disable the upper wing surface anti-ice system for those airplanes on which STC SA6061NM has been installed. The Honeywell anti-ice system specified in that service bulletin is identical to the Allied Signal overwing heater blanket system referred to in AD 2001–06–16 COR.

Explanation of Requirements of Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of this same type design, this AD supersedes AD 2001–06–16 COR. This AD continues to require revisions to the Airplane Flight Manual (AFM); installation of inspection aids on the wing upper surfaces; and, among other actions, installation of an overwing heater blanket system or primary upper wing ice detection system, and installation of a heater protection panel or an equipment protection device on certain overwing heater blanket systems. This AD also requires disabling the anti-ice systems for the upper wing surface of airplanes equipped with Honeywell Anti-Ice Systems installed per STC SA6061NM, per the service bulletin described previously.

Explanation of Change to Applicability

The FAA has revised the applicability of the AD 2002–06–16 COR to identify model designations as published in the most recent type certificate date sheet for the affected models.

Removal of Note 6 From AD 2001–06–16 COR

Note 6 of AD 2001–06–16 COR states that installation of an overwing heater blanket system per Allied Signal STC SA6061NM (also known and specified in this AD as a Honeywell anti-ice system installed per STC SA6061NM) is an approved means of compliance with the requirements of paragraph (f)(2)(ii) of that AD. Since this AD requires disabling the system installed per STC SA6061NM, the FAA has removed the wording of the previous Note 6 from this AD, and renumbered the notes accordingly.

Interim Action

This is considered to be interim action. Honeywell has advised that it currently is developing a modification that will address reactivating the anti-ice system that will be disabled per the requirements of this AD. Once this modification is developed, approved, and available, the FAA may consider additional rulemaking.

Determination of Rule's Effective Date

Since a situation exists that requires the immediate adoption of this regulation, it is found that notice and opportunity for prior public comment hereon are impracticable, and that good cause exists for making this amendment effective in less than 30 days.

Comments Invited

Although this action is in the form of a final rule that involves requirements affecting flight safety and, thus, was not preceded by notice and an opportunity for public comment, comments are invited on this rule. Interested persons are invited to comment on this rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified under the caption **ADDRESSES.** All communications received on or before the closing date for comments will be considered, and this rule may be amended in light of the comments received. Factual information that supports the commenter's ideas and suggestions is extremely helpful in evaluating the effectiveness of the AD action and determining whether additional rulemaking action would be needed.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.
- For each issue, state what specific change to the AD is being requested.
- Include justification (*e.g.*, reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the rule that might suggest a need to modify the rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by

interested persons. A report that summarizes each FAA-public contact concerned with the substance of this AD will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this rule must submit a self-addressed, stamped postcard on which the following statement is made: ''Comments to Docket Number 2002–NM–216–AD.'' The postcard will be date stamped and returned to the commenter.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

The FAA has determined that this regulation is an emergency regulation that must be issued immediately to correct an unsafe condition in aircraft, and that it is not a ''significant regulatory action'' under Executive Order 12866. It has been determined further that this action involves an emergency regulation under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979). If it is determined that this emergency regulation otherwise would be significant under DOT Regulatory Policies and Procedures, a final regulatory evaluation will be prepared and placed in the Rules Docket. A copy of it, if filed, may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES.**

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by removing amendment AD 2001–06–16 COR, 39–12163 (66 FR 31121, June 11, 2001), and by adding a new airworthiness directive (AD), amendment 39–12912, to read as follows:

2002–21–06 **McDonnell Douglas**: Amendment 39–12912. Docket 2002–NM–216–AD. Supersedes AD 2001–06–16 COR, Amendment 39–12163.

Applicability: All Model DC–9–81 (MD–81), DC–9–82 (MD–82), DC–9–83 (MD–83), DC–9–87 (MD–87), and MD–88 airplanes; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (l)(1) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent damage to the upper wing skin surface and its structure, due to prolonged short-circuit electrical arcing of the anti-ice system; accomplish the following:

Restatement of AD 2001–06–16 COR

Airplane Flight Manual Revision

(a) Within 10 days after January 17, 1992 (the effective date of AD 92–03–02, amendment 39–8156), revise the Limitations Section of the FAA-approved Airplane Flight Manual (AFM) to include the following. This may be accomplished by inserting a copy of this AD in the AFM.

''Ice on Wing Upper Surfaces

CAUTION

Ice shedding from the wing upper surface during takeoff can cause severe damage to one or both engines, leading to surge, vibration, and complete thrust loss. The formation of ice can occur on wing surfaces during exposure of the airplane to normal icing conditions. Clear ice can also occur on the wing upper surfaces when cold-soaked fuel is in the main wing fuel tanks, and the airplane is exposed to conditions of high humidity, rain, drizzle, or fog at ambient temperatures well above freezing. Often, the ice accumulation is clear and difficult to detect visually. The ice forms most frequently on the inboard, aft corner of the main wing tanks. [END OF CAUTIONARY NOTE]

The wing upper surfaces must be physically checked for ice when the airplane has been exposed to conditions conducive to ice formation. Takeoff may not be initiated unless the flight crew verifies that a visual check and a physical (hands-on) check of the wing upper surfaces have been accomplished, and that the wing is clear of ice accumulation when any of the following conditions occur:

(1) When the ambient temperature is less than 50 degrees F and high humidity or visible moisture (rain, drizzle, sleet, snow, fog, etc.) is present;

(2) When frost or ice is present on the lower surface of either wing;

(3) After completion of de-icing.

When inspection aids (*i.e.* tufts, decals, mount pads, painted symbols, and paint stripes) are installed in accordance with McDonnell Douglas MD–80 Service Bulletin 30–59, the physical check may be made by assuring that all installed tufts move freely.

NOTE

This limitation does not relieve the requirement that aircraft surfaces are free of frost, snow, and ice accumulation, as required by Federal Aviation Regulations Sections 91.527 and 121.629. [END OF NOTE]''

AFM Configuration Deviation List Revision

(b) Within 10 days after January 17, 1992, revise the Configuration Deviation List (CDL) Appendix of the FAA-approved AFM to include the following. This may be accomplished by inserting a copy of this AD in the AFM.

''30–80–01 Triangular Decal and Tuft Assemblies

Up to two (2) decals or tufts per side may be missing, provided:

(a) At least one decal and tuft on each side is located along the aft spar line; and

(b) The tufts are used for performing the physical check to determine that the upper wing is free of ice by observing that the tufts move freely.

Up to eight (8) decals and/or tufts may be missing, provided:

(a) Takeoff may not be initiated unless the flight crew verifies that a physical (hands-on) check is made of the upper wing in the location of the missing decals and/or tufts to assure that there is no ice on the wing when icing conditions exist;

OR

(b) When the ambient temperature is more than 50 degrees F.''

Installation of Inspection Aids

(c) Within 30 days after January 17, 1992, install inspection aids (*i.e.*, tufts, decals, mount pads, painted symbols, and paint stripes) on the inboard side of the wings' upper surfaces, in accordance with McDonnell Douglas Service Bulletin 30–59, dated September 18, 1989; Revision 1, dated January 5, 1990; or Revision 2, dated August 15, 1990.

Repetitive Tests and One-Time Inspection

(d) For airplanes on which an overwing heater blanket system was installed without installation of a heater protection panel (HPP) or an equipment protection device (EPD) prior to May 7, 2001 (the effective date of 2001–06–16 COR, amendment 39–12163): Within 60 days days after May 7, 2001, accomplish the actions specified in paragraph (d)(1) or (d)(2) of this AD, as applicable.

(1) For airplanes on which the overwing heater blanket system was installed in accordance with McDonnell Douglas Service Bulletin MD80–30–071, Revision 02, dated February 6, 1996; or McDonnell Douglas Service Bulletin MD80–30–078, Revision 01, dated April 8, 1997: Accomplish paragraphs (d)(1)(i) and (d)(1)(ii) of this AD.

(i) Remove secondary access covers, and perform a one-time detailed visual inspection to detect discrepancies (mechanical damage or punctures in the upper skin of the blanket, prying damage on the panel, and fuel leakage) of the overwing heater blanket, in accordance with McDonnell Douglas Alert Service Bulletin MD80–30A087, dated September 22, 1997. And,

(ii) Accomplish paragraph (d)(1)(ii)(A) or (d)(1)(ii)(B) of this AD.

(A) Perform dielectric withstanding voltage and resistance tests in accordance with McDonnell Douglas Alert Service Bulletin MD80–30A087, dated September 22, 1997. Repeat the tests thereafter at intervals not to exceed 150 days, until installation of an HPP in accordance with paragraph (f)(1)(i) or (f)(1)(ii) of this AD, as applicable.

(B) Deactivate the overwing heater blanket system until accomplishment of dielectric withstanding voltage and resistance tests specified in paragraph (d)(1)(ii)(A). If the overwing heater blanket system is deactivated as provided by this paragraph, continue to accomplish the requirements of paragraphs (a), (b), and (c) of this AD.

Note 2: For the purposes of this AD, a detailed visual inspection is defined as: ''An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required.''

(2) For airplanes on which the overwing heater blanket system was installed in accordance with TDG Aerospace, Inc., STC SA6042NM: Accomplish paragraphs (d)(2)(i) and (d)(2)(ii) of this AD.

(i) Remove secondary access covers, and perform a one-time detailed visual inspection to detect discrepancies (mechanical damage or punctures in the upper skin of the blanket, prying damage on the panel, and fuel leakage) of the overwing heater blanket, in accordance with McDonnell Douglas Alert Service Bulletin MD80–30A087, dated September 22, 1997. And,

(ii) Accomplish paragraph (d)(2)(ii)(A) or (d)(2)(ii)(B) of this AD.

(A) Perform dielectric withstanding voltage and resistance tests in accordance with McDonnell Douglas Alert Service Bulletin MD80–30A087, dated September 22, 1997. Repeat the tests thereafter at intervals not to exceed 150 days, until installation of an EPD in accordance with paragraph (f)(2)(i) of this AD.

(B) Deactivate overwing heater blanket system until accomplishment of dielectric withstanding voltage and resistance tests specified in paragraph (d)(2)(ii)(A). If the overwing heater blanket system is deactivated as provided by this paragraph, continue to accomplish the requirements of paragraphs (a), (b), and (c) of this AD.

Corrective Action

(e) If any discrepancy is detected during any inspection or test performed in accordance with paragraph (d) of this AD, prior to further flight, repair or replace the affected heater blanket, in accordance with McDonnell Douglas Alert Service Bulletin MD80–30A087, dated September 22, 1997; except as provided in paragraph (h) of this AD.

Note 3: McDonnell Douglas Alert Service Bulletin MD80–30A087, dated September 22, 1997, references TDG Aerospace Document E95–451, Revision B, dated January 31, 1996, as an additional source of service information for accomplishment of repair or replacement of the overwing heater blanket.

Installation of Overwing Heater Blanket or Primary Upper Wing Ice Detection System

(f) Within 3 years after May 7, 2001, do the requirements of either paragraph (f)(1) or (f)(2) of this AD.

(1) Do the actions specified in paragraph (f)(1)(i) or (f)(1)(ii) of this AD, as applicable.

(i) For airplanes listed in Group 1 in McDonnell Douglas Service Bulletin MD80–30–090, dated October 19, 1999: Install an overwing heater blanket system in accordance with McDonnell Douglas Service Bulletin MD80–30–071, Revision 02, dated February 6, 1996; and modify and reidentify the existing HPP in accordance with McDonnell Douglas Service Bulletin MD80–30–090. Modification of the existing HPP in accordance with this paragraph constitutes terminating action for the repetitive inspections required by (d)(1)(ii)(A) of this AD.

(ii) For airplanes listed in Group 2 in McDonnell Douglas Service Bulletin MD80–30–090, dated October 19, 1999: Install an overwing heater blanket system in accordance with McDonnell Douglas Service Bulletin MD80–30–078, Revision 01, dated April 8, 1997; and install an HPP and associated wiring in accordance with McDonnell Douglas Service Bulletin MD80–30–090. Installation of an HPP and associated wiring in accordance with this paragraph constitutes terminating action for the repetitive inspections required by (d)(1)(ii)(A) of this AD.

Note 4: For other airplanes, accomplishment of the requirements of paragraph (f)(1)(i) or (f)(1)(ii) of this AD may be acceptable per paragraph (i)(1) of this AD.

(2) Accomplish the actions specified in either paragraph (f)(2)(i), (f)(2)(ii), or (f)(2)(iii) of this AD.

(i) Install an overwing heater blanket system, and install an EPD that provides a circuit protection function to the overwing heater blanket, in accordance with a method approved by the Manager, Los Angeles Aircraft Certification Office (ACO), FAA. Installation of an EPD in accordance with this paragraph constitutes terminating action for the repetitive inspections required by (d)(2)(ii)(A) of this AD.

Note 5: Installation of an overwing heater blanket system and installation of an EPD that provides a circuit protection function to the overwing heater blanket, in accordance with TDG Aerospace, Inc., SA6042NM, or TDG Master Drawing List (MDL) E93–104, Revision R, dated October 25, 2000; is an approved means of compliance with the requirements of paragraph (f)(2)(i) of this AD.

(ii) Install an overwing heater blanket system in accordance with a method approved by the Manager, Los Angeles ACO.

(iii) Install an FAA-approved primary upper wing ice detection system in accordance with a method approved by the Manager, Los Angeles ACO.

Note 6: Boeing (McDonnell Douglas) has received FAA approval of a primary upper wing ice detection system that is considered to be an alternative method of compliance (AMOC) with the requirements of paragraph (f)(2)(iii) of this AD. Information concerning such AMOCs may be obtained from the Los Angeles ACO.

AFM Revision

(g) Except as provided by paragraph (h) of this AD, prior to further flight after accomplishment of the installation required by paragraph (f)(1) or (f)(2) of this AD, revise the Limitations Section of the FAA-approved AFM to include the following. This may be accomplished by inserting a copy of this AD in the AFM. After accomplishment of the installation required by paragraph (f)(1) or (f)(2) of this AD and this AFM revision, the AFM revisions required by paragraphs (a) and (b) of this AD may be removed from the AFM, and the inspection aids required by paragraph (c) of this AD may be removed from the airplane.

''Ice on Wing Upper Surfaces

CAUTION

Ice shedding from the wing upper surface during takeoff can cause severe damage to one or both engines, leading to surge, vibration, and complete thrust loss. The formation of ice can occur on wing surfaces during exposure of the airplane to normal icing conditions. Clear ice can also occur on the wing upper surfaces when cold-soaked fuel is in the main wing fuel tanks, and the airplane is exposed to conditions of high humidity, rain, drizzle, or fog at ambient temperatures well above freezing. Often, the ice accumulation is clear and difficult to detect visually. The ice forms most frequently on the inboard, aft corner of the main wing tanks. [END OF CAUTIONARY NOTE]''

MMEL Provision

(h) An airplane may be operated with an inoperative overwing heater blanket or primary upper wing ice detection system for 10 days per the Master Minimum Equipment List (MMEL), provided that the actions specified in paragraphs (h)(1), (h)(2), and (h)(3) of this AD are done before further flight.

(1) Revise the Limitations Section of the FAA-approved AFM to include the following. This may be accomplished by inserting a copy of this AD in the AFM.

''Ice on Wing Upper Surfaces

CAUTION

The wing upper surfaces must be physically checked for ice when the airplane has been exposed to conditions conducive to ice formation. Takeoff may not be initiated unless the flight crew verifies that a visual check and a physical (hands-on) check of the wing upper surfaces have been accomplished, and that the wing is clear of ice accumulation when any of the following conditions occur:

(1) When the ambient temperature is less than 50 degrees F and high humidity or visible moisture (rain, drizzle, sleet, snow, fog, etc.) is present;

(2) When frost or ice is present on the lower surface of either wing;

(3) After completion of de-icing.

When inspection aids (*i.e.* tufts, decals, mount pads, painted symbols, and paint stripes) are installed in accordance with McDonnell Douglas MD–80 Service Bulletin 30–59, the physical check may be made by assuring that all installed tufts move freely.

NOTE

This limitation does not relieve the requirement that aircraft surfaces are free of frost, snow, and ice accumulation, as required by Federal Aviation Regulations Sections 91.527 and 121.629. [END OF NOTE]''

(2) Revise the CDL Appendix of the FAA-approved AFM to include the following. This may be accomplished by inserting a copy of this AD in the AFM.

''30–80–01 Triangular Decal and Tuft Assemblies

Up to two (2) decals or tufts per side may be missing, provided:

(a) At least one decal and tuft on each side is located along the aft spar line; and

(b) The tufts are used for performing the physical check to determine that the upper wing is free of ice by observing that the tufts move freely.

Up to eight (8) decals and/or tufts may be missing, provided:

(a) Takeoff may not be initiated unless the flight crew verifies that a physical (hands-on) check is made of the upper wing in the location of the missing decals and/or tufts to assure that there is no ice on the wing when icing conditions exist;

OR

(b) When the ambient temperature is more than 50 degrees F.''

(3) Install inspection aids (*i.e.*, tufts, decals, mount pads, painted symbols, and paint stripes) on the inboard side of the wings' upper surfaces, in accordance with McDonnell Douglas Service Bulletin 30–59, dated September 18, 1989; Revision 1, dated January 5, 1990; or Revision 2, dated August 15, 1990.

New Requirements of This AD

Note 7: The Honeywell Anti-Ice System specified in paragraphs (i), (j), and (k) of this AD, is also known and specified as an overwing heater blanket system installed in accordance with AlliedSignal Supplemental Type Certificate (STC) STC SA6061NM.

For Airplanes Equipped With a Honeywell Anti-Ice System Installed per STC SA6061NM

(i) For airplanes equipped with a Honeywell Anti-Ice System installed per STC SA6061NM: Accomplish the actions specified in paragraphs (i)(1), (i)(2), (i)(3), and (i)(4) of this AD, at the times specified in those paragraphs.

(1) Within 72 hours after the effective date of this AD, disable the Honeywell Anti-Ice System installed per STC SA6061NM, per Honeywell Alert Service Bulletin 109XXXX–30–38, dated August 8, 2002.

(2) Within 72 hours after the effective date of this AD, revise the Limitations Section of the FAA-approved AFM to include the following (this may be accomplished by inserting a copy of this AD in the AFM):

''Ice on Wing Upper Surfaces

CAUTION

Ice shedding from the wing upper surface during takeoff can cause severe damage to one or both engines, leading to surge, vibration, and complete thrust loss. The formation of ice can occur on wing surfaces during exposure of the airplane to normal icing conditions. Clear ice can also occur on the wing upper surfaces when cold-soaked fuel is in the main wing fuel tanks, and the airplane is exposed to conditions of high humidity, rain, drizzle, or fog at ambient temperatures well above freezing. Often, the ice accumulation is clear and difficult to detect visually. The ice forms most frequently on the inboard, aft corner of the main wing tanks. [END OF CAUTIONARY NOTE]

The wing upper surfaces must be physically checked for ice when the airplane has been exposed to conditions conducive to ice formation. Takeoff may not be initiated unless the flight crew verifies that a visual check and a physical (hands-on) check of the wing upper surfaces have been accomplished, and that the wing is clear of ice accumulation when any of the following conditions occur:

(1) When the ambient temperature is less than 50 degrees F and high humidity or visible moisture (rain, drizzle, sleet, snow, fog, etc.) is present;

(2) When frost or ice is present on the lower surface of either wing;

(3) After completion of de-icing.

When inspection aids (*i.e.* tufts, decals, mount pads, painted symbols, and paint stripes) are installed in accordance with McDonnell Douglas MD–80 Service Bulletin 30–59, the physical check may be made by assuring that all installed tufts move freely.

NOTE

This limitation does not relieve the requirement that aircraft surfaces are free of frost, snow, and ice accumulation, as required by Federal Aviation Regulations Sections 91.527 and 121.629. [END OF NOTE]''

AFM Configuration Deviation List Revision

(3) Within 72 hours after the effective date of this AD, revise the CDL Appendix of the FAA-approved AFM to include the following (this may be accomplished by inserting a copy of this AD in the AFM):

''30–80–01 Triangular Decal and Tuft Assemblies

Up to two (2) decals or tufts per side may be missing, provided:

(a) At least one decal and tuft on each side is located along the aft spar line; and

(b) The tufts are used for performing the physical check to determine that the upper wing is free of ice by observing that the tufts move freely.

Up to eight (8) decals and/or tufts may be missing, provided:

(a) Takeoff may not be initiated unless the flight crew verifies that a physical (hands-on) check is made of the upper wing in the location of the missing decals and/or tufts to assure that there is no ice on the wing when icing conditions exist;

OR

(b) When the ambient temperature is more than 50 degrees F.''

Installation of Inspection Aids

(4) Within 30 days after the effective date of this AD, install inspection aids (*i.e.*, tufts, decals, mount pads, painted symbols, and paint stripes) on the inboard side of the wings' upper surfaces, in accordance with McDonnell Douglas Service Bulletin 30–59, dated September 18, 1989; Revision 1, dated January 5, 1990; or Revision 2, dated August 15, 1990.

Note 8: Operators should note that certain AMOCs have been approved as acceptable methods of compliance with paragraph (i)(4) of this AD. Information concerning such AMOCs may be obtained from the Manager, Los Angeles ACO.

Installation of Overwing Heater Blanket or Primary Upper Wing Ice Detection System

(j) For airplanes equipped with disabled Honeywell Anti-Ice Systems installed per STC SA6061NM: Within 3 years after May 7, 2001, accomplish the requirements of paragraph (j)(l), (j)(2), or (j)(3) of this AD.

(1) Install an overwing heater blanket system, and install an EPD that provides a circuit-protection function to the overwing heater blanket, in accordance with a method approved by the Manager, Los Angeles ACO, FAA.

Note 9: Installation of an overwing heater blanket system and installation of an EPD that provides a circuit-protection function to the overwing heater blanket, in accordance with TDG Aerospace, Inc., SA6042NM, or TDG Master Drawing List (MDL) E93–104, Revision R, dated October 25, 2000; is an approved means of compliance with the requirements of paragraph (j)(1) of this AD.

(2) Install an overwing heater blanket system in accordance with a method approved by the Manager, Los Angeles ACO.

(3) Install an FAA-approved primary upper wing ice detection system in accordance with a method approved by the Manager, Los Angeles ACO.

Note 10: Boeing (McDonnell Douglas) has received FAA approval of an acceptable primary upper wing ice detection system, which is considered to be an acceptable method of compliance with the requirements of paragraph (j)(3) of this AD when

accomplished in accordance with a method approved by the Manager, Los Angeles ACO.

AFM Revision

(k)(1) For airplanes equipped with a disabled Honeywell Anti-Ice Systems installed per STC SA6061NM: Prior to further flight after accomplishment of the installation required by paragraph (j)(1), (j)(2), or (j)(3) of this AD, revise the Limitations Section of the FAA-approved AFM to include the following (this may be accomplished by inserting a copy of this AD in the AFM):

''Ice on Wing Upper Surfaces

CAUTION

Ice shedding from the wing upper surface during takeoff can cause severe damage to one or both engines, leading to surge, vibration, and complete thrust loss. The formation of ice can occur on wing surfaces during exposure of the airplane to normal icing conditions. Clear ice can also occur on the wing upper surfaces when cold-soaked fuel is in the main wing fuel tanks, and the airplane is exposed to conditions of high humidity, rain, drizzle, or fog at ambient temperatures well above freezing. Often, the ice accumulation is clear and difficult to detect visually. The ice forms most frequently on the inboard, aft corner of the main wing tanks. [END OF CAUTIONARY NOTE]''

(2) After accomplishment of the installation required by paragraph (j)(1) of this AD and this AFM revision, the AFM revisions and CDLs required by paragraphs (i)(2) and (i)(3) of this AD may be removed from the AFM, and the inspection aids required by paragraph (i)(4) of this AD may be removed from the airplane.

Alternative Methods of Compliance (AMOCs)

(l)(1) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Los Angeles ACO, FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Los Angeles ACO.

(2) The following AMOCs were approved previously per AD 92–03–02, amendment 39–8156, and are approved as AMOCs with the indicated paragraphs of this AD:

(i) Installation of a non-skid, striped triangular symbol per Option 5 of McDonnell Douglas Service Bulletin MD80–30–059, Revision 4 through Revision 7, is approved as an AMOC with paragraphs (b) and (i)(2) of this AD; and

(ii) Revision of the Configuration Deviation List (CDL) Appendix of the AFM by inserting a copy of CDL Appendix, Section I, Page 2A, dated March 10, 1993, into the AFM, is approved as an AMOC with paragraphs (c) and (i)(3) of this AD.

Note 11: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Los Angeles ACO.

Special Flight Permits

(m) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(n) Unless otherwise specified in this AD, the actions shall be done in accordance with the applicable service document identified in the following table:

Service Document	Revision Level	Date
Honeywell Alert Service Bulletin 109XXXX–30–38	Original	August 8, 2002
McDonnell Douglas Alert Service Bulletin MD80–30A087	Original	September 22, 1997
McDonnell Douglas Service Bulletin 30–59	Original	September 18, 1989
McDonnell Douglas Service Bulletin 30–59	1	January 5, 1990
McDonnell Douglas Service Bulletin 30–59	2	August 15, 1990
McDonnell Douglas Service Bulletin MD80–30–071	02	February 6, 1996
McDonnell Douglas Service Bulletin MD80–30–078	01	April 8, 1997
McDonnell Douglas Service Bulletin MD80–30–090	Original	October 19, 1999

(1) The incorporation by reference of Honeywell Alert Service Bulletin 109XXXX–30–38, dated August 8, 2002, is approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The incorporation by reference of McDonnell Douglas Service Bulletin 30–59, dated September 18, 1989; McDonnell Douglas Service Bulletin 30–59, Revision 1, dated January 5, 1990; and McDonnell Douglas Service Bulletin 30–59, Revision 2, dated August 15, 1990; was approved previously by the Director of the Federal Register as of January 17, 1992 (57 FR 2014, January 17, 1992).

(3) The incorporation by reference of the remaining service bulletins listed in Table 1 of this AD, was approved previously by the Director of the Federal Register as of May 7, 2001 (66 FR 17499, April 2, 2001).

(4) Copies may be obtained from Boeing Commercial Aircraft Group, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1–L5A (D800–0024). Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California; or at the Office of the Federal Register, 800 North Capitol Street, NW., Suite 700, Washington, DC.

Effective Date

(o) This amendment becomes effective on November 8, 2002.

Issued in Renton, Washington, on October 9, 2002.

Ali Bahrami,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 02–26480 Filed 10–23–02; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2000–NM–57–AD; Amendment 39–12915; AD 2002–21–09]

RIN 2120–AA64

Airworthiness Directives; McDonnell Douglas Model DC–9–10, –20, –30, –40, and –50 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain McDonnell Douglas Model DC–9–10, –20, –30, –40, and –50 series airplanes. This AD requires, among other actions, various