

Actions	Compliance	Procedures
(4) Only install, on any affected airplane, a flap control torque tube or flap control torque tube assembly and end plugs that incorporate the part numbers specified in paragraph (d) of this AD.	As of the effective date of this AD	In accordance with sections 3 through 10 of the ACCOMPLISHMENT INSTRUCTIONS section of Piper Mandatory Service Bulletin No. 1051B, dated November 5, 2002.

(e) *Can I comply with this AD in any other way?* You may use an alternative method of compliance or adjust compliance time if:

(1) Your alternative method of compliance provides an equivalent level of safety; and

(2) The Manager, Atlanta Aircraft Certification Office, approves your alternative. Submit your request through an FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Atlanta Aircraft Certification Office.

Note: This AD applies to each airplane identified in paragraph (a) of this AD, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if you have not eliminated the unsafe condition, specific actions you propose to address it.

(f) *Where can I get information about any already-approved alternative methods of compliance?* Contact Hassan Amini, Aerospace Engineer, FAA, Atlanta Aircraft Certification Office, One Crown Center, 1895 Phoenix Boulevard, Suite 450, Atlanta, Georgia 30349; telephone: (770) 703-6080; facsimile: (770) 703-6097.

(g) *What if I need to fly the airplane to another location to comply with this AD?* The FAA can issue a special flight permit under sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate your airplane to a location where you can accomplish the requirements of this AD.

(h) *How do I get copies of the documents referenced in this AD?* You may get copies of the documents referenced in this AD from The New Piper Aircraft, Inc., Customer Services, 2926 Piper Drive, Vero Beach, Florida 32960; telephone: (561) 567-4361; facsimile: (772) 978-6573. You may view these documents at FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri 64106.

Issued in Kansas City, Missouri, on January 15, 2003.

Michael Gallagher,

Manager, Small Airplane Directorate, ,
Aircraft Certification Service.

[FR Doc. 03-1679 Filed 1-24-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 93-CE-37-AD]

RIN 2120-AA64

Airworthiness Directives; Raytheon Aircraft Company Beech Models C35, D35, E35, F35, G35, H35, J35, K35, M35, N35, P35, S35, V35, V35A, and V35B Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes to revise Airworthiness Directive (AD) 94-20-04 R1, which currently requires ruddervator inspections and modifications on certain Raytheon Aircraft Company (Raytheon) Beech Models C35, D35, E35, F35, G35, H35, J35, K35, M35, N35, P35, S35, V35, V35A, and V35B airplanes. This proposed AD is the result of the need to add a repetitive inspection of the fuselage bulkheads and change other inspections from a repetitive to a one-time action. The proposed AD would make these changes to AD 94-20-04 R1. The actions specified by the proposed AD are intended to prevent structural failure of the V-tail, which could result in loss of control of the airplane.

DATES: The Federal Aviation Administration (FAA) must receive any comments on this proposed rule on or before March 3, 2003.

ADDRESSES: Submit comments in triplicate to FAA, Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 93-CE-37-AD, 901 Locust, Room 506, Kansas City, Missouri 64106. Comments may be inspected at this location between 8 a.m. and 4 p.m., Monday through Friday, holidays excepted.

Service information that applies to the proposed AD may be obtained from the Raytheon Aircraft Company, PO Box 85, Wichita, Kansas 67201-0085; telephone: (800) 625-7043 or (316) 676-4556. This information also may be examined at the Rules Docket at the address above.

FOR FURTHER INFORMATION CONTACT: Mr. T.N. Baktha, Aerospace Engineer, FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Mid-Continent Airport, Wichita, Kansas 67209; telephone: (316) 946-4155; facsimile: (316) 946-4407.

SUPPLEMENTARY INFORMATION:

Comments Invited

How Do I Comment on the Proposed AD?

The FAA invites comments on this proposed rule. You may submit whatever written data, views, or arguments you choose. You need to include the rule's docket number and submit your comments in triplicate to the address specified under the caption **ADDRESSES**. The FAA will consider all comments received on or before the closing date. We may amend the proposed rule in light of comments received. Factual information that supports your ideas and suggestions is extremely helpful in evaluating the effectiveness of the proposed AD action and determining whether we need to take additional rulemaking action.

Are There Any Specific Portions of the Proposed AD I Should Pay Attention to?

The FAA specifically invites comments on the overall regulatory, economic, environmental, and energy aspects of the proposed rule that might suggest a need to modify the rule. You may examine all comments we receive before and after the closing date of the rule in the Rules Docket. We will file a report in the Rules Docket that summarizes each FAA contact with the public that concerns the substantive parts of the proposed AD.

How Can I Be Sure FAA Receives My Comment?

If you want us to acknowledge the receipt of your comments, you must include a self-addressed, stamped postcard. On the postcard, write "Comments to Docket No. 93-CE-37-AD." We will date stamp and mail the postcard back to you.

Discussion

Has FAA Taken Any Action to This Point?

On October 15, 2002, FAA issued AD 94-20-04 R1, Amendment 39-12919 (67 FR 64794, October 22, 2002), to require ruddervator inspections and modifications on certain Raytheon Aircraft Company (Raytheon) Beech Models C35, D35, E35, F35, G35, H35, J35, K35, M35, N35, P35, S35, V35, V35A, and V35B airplanes.

The intent of this AD was to maintain the inspection and modification requirements of AD 94-20-04, Amendment 39-9032 (59 FR 49785, September 30, 1994), but condense and clarify the information presented in AD 94-20-04.

What Has Happened Since AD 94-20-04 R1 To Initiate This Action?

Comments from the public since issuance of AD 94-20-04 R1 indicate a need for a revision to that AD. Specifically, the visual inspection of the empennage, aft fuselage, and ruddervator control system with any subsequent repair and the setting of the elevator controls, rudder and tab system controls, cable tensions, and rigging should all be a one time action. Currently, they are to be accomplished repetitively at 100-hour time-in-service (TIS) intervals.

In addition, we inadvertently did not include the 100-hour TIS interval

repetitive inspections of the fuselage bulkheads that were required by AD 94-20-04.

The FAA's Determination and Explanation of the Provisions of the Proposed AD

What Has FAA Decided?

After examining the circumstances and reviewing all available information related to the incidents described above, we have determined that:

- The unsafe condition referenced in this document still exists or could develop on other Raytheon Beech Models C35, D35, E35, F35, G35, H35, J35, K35, M35, N35, P35, S35, V35, V35A, and V35B of the same type design;
- The actions of AD 94-20-04 R1 should be maintained except for adding the fuselage bulkhead inspections and making certain repetitive actions a one-time action; and
- AD action should be taken in order to continue to prevent structural failure of the V-tail, which could result in loss of control of the airplane.

What Would the Proposed AD Require?

This proposed AD would revise AD 94-20-04 R1, which applies to Raytheon Beech Models C35, D35, E35, F35, G35, H35, J35, K35, M35, N35, P35, S35, V35, V35A, and V35B. The proposed AD would:

- Maintain the actions of AD 94-20-04 R1, but would make the repetitive visual inspection of the empennage, aft fuselage, and ruddervator control system with any subsequent repair and the setting of the elevator controls, rudder and tab system controls, cable tensions, and rigging a one time action; and
- Would add repetitive inspections of the fuselage bulkheads that were required by AD 94-20-04.

Cost Impact

How Many Airplanes Would the Proposed AD Impact?

We estimate that the proposed AD affects 10,200 airplanes in the U.S. registry.

What Would Be the Cost Impact of the Proposed AD on Owners/Operators of the Affected Airplanes?

We estimate the following costs to accomplish the proposed initial inspections. These cost figures are exactly the same as what is currently required by AD 94-20-04 R1. Although we are adding the inspection of the fuselage bulkheads to the proposed AD, we had already incorporated the costs of this inspection into the Cost Impact of AD 94-20-04 R1. Therefore, this proposed AD presents no new costs upon the public:

Labor cost	Parts cost	Total cost per airplane	Total cost on U.S. operators
40 workhours at \$60 per hour = \$2,400	Not applicable	\$2,400	\$24,480,000

The above figures are based only on the initial inspections and do not take into account the cost of repetitive inspections or adjustments, repairs, or replacements that would be necessary based on the results of the inspections. We have no way of determining the number of repetitive inspections each owner/operator of the affected airplanes would incur or what adjustments, repairs, or replacements may be necessary based on the results of the inspections.

Regulatory Impact

Would This Proposed AD Impact Various Entities?

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the

various levels of government. Therefore, it is determined that this proposed rule would not have federalism implications under Executive Order 13132.

Would This Proposed AD Involve a Significant Rule or Regulatory Action?

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action has been placed in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the

location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend 14 CFR part 39 of the Federal Aviation Regulations as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. FAA amends § 39.13 by removing Airworthiness Directive (AD) 94-20-04

R1, Amendment 39–12919 (67 FR 64794, October 22, 2002), and by adding a new AD to read as follows:

Raytheon Aircraft Company (Beech Aircraft Corporation formerly held Type Certificate (TC) No. A-777 and TC No. 3A15): Docket No. 93–CE–37–AD; Revises AD 94–20–04 R1, Amendment 39–12919, which revised AD 94–20–04, Amendment 39–9032.

(a) *What airplanes are affected by this AD?* This AD affects the following airplanes that are certificated in any category:

(1) Beech Models C35, D35, E35, F35, G35, H35, J35, K35, M35, N35, and P35 airplanes, all serial numbers; and

(2) Beech Models S35, V35, V35A, and V35B airplanes, all serial numbers, that do not have the straight tail conversion modification incorporated in accordance with Supplemental Type Certificate (STC) SA2149CE.

Note 1: Beech Models 35, 35R, A35, B35 airplanes were included in the Applicability of AD 94–20–04. We have removed Beech Models 35, 35R, A35, and B35 airplanes from the Applicability section of this AD and incorporated the actions applicable to these airplanes into another AD action. Part of this

other AD action is the incorporation of Raytheon Service Raytheon Service Bulletin 27–3358.

(b) *Who must comply with this AD?* Anyone who wishes to operate any of the airplanes identified in paragraphs (a), (a)(1), and (a)(2) of this AD must comply with this AD.

(c) *What problem does this AD address?* The actions specified by this AD are intended to prevent structural failure of the V-tail, which could result in loss of control of the airplane.

(d) *What actions must I accomplish to address this problem?* To address this problem, you must accomplish the following:

Actions	Compliance	Procedures
(1) Verify that the ruddervator balance is within manufacturer's specified limits as defined in the applicable shop or maintenance manual. If the ruddervator is outside the specified limits, balance the ruddervator control surfaces.	Accomplish the verification within the next 100 hours TIS after November 28, 1994 (the effective date of AD 94–20–04) and thereafter prior to further flight after the ruddervators are repaired or repainted (even if stripes are added or paint is touched up). Accomplish the balancing prior to further flight after the ruddervator is found outside the specified limits.	Verify in accordance with the applicable the shop or maintenance manual. Balance the ruddervator control surfaces in accordance with Section 3 of Beech Shop Manual 35–590096B19, or subsequent revisions.
(2) Visually inspect the empennage, aft fuselage, and ruddervator control system for damage. (i) Repair or replace any damaged parts; and (ii) Set the elevator controls, rudder and tab system controls, cable tensions, and rigging.	Inspect and set the controls, tension, and rigging within the next 100 hours TIS after November 28, 1994 (the effective date of AD 94–20–04). Accomplish any repairs and replacements prior to further flight after the inspection.	In accordance with the procedures and as specified in the instructions to Beech Kit 35–4017–1 “Kit Information Empennage and Aft Fuselage Inspection”, as specified in Beech Service Bulletin No. 2188, dated May, 1987.
(3) Accomplish the following actions: (i) Visually inspect the fuselage bulkheads at Fuselage Station (FS) 256.9 and FS 272 for damage (cracks, distortion, loose rivets, etc.); (ii) Visually inspect the fuselage skin around the bulkhead for damage (wrinkles or cracks); and (iii) Repair or replace any damaged parts.	Initially inspect within the 100 hours time-in-procedures after the effective date of this AD. Repetitively inspect thereafter at intervals not to exceed 100 hours TIS. Repair or replace prior to further flight where damage is found.	In accordance with the procedures in the instructions to Beech Kit 35–4017–1 “Kit Information Empennage & Aft Fuselage Inspection”, as specified in Beech SB 2188, dated May 1987.
(4) Remove all external stabilizer reinforcements installed during incorporation of either Supplement Type Certificate (STC) SA845GL, STC SA846GL, STC SA1650CE, STC SA2286NM, or STC SA2287NM, as applicable. (i) Seal or fill any residual holes with appropriate size rivets. (ii) The internal stub spar incorporated through STC SA1649CE and STC SA1650CE may be retained. (iii) The external angles incorporated through STC SA1649CE may also be retained by properly trimming the leading edges section to permit installation of the stabilizer reinforcement referenced in paragraph (d)(5)(i) of this AD. (iv) For the Beech Models S35, V35, V35A, and V35B airplanes, you may retain and use the tail-safe external angles that were installed in accordance with STC SA1649CE instead of the stabilizer reinforcement specified in paragraph (d)(5)(i) of this AD.	Within the next 100 hours TIS after November 28, 1994 (the effective date of AD 94–20–04), unless already accomplished.	In accordance with the applicable maintenance information.
(5) Accomplish the following: (i) Install stablizer reinforcements; (ii) Set the elevator nosedown trim; and (iii) Replace the ruddervator tab control cables with larger diameter cables.	Within the next 100 hours TIS after November 28, 1994 (the effective date of AD 94–20–04), unless already accomplished.	In accordance with RAC Kit No. 35–4016–3, 35–4016–5, 35–4016–7, or 35–4016–9, as applicable and as specified in Beech SB No. 2188, dated May, 1987.

Actions	Compliance	Procedures
(6) Verify the accuracy of the airplane basic weight and balance information and correct any discrepancies.	Accomplish the airplane basic weight and balance accuracy verification within the next 100 hours TIS after November 28, 1994 (the effective date the of AD 94-20-04), unless already accomplished. Correct any discrepancies. prior to further flight after the verification.	Use the procedures contained in the Appendix to this AD.

(e) *Can I comply with this AD in any other way?*

(1) You may use an alternative method of compliance or adjust the compliance time if:

(i) Your alternative method of compliance provides an equivalent level of safety; and

(ii) The Manager, Wichita Aircraft Certification Office (ACO), approves your alternative. Submit your request through an FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Wichita ACO.

(2) Alternative methods of compliance approved in accordance with AD 94-20-04 R1 or AD 94-20-04 are approved as alternative methods of compliance with this AD.

Note 2: This AD applies to each airplane identified in paragraphs (a), (a)(1), and (a)(2) of this AD, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if you have not eliminated the unsafe condition, specific actions you propose to address it.

(f) *Where can I get information about any already-approved alternative methods of compliance?* Contact Mr. T.N. Baktha, Aerospace Engineer, FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Mid-Continent Airport, Wichita, Kansas 67209; telephone: (316) 946-4155; facsimile: (316) 946-4407.

(g) *What if I need to fly the airplane to another location to comply with this AD?* The FAA can issue a special flight permit under sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate your airplane to a location where you can accomplish the requirements of this AD.

(h) *How do I get copies of the documents referenced in this AD?* You may obtain copies of the documents referenced in this AD from the Raytheon Aircraft Company, PO Box 85, Wichita, Kansas 67201-0085. You may examine these documents at FAA, Central Region, Office of the Regional Counsel,

901 Locust, Room 506, Kansas City, Missouri 64106.

(i) *Does this AD action affect any existing AD actions?* This amendment revises AD 94-20-04 R1, Amendment 39-12919.

Appendix to Docket No. 93-CE-37-AD

Weight and Balance Accuracy Method No. 1

1. Review existing weight and balance documentation to assure completeness and accuracy of the documentation from the most recent FAA-approved weighing or from factory delivery to date of compliance with this AD.

2. Compare the actual configuration of the airplane to the configuration described in the weight and balance documentation.

3. If equipment additions or deletions are not reflected in the documentation or if modifications affecting the location of the center of gravity (e.g., paint or structural repairs) are not documented, determine the accuracy of the airplane weight and balance data in accordance with Method No. 2.

Weight and Balance Information Accuracy Method No. 2

1. Determine the basic empty weight and center of gravity (CG) of the empty airplane using the Weighing Instructions in the Weight and Balance section of the airplane flight manual/pilot's operating handbook (AFM/POH).

2. Record the results in the airplane records, and use these new values as the basis for computing the weight and CG information as specified in the Weight and Balances section of the AFM/POH.

Issued in Kansas City, Missouri, on January 15, 2003.

Michael Gallagher,

Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-1678 Filed 1-24-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-CE-56-AD]

RIN 2120-AA64

Airworthiness Directives; British Aerospace Model HP.137 Jetstream Mk.1, Jetstream Series 200, Jetstream Series 3101, and Jetstream Model 3201 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes to adopt a new airworthiness directive (AD) that would apply to all British Aerospace Model HP.137 Jetstream Mk.1, Jetstream Series 200, Jetstream Series 3101, and Jetstream Model 3201 airplanes. This proposed AD would require you to inspect the steering jack piston rod for cracks and replace if necessary; measure the torque setting of the steering jack piston rod end fitting and stop bolt; and measure the thickness of the tab washers. This proposed AD would also require you to calculate a new safe life limit for the steering jack piston rod based on the results of the proposed inspection and the proposed measurements. This proposed AD is the result of mandatory continuing airworthiness information (MCAI) issued by the airworthiness authority for the United Kingdom. The actions specified by this proposed AD are intended to detect, correct, and prevent cracks in the steering jack piston rod, which could result in failure of the steering jack piston rod. Such failure could lead to loss of steering control of the airplane during takeoff, landing, and taxi operations.

DATES: The Federal Aviation Administration (FAA) must receive any comments on this proposed rule on or before February 28, 2003.

ADDRESSES: Submit comments to FAA, Central Region, Office of the Regional Counsel, Attention: Rules Docket No. 2002-CE-56-AD, 901 Locust, Room