

(j) Material Incorporated by Reference

None.

Issued in Renton, Washington, on January 2, 2018.

Michael Kaszycki,

*Acting Director, System Oversight Division,
Aircraft Certification Service.*

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DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2017-0707; Product Identifier 2016-NM-014-AD; Amendment 39-19185; AD 2018-03-12]

RIN 2120-AA64

Airworthiness Directives; Airbus Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain Airbus Model A318 series airplanes; Model A319 series airplanes; Model A320-211, -212, -214, -231, -232, and -233 airplanes; and Model A321-111, -112, -131, -211, -212, -213, -231, and -232 airplanes. This AD was prompted by reports of fatigue damage in the structure for the door stop fittings on certain fuselage frames (FR). This AD requires repetitive rototest inspections for cracking of the fastener holes in certain door stop fittings, and repair if necessary. We are issuing this AD to address the unsafe condition on these products.

DATES: This AD is effective March 19, 2018.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of March 19, 2018.

ADDRESSES: For service information identified in this final rule, contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone: +33 5 61 93 36 96; fax: +33 5 61 93 44 51; email: account.airworth-eas@airbus.com; internet: <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Standards Branch, 1601 Lind Avenue SW, Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the internet at <http://www.regulations.gov> by searching

for and locating Docket No. FAA-2017-0707.

Examining the AD Docket

You may examine the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0707; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone: 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE, Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT:

Sanjay Ralhan, Aerospace Engineer, International Section, Transport Standards Branch, FAA, 1601 Lind Avenue SW, Renton, WA 98057-3356; telephone: 425-227-1405; fax: 425-227-1149.

SUPPLEMENTARY INFORMATION:**Discussion**

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to certain Airbus Model A318 series airplanes; Model A319 series airplanes; Model A320-211, -212, -214, -231, -232, and -233 airplanes; and Model A321-111, -112, -131, -211, -212, -213, -231, and -232 airplanes. The NPRM published in the **Federal Register** on July 25, 2017 (82 FR 34449) (“the NPRM”). The NPRM was prompted by reports of fatigue damage in the structure for the door stop fittings on certain fuselage frames. The NPRM proposed to require repetitive rototest inspections for cracking of the fastener holes in certain door stop fittings, and repair if necessary. We are issuing this AD to detect and correct cracking at the door stop fitting holes of fuselage FR66 and FR68. Such cracking could result in reduced structural integrity of the airplane due to the failure of structural components.

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA AD 2016-0238, dated December 2, 2016; corrected January 4, 2017 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition for certain Airbus Model A318 series airplanes; Model A319 series airplanes;

Model A320-211, -212, -214, -231, -232, and -233 airplanes; and Model A321-111, -112, -131, -211, -212, -213, -231, and -232 airplanes. The MCAI states:

During an A320 fatigue test campaign, it was determined that fatigue damage could appear at the door stop fitting holes of fuselage frame (FR) 66 and FR 68 on left hand (LH) and right hand (RH) sides.

This condition, if not detected and corrected, could affect the structural integrity of the airframe.

Two inspections, Airworthiness Limitations Item (ALI) tasks 534129 and 534130, were introduced in the Airworthiness Limitations Section (ALS) Part 2 with the April 2012 revision and with some compliance time changes with Revision 3 of ALS Part 2 of October 2014.

Since these ALI tasks were implemented, a significant number of reports [were] received concerning non-critical damage and early crack findings. Prompted by these reports, Airbus published SB A320-53-1288 and SB A320-53-1290, providing inspection instructions to improve damage management and modification instructions.

Consequently, EASA issued AD 2016-0015, requiring repetitive rototest inspections of the affected door stop fitting holes and, depending on findings, repair of any cracked area(s).

Since that [EASA] AD was issued, ALS Part 2 Revision 04 and later on Revision 05 were published, introducing updated thresholds and/or intervals for some tasks as specified in Airbus SB A320-53-1288, introducing new configuration of aeroplane with RETRO WING having accomplished SB A320-57-1193 (mod 160080), and keeping the threshold or interval only in flight cycles (FC).

For the reasons described above, this [EASA] AD retains the requirements of EASA AD 2016-0015, which is superseded, but requires those actions within the updated thresholds and intervals. In addition, a corrected threshold for pre-mod 160021 A321 aeroplanes is introduced and the Applicability is reduced to exclude configurations that are not affected.

This [EASA] AD is republished to clarify some requirements in Appendix 1 [in this EASA AD].

You may examine the MCAI in the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0707.

Comments

We gave the public the opportunity to participate in developing this AD. The following presents the comments received on the NPRM and the FAA's response to each comment.

Request To Add a Grace Period for Certain Repetitive Inspections

United Airlines (UAL) requested that we revise paragraph (h) of the proposed AD to allow a 60-day grace period after

the effective date of this AD to give operators time to update their maintenance programs. UAL noted that for airplanes on which inspections were previously accomplished as specified in airworthiness limitation item (ALI) task 534129 or 534130, paragraph (h) of the proposed AD requires future inspections be done in accordance with Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016. UAL noted that operators who are not yet incorporating Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016, may have to schedule special inspection visits instead of doing the inspections during scheduled maintenance.

We agree with the commenter's request to add a grace period to paragraph (h) of this AD to allow operators to plan for the new inspection interval. However, since the commenter did not provide adequate justification to support a 60-day grace period, we have determined that a 30-day grace period is appropriate. Additionally, under the provisions of paragraph (q)(1) of this AD, we will consider requests for approval of an extension of the compliance time if sufficient data are submitted to substantiate that the new compliance time would provide an acceptable level of safety. We have revised paragraph (h) of this AD to include a 30-day grace period.

Request To Allow Deviations From the Service Information for Certain Modified Airplanes

UAL requested that either the service information or the proposed AD be revised to provide alternate instructions for airplanes with modified hardware. UAL noted that “paragraph (i)” of the proposed AD requires repetitive inspections on airplanes modified by cold working fastener holes, which includes installing oversize hardware. UAL pointed out that the inspections must be done in accordance with Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016, which requires using nominal size hardware that no longer exists on modified airplanes.

We infer that the commenter meant to refer to paragraph (j) of the proposed AD, which discusses post-modification inspections, rather than paragraph (i) of the proposed AD, which discusses an optional modification. We agree with the commenter's request. We acknowledge that Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03,

dated October 3, 2016, does not specifically address oversize hardware; however, EASA has stated that “the same inspection principle applies for post SB [Service Bulletin] 53–1290 configuration.” Therefore, we have retained Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016, in paragraph (j) of this AD. We have also revised paragraph (j) of this AD to include an option for operators to obtain inspection instructions using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA Design Organization Approval (DOA).

Request To Remove a Reference to a Non-Terminating Action

UAL requested that we remove the statement “repair of an airplane as required by this paragraph does not constitute terminating action for the repetitive inspections required by paragraph (g) or (j) of this AD for that repair, unless specified otherwise” from paragraph (k) of the proposed AD. UAL suggested that statement be replaced with one instructing operators to accomplish inspections as specified in the repair instructions.

UAL noted that paragraph (k) of the proposed AD states that a crack repair must be done using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA and says that such a repair does not constitute terminating action for the repetitive inspections done in accordance with Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016, unless specified otherwise.

UAL pointed out that Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016, contains language to allow operators to accomplish crack repairs in accordance with structural repair manual (SRM) 53–41–12, and then perform inspections of the repaired area in accordance with SRM 53–41–12. UAL noted that the SRM repair instructions do not state that they terminate the inspections in Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016. UAL further noted that Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016, only applies to unrepaired areas with nominal size holes (the repaired areas would have oversized holes).

We disagree with the commenter's request. We acknowledge that Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016, allows repairs to be done using an SRM. However, this AD does not include that allowance since SRMs published before the effective date of this AD might not address the unsafe condition identified in this AD. Therefore, paragraph (k) of this AD requires repairs to be done using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA.

Additionally, the statement that the commenter requested us to remove from paragraph (k) of this proposed AD aligns with the MCAI. The statement is meant to clarify that doing a repair does not necessarily terminate the repetitive inspections; the repetitive inspections would only be terminated if the repair approved by the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA specifically states that the inspections are terminated. If the approved repair does not state that the inspections are terminated, operators must continue to inspect using Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016, or using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA. Therefore, we have not changed the requirements in paragraph (k) of this AD nor have we removed the statement identified by the commenter. However, we have revised paragraph (g) of this AD to include an option for operators to obtain inspection instructions using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA.

Request To Remove Requirement To Obtain Certain Inspection Instructions

UAL requested that we remove paragraph (l)(2) of the proposed AD because it has no real purpose. UAL noted that paragraph (l)(2) of the proposed AD requires operators to obtain inspection instructions and corrective actions for all repaired fastener holes by contacting the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA. UAL claimed that if a repair was accomplished using the instructions in an SRM or repair design approval sheet (RDAS), the repair approval contains, at a minimum, the initial compliance threshold. UAL added that it is standard

practice for operators to contact Airbus prior to the inspection threshold if the compliance method and intervals are not yet defined.

We disagree with the commenter's request. Airbus intends to provide specific instructions for airplanes inspected in accordance with ALI task 534129 or task 534130 and repaired in accordance with an SRM or RDAS published before the effective date of this AD. Since repair instructions published before the effective date of this AD might not address the unsafe condition identified in this AD, the SRM or RDAS instructions might need to be re-evaluated or revised to address the unsafe condition. In addition, we do not rely on an operator's standard practices, and instead require operators to obtain inspections and corrective actions to address the unsafe condition. We have not revised this AD regarding this issue.

Request To Clarify Actions for Airplanes With Certain Repairs

UAL requested that we delete paragraph (n) of the proposed AD. UAL noted that paragraph (n) of the proposed AD requires operators to determine if a repair was done using an RDAS that is unrelated to ALI task 534129 or task 534130. UAL suggested that the repair instructions would have to state that the damage was found as a result of the applicable ALI, but noted that the ALI task is an inspection that may not be referenced in a documented repair. UAL questioned the relevance of whether or not a repair was related to ALI task 534129 or task 534130, noting that the same considerations are given to repair instructions, regardless of how damage was found. UAL stated that operators would know to seek an alternative method of compliance (AMOC) if they cannot inspect a previously repaired area in accordance with Airbus Service Bulletin A320–53–1288, Revision 01,

including Appendixes 01, 02, and 03, dated October 3, 2016.

We disagree with the commenter's request. The intent of paragraph (n) of the proposed AD is to require operators to re-evaluate existing repairs performed using an Airbus RDAS unrelated to ALI task 534129 or task 534130 because those repairs may not address the findings from the specific inspection types required by the ALI tasks. Therefore, the corresponding repairs might not address the unsafe condition and operators might need new instructions. We have not changed this AD in this regard.

Request To Verify the Latest Service Information is Referenced

UAL requested that, prior to the release of this final rule, we verify that we are referencing the latest revisions of Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016; and Airbus Service Bulletin A320–53–1290, Revision 01, dated October 3, 2016.

We agree with the commenter's request. We have verified that no later revisions of the service information have been issued, and no change is needed to this AD.

Explanation of Change to the Final Rule

In the proposed AD, Table 1 to paragraphs (g) and (j) of this AD and Table 2 to paragraphs (g) and (j) of this AD included a compliance time that stated “. . . or before November 30, 2017. . . .” Since this final rule will become effective after November 30, 2017, we have changed this statement to read “. . . or within 30 days after the effective date of this AD. . . .” We have determined that this revised compliance time addresses the unsafe condition.

Conclusion

We reviewed the relevant data, considered the comments received, and

determined that air safety and the public interest require adopting this AD with the changes described previously and minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

We also determined that these changes will not increase the economic burden on any operator or increase the scope of this AD.

Related Service Information Under 1 CFR Part 51

Airbus has issued the following service information.

- Airbus Service Bulletin A320–53–1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016. This service information describes procedures for rototest inspections for cracking of the fastener holes in the airframe structure for the door stop fittings installation in FR66 and FR68.
- Airbus Service Bulletin A320–53–1290, Revision 01, dated October 3, 2016. This service information describes procedures for cold working the fastener holes in the airframe structure for the door stop fittings installation in FR66 and FR68.

This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section.

Costs of Compliance

We estimate that this AD affects 1,084 airplanes of U.S. registry.

We estimate the following costs to comply with this AD:

ESTIMATED COSTS

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Inspections	23 work-hours × \$85 per hour = \$1,955 per inspection cycle.	\$0	\$1,955 per inspection cycle ..	\$2,119,220 per inspection cycle.

We estimate the following costs to do any necessary repairs that would be

required based on the results of the required inspection. We have no way of

determining the number of aircraft that might need this repair.

ON-CONDITION COSTS

Action	Labor cost	Parts cost	Cost per product
Repair	27 work-hours × \$85 per hour = \$2,295	\$610	\$2,905

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

This AD is issued in accordance with authority delegated by the Executive Director, Aircraft Certification Service, as authorized by FAA Order 8000.51C. In accordance with that order, issuance of ADs is normally a function of the Compliance and Airworthiness Division, but during this transition period, the Executive Director has delegated the authority to issue ADs applicable to transport category airplanes to the Director of the System Oversight Division.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

1. Is not a "significant regulatory action" under Executive Order 12866;

2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);

3. Will not affect intrastate aviation in Alaska; and

4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2018-03-12 Airbus: Amendment 39-19185; Docket No. FAA-2017-0707; Product Identifier 2016-NM-014-AD.

(a) Effective Date

This AD is effective March 19, 2018.

(b) Affected ADs

None.

(c) Applicability

This AD applies to Airbus Model A318-111, -112, -121, and -122 airplanes; Model A319-111, -112, -113, -114, -115, -131, -132, and -133 airplanes; Model A320-211, -212, -214, 231, -232, and -233 airplanes; and Model A321-111, -112, -131, -211, -212, -213, -231, and -232 airplanes; certificated in any category; all manufacturer serial numbers, except airplanes specified in paragraphs (c)(1), (c)(2), and (c)(3) of this AD.

(1) Airplanes on which Airbus modification (Mod) 157039 has been embodied in production.

(2) Model A319 series airplanes on which Mod 28238, Mod 28162, and Mod 28342 have been embodied in production.

(3) Model A318 series airplanes on which Mod 39195 has been embodied in production or Airbus Service Bulletin A320-00-1219 has been embodied in service.

(d) Subject

Air Transport Association (ATA) of America Code 53, Fuselage.

(e) Reason

This AD was prompted by reports of fatigue damage in the structure for the door stop fittings on certain fuselage frames (FR). We are issuing this AD to detect and correct cracking at the door stop fitting holes of fuselage FR66 and FR68. Such cracking could result in reduced structural integrity of the airplane due to the failure of structural components.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Repetitive Rototest Inspections

Within the applicable compliance times specified in table 1 to paragraphs (g) and (j) of this AD and table 2 to paragraphs (g) and (j) of this AD: Do a rototest inspection of all holes below each door stop fitting at fuselage FR66 and FR68, both left-hand (LH) and right-hand (RH) sides, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-53-1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016; or using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). Repeat the inspections thereafter at the applicable compliance times specified in table 1 to paragraphs (g) and (j) of this AD and table 2 to paragraphs (g) and (j) of this AD, until the modification specified in paragraph (i) of this AD is done. Where the "Threshold" column of table 1 to paragraphs (g) and (j) of this AD and table 2 to paragraphs (g) and (j) of this AD, specifies compliance times in "FC" (flight cycles), those compliance times are total flight cycles since the first flight of the airplane.

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Table 1 to paragraphs (g) and (j) of this AD – Aft passenger/crew door cut-out door stop fittings holes at FR 66 WEB LH/RH

Airplanes affected	Threshold	Interval (not to exceed)
A318-PAX (A318-passenger)	Before 33,800 FC	5,900 FC
A319-PAX pre-mod 160001 and pre-mod 160080	Before 42,700 FC	7,500 FC
A319-PAX post-mod 160001 OR A319-PAX post-mod 160080	Before 40,300 FC	7,200 FC
A320 pre-mod 160001 and pre-mod 160080	Before 48,000 FC	9,700 FC
A320 post-mod 160001 OR A320 post-mod 160080	Before 45,000 FC	7,800 FC
A321 pre-mod 160021	Before 34,500 FC or within 30 days after the effective date of this AD, whichever is later without exceeding the accumulation of 42,300 FC since first flight	17,000 FC
A321 post-mod 160021	39,400 FC	8,500 FC

Table 2 to paragraphs (g) and (j) of this AD - Aft passenger/crew door cut-out door stop fittings holes at FR68 WEB LH/RH

Airplanes affected	Threshold	Interval (not to exceed)
A318-PAX	Before 30,800 FC	5,900 FC
A319-PAX pre-mod 160001 and pre-mod 160080	Before 34,400 FC	7,500 FC
A319-PAX post-mod 160001 OR A319-PAX post-mod 160080	Before 33,500 FC	7,200 FC

Airplanes affected	Threshold	Interval (not to exceed)
A320	Before 40,900 FC	9,700 FC
	Before 24,400 FC or within 30 days after the effective date of this AD, whichever is later, without exceeding the accumulation of 39,300 FC since first flight	
A321 pre-mod 160021	Before 39,300 FC	13,600 FC
A321 post-mod 160021	Before 39,300 FC	8,500 FC

BILLING CODE 4910-13-C**(h) Airworthiness Limitations Item (ALI) Inspections Accomplished Before the Effective Date of This AD**

Inspections accomplished as specified in ALI task 534129 or task 534130 before the effective date of this AD are acceptable for compliance with the initial inspection required by paragraph (g) of this AD. As of 30 days after the effective date of this AD, repetitive inspections must be continued as required by paragraph (g) of this AD.

(i) Optional Modification

For airplanes on which no cracks were detected during any rototest inspection required by paragraph (g) of this AD: Modifying the affected area by cold working the fastener holes before further flight after no cracks were detected, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-53-1290, Revision 01, dated October 3, 2016, terminates the repetitive inspections required by paragraph (g) of this AD for the modified area only.

(j) Post-Modification Repetitive Inspections

For airplanes on which the modification specified in paragraph (i) of this AD has been done: At the compliance time specified in paragraphs (j)(1), (j)(2), or (j)(3) of this AD, as applicable, accomplish a rototest inspection of all holes at the door stop fitting locations at fuselage FR66 and FR68, both LH and RH sides, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-53-1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016; or using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). Repeat the inspection thereafter at intervals not to exceed the applicable compliance times specified in table 1 to paragraphs (g) and (j) of this AD and table 2 to paragraphs (g) and (j) of this AD.

(1) For airplanes with less than 1,800 flight cycles accumulated since first flight of the airplane at the time of accomplishing the modification specified in paragraph (i) of this

AD: At the applicable initial compliance time specified in table 1 to paragraphs (g) and (j) of this AD and table 2 to paragraphs (g) and (j) of this AD.

(2) For airplanes with 1,800 flight cycles or more and less than 13,800 flight cycles accumulated since first flight of the airplane at the time of accomplishing the modification specified in paragraph (i) of this AD: Before the accumulation of 48,000 flight cycles since first flight of the airplane.

(3) For airplanes with 13,800 flight cycles or more accumulated since first flight of the airplane at the time of accomplishing the modification specified in paragraph (i) of this AD: Before the accumulation of 60,000 flight cycles since first flight of the airplane.

(k) Repair

If, during any inspection required by paragraph (g) or (j) of this AD, any crack is detected, before further flight, repair using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA. Repair of an airplane as required by this paragraph does not constitute terminating action for the repetitive inspections required by paragraph (g) or (j) of this AD for that airplane, unless specified otherwise in instructions obtained using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA.

(l) Post-Repair Actions for Certain Airplanes

For an airplane that has been inspected as specified in ALI task 534129 or task 534130 and repaired before the effective date of this AD as specified in the applicable structural repair manual (SRM) or as specified in an Airbus repair design approval sheet (RDAS): Comply with the requirements of paragraphs (l)(1) and (l)(2) of this AD.

(1) For all fastener holes where no damage or cracks were detected (*i.e.*, those not repaired), accomplish the actions required by paragraph (g) of this AD, unless the terminating action specified in paragraph (m) of this AD has been done.

(2) For all repaired fastener holes: Within 30 days after the effective date of this AD, or within a compliance time approved by the

Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA, whichever occurs later, contact the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA; for inspection instructions and applicable corrective actions, and do the inspections and applicable corrective actions accordingly.

(m) Terminating Action for Certain Airplanes

For airplanes that have been inspected, as specified in ALI task 534129 or task 534130, and repaired before the effective date of this AD, as specified in the applicable SRM, or as specified in an Airbus RDAS: Modification of the four fastener holes at door stop locations where no damage or crack was detected (*i.e.*, door stop locations not repaired) by cold working holes before further flight after no cracks were detected, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-53-1290, Revision 01, dated October 3, 2016, constitutes terminating action for the repetitive inspections of those four fastener holes at those door stop locations as required by paragraphs (g) or (l)(1) of this AD for that airplane.

(n) Actions for Airplanes With Certain Repairs

For an airplane that has been repaired before the effective date of this AD in the areas described in this AD using an Airbus RDAS unrelated to ALI task 534129 or task 534130: Before exceeding the compliance times specified in paragraph (g) of this AD, contact the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA; for corrective action instructions and accomplish those instructions accordingly. Accomplishment of corrective action(s) on an airplane, as required by this paragraph, does not constitute terminating action for the repetitive inspections as required by paragraphs (g) or (j) of this AD for that airplane, as applicable, unless specified otherwise in the instructions.

(o) Terminating Action for ALI Tasks

(1) Accomplishment of inspections on an airplane, as required by paragraphs (g), (j), or (l) of this AD, as applicable, constitutes terminating action for the inspection requirements of ALI task 534129 or task 534130, as applicable, for that airplane.

(2) Modification of the four fastener holes at a door stop location of an airplane as specified in paragraphs (i) or (m) of this AD, as applicable, and subsequent initial inspection required by paragraph (j) of this AD, constitutes terminating action for the inspection requirements of ALI task 534129 or task 534130, as applicable, for those holes for that airplane. Subsequent repetitive inspections are required by paragraph (j) of this AD.

(p) Credit for Previous Actions

(1) This paragraph provides credit for actions required by paragraphs (g) and (j) of this AD, if those actions were performed before the effective date of this AD using Airbus Service Bulletin A320-53-1288, including Appendixes 01 and 02, dated October 10, 2014.

(2) This paragraph provides credit for actions required by paragraphs (i) and (m) of this AD, if those actions were performed before the effective date of this AD using Airbus Service Bulletin A320-53-1290, dated October 10, 2014.

(q) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, International Section, Transport Standards Branch, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Section, send it to the attention of the person identified in paragraph (r)(2) of this AD. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(2) *Contacting the Manufacturer*: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Section, Transport Standards Branch, FAA; or EASA; or Airbus's EASA DOA. If approved by the DOA, the approval must include the DOA-authorized signature.

(3) *Required for Compliance (RC)*: If any service information contains procedures or tests that are identified as RC, those procedures and tests must be done to comply with this AD; any procedures or tests that are not identified as RC are recommended. Those procedures and tests that are not identified as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided

the procedures and tests identified as RC can be done and the airplane can be put back in an airworthy condition. Any substitutions or changes to procedures or tests identified as RC require approval of an AMOC.

(r) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA AD 2016-0238, dated December 2, 2016; corrected January 4, 2017, for related information. This MCAI may be found in the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0707.

(2) For more information about this AD, contact Sanjay Ralhan, Aerospace Engineer, International Section, Transport Standards Branch, FAA, 1601 Lind Avenue SW, Renton, WA 98057-3356; telephone: 425-227-1405; fax: 425-227-1149.

(3) Service information identified in this AD that is not incorporated by reference is available at the addresses specified in paragraphs (s)(3) and (s)(4) of this AD.

(s) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(i) Airbus Service Bulletin A320-53-1288, Revision 01, including Appendixes 01, 02, and 03, dated October 3, 2016.

(ii) Airbus Service Bulletin A320-53-1290, Revision 01, dated October 3, 2016.

(3) For service information identified in this AD, contact Airbus, Airworthiness Office—ELAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone: +33 5 61 93 36 96; fax: +33 5 61 93 44 51; email: account.airworth-eas@airbus.com; internet: <http://www.airbus.com>.

(4) You may view this service information at the FAA, Transport Standards Branch, 1601 Lind Avenue SW, Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on January 29, 2018.

Michael Kaszycki,

Acting Director, System Oversight Division, Aircraft Certification Service.

[FR Doc. 2018-02359 Filed 2-9-18; 8:45 am]

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DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2017-0901; Product Identifier 2017-NM-106-AD; Amendment 39-19183; AD 2018-03-10]

RIN 2120-AA64

Airworthiness Directives; The Boeing Company Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for all The Boeing Company Model 757-300 series airplanes. This AD was prompted by reports of scribe line damage on fuselage skin. This AD requires detailed inspections of fuselage skin for the presence of scribe lines, and applicable on-condition actions. We are issuing this AD to address the unsafe condition on these products.

DATES: This AD is effective March 19, 2018.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of March 19, 2018.

ADDRESSES: For service information identified in this final rule, contact Boeing Commercial Airplanes, Attention: Contractual & Data Services (C&DS), 2600 Westminister Blvd., MC 110-SK57, Seal Beach, CA 90740-5600; telephone 562-797-1717; internet <https://www.myboeingfleet.com>. You may view this service information at the FAA, Transport Standards Branch, 1601 Lind Avenue SW, Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0901.

Examining the AD Docket

You may examine the AD docket on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0901; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this final rule, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building