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FOR FURTHER INFORMATION CONTACT:

Stewart Schneider, Office of Nuclear Material Safety and Safeguards, telephone: 301-415-4123, 3453, email: Stewart.Schneider@nrc.gov, or On Yee, Office of Nuclear Reactor Regulation, telephone: 301-415-1905, email: On.Yee@nrc.gov. Both are staff of the U.S. Nuclear Regulatory Commission, Washington, DC 20555-0001.

SUPPLEMENTARY INFORMATION: On October 2, 2020 (85 FR 62199), the NRC published a direct final rule amending its regulations in appendix H, "Reactor Vessel Material Surveillance Program Requirements" (appendix H), to part 50 of title 10 of the *Code of Federal Regulations* (10 CFR), "Domestic Licensing of Production and Utilization Facilities," to revise the NRC's reactor vessel material surveillance program requirements for commercial light-water reactors. The direct final rule revises the requirements in appendix H to 10 CFR part 50 associated with the testing of specimens contained within surveillance capsules and reporting the surveillance test results. The direct final rule also clarifies the requirements for the design of surveillance programs and the capsule withdrawal schedules for surveillance capsules in reactor vessels purchased after 1982.

In the direct final rule published on October 2, 2020, the NRC stated that if no significant adverse comments were received, the direct final rule would become effective on February 1, 2021. The NRC received and docketed two comment submissions on the companion proposed rule (85 FR 62234; October 2, 2020). Electronic copies of the comments can be obtained from the Federal Rulemaking website at <https://www.regulations.gov> under Docket ID NRC-2017-0151 and are also available in ADAMS under Accession Nos. ML20301A624 and ML20308A229, respectively.

The NRC determined that the two comment submissions addressed issues that were outside the scope of the direct final rule or were not significant nor adverse. One comment submission questioned (1) the use of Charpy V-notch testing procedures, (2) the subsequent license renewal and extended power uprate for Turkey Point Nuclear Generating Units 3 and 4, (3) the use of Integrated Surveillance Programs, and (4) the timing associated with specimen testing and capsule report submittal. With respect to the items 1-3, these issues are outside the scope of this direct final rule. While item 4 is within the scope of the direct final rule, the NRC determined that the comment was not significant nor adverse. The other comment submission is outside the scope of this direct final rule. Therefore the direct final rule will become effective as scheduled.

Paperwork Reduction Act Statement

The direct final rule contains a new or amended collection of information subject to the Paperwork Reduction Act of 1995 (44 U.S.C. 3501 *et seq.*). The collections of information were approved by the Office of Management and Budget (OMB), approval number 3150-0011.

Public Protection Notification

The NRC may not conduct or sponsor, and a person is not required to respond to a collection of information unless the document requesting or requiring the collection displays a currently valid OMB control number.

Dated: December 21, 2020.

For the Nuclear Regulatory Commission.

Pamela J. Shepherd-Vladimir,

Acting Chief, Regulatory Analysis and Rulemaking Support Branch, Division of Rulemaking, Environmental, and Financial Support, Office of Nuclear Material Safety and Safeguards.

[FR Doc. 2020-28814 Filed 12-28-20; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2020-0858; Project Identifier MCAI-2020-00949-T; Amendment 39-21370; AD 2020-26-15]

RIN 2120-AA64

Airworthiness Directives; Airbus SAS Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: The FAA is superseding Airworthiness Directive (AD) 2016-07-14, which applied to certain Airbus Model A319-111, -112, -113, -114, -115, -131, -132, and -133 airplanes; Model A320-211, -212, -214, -231, -232, and -233 airplanes; and Model A321-111, -112, -131, -211, -212, -213, -231, and -232 airplanes. AD 2016-07-14 required replacing the clips, shear webs, and angles, related investigative actions, and repair if necessary. This AD retains the actions of AD 2016-07-14, and requires modifying (replacing) the clips, shear webs, and angles at a certain rear fuselage area with new parts, as specified in a European Union Aviation Safety Agency (EASA) AD, which is incorporated by reference. The FAA has also determined that additional airplanes are subject to the unsafe condition. This AD was prompted by fatigue testing that determined that fatigue damage could appear on clips, shear webs, and angles at certain rear fuselage sections and certain frames. The FAA is issuing this AD to address the unsafe condition on these products.

DATES: This AD is effective February 2, 2021.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of February 2, 2021.

ADDRESSES: For material incorporated by reference (IBR) in this AD, contact the EASA, Konrad-Adenauer-Ufer 3, 50668 Cologne, Germany; telephone +49 221 8999 000; email ADs@easa.europa.eu; internet www.easa.europa.eu. You may find this IBR material on the EASA website at <https://ad.easa.europa.eu>. You may view this IBR material at the FAA, Airworthiness Products Section, Operational Safety Branch, 2200 South 216th St., Des Moines, WA. For information on the availability of this material at the FAA, call 206-231-3195.

It is also available in the AD docket on the internet at <https://www.regulations.gov> by searching for and locating Docket No. FAA–2020–0858.

Examining the AD Docket

You may examine the AD docket on the internet at <https://www.regulations.gov> by searching for and locating Docket No. FAA–2020–0858; or in person at Docket Operations between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this final rule, any comments received, and other information. The address for Docket Operations is U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE, Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Sanjay Ralhan, Aerospace Engineer, Large Aircraft Section, International Validation Branch, FAA, 2200 South 216th St., Des Moines, WA 98198; telephone and fax 206–231–3223; email Sanjay.Ralhan@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

The EASA, which is the Technical Agent for the Member States of the European Union, has issued EASA AD 2020–0153, dated July 10, 2020 (EASA AD 2020–0153) (also referred to as the Mandatory Continuing Airworthiness Information, or the MCAI), to correct an unsafe condition for certain Airbus Model A319–111, –112, –113, –114, –115, –131, –132, and –133 airplanes; Model A320–211, –212, –214, –215 –216, –231, –232, and –233 airplanes; and Model A321–111, –112, –131, –211, –212, –213, –231, and –232 airplanes.

EASA AD 2020–0153 superseded EASA AD 2014–0177, dated July 25, 2014 (which corresponds to FAA AD 2016–07–14, 39–18459 [81 FR 21244, April 11, 2016] (AD 2016–07–14)). Model A320–215 airplanes are not certificated by the FAA and are not included on the U.S. type certificate data sheet; this AD therefore does not include those airplanes in the applicability.

The FAA issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to supersede AD 2016–07–14. AD 2016–07–14 applied to certain Airbus Model A319–111, –112, –113, –114, –115, –131, –132, and –133 airplanes; Model A320–211, –212, –214, –231, –232, and –233 airplanes; and Model A321–111, –112, –131, –211, –212, –213, –231, and –232 airplanes. The NPRM published in the **Federal Register** on October 1, 2020 (85 FR 61892). The NPRM was prompted by fatigue testing that determined that fatigue damage could appear on clips, shear webs, and angles at certain rear fuselage sections and certain frames. The NPRM proposed to retain the actions of AD 2016–07–14, and require modifying (replacing) the clips, shear webs, and angles at a certain rear fuselage area with new parts, as specified in EASA AD 2020–0153. The NPRM also proposed to apply to additional airplanes subject to the unsafe condition.

The FAA is issuing this AD to address fatigue damage on the clips, shear webs, and angles, which could affect the structural integrity of the airplane. See the MCAI for additional background information.

Comments

The FAA gave the public the opportunity to participate in developing

this final rule. The FAA has considered the comments received. United Airlines and an anonymous commenter indicated their support for the NPRM.

Conclusion

The FAA reviewed the relevant data, considered the comments received, and determined that air safety and the public interest require adopting this final rule as proposed, except for minor editorial changes. The FAA has determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM for addressing the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

Related Service Information Under 1 CFR Part 51

EASA AD 2020–0153 describes procedures for replacement of affected parts (as required by FAA AD 2016–07–14). EASA AD 2020–0153 also describes procedures for a modification by replacing the clips, shear webs, and angles at the rear fuselage area of section 19 at frame 72 and frame 74 with new parts without pilot holes, and installing oversized Hi-Loks, nominal aluminum rivets, and nominal Hi-Loks in certain positions. This material is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

Costs of Compliance

The FAA estimates that this AD affects 219 airplanes of U.S. registry. The FAA estimates the following costs to comply with this AD:

ESTIMATED COSTS FOR REQUIRED ACTIONS

Action	Labor cost	Parts cost	Cost per product	Cost on U.S. operators
Retained actions from AD 2016–07–14 (for 44 airplanes affected).	Up to 110 work-hours × \$85 per hour = Up to \$9,350.	\$10,000	Up to \$19,350 ..	Up to \$851,400.
New actions	126 work-hours × \$85 per hour = \$10,710	51,750	62,460	13,678,740.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

The FAA is issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section

44701: General requirements. Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

(1) Is not a “significant regulatory action” under Executive Order 12866,
(2) Will not affect intrastate aviation in Alaska, and

(3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by:

■ a. Removing Airworthiness Directive (AD) 2016–07–14, Amendment 39–18459 (81 FR 21244, April 11, 2016), and

■ b. Adding the following new AD:

2020–26–15 Airbus SAS: Amendment 39–21370; Docket No. FAA–2020–0858; Project Identifier MCAI–2020–00949–T.

(a) Effective Date

This airworthiness directive (AD) is effective February 2, 2021.

(b) Affected ADs

This AD replaces AD 2016–07–14, Amendment 39–18459 (81 FR 21244, April 11, 2016) (AD 2016–07–14).

(c) Applicability

This AD applies to Airbus SAS airplanes specified in paragraphs (c)(1) through (3) of this AD, certificated in any category, as identified in European Union Aviation Safety Agency (EASA) AD 2020–0153, dated July 10, 2020 (EASA AD 2020–0153).

(1) Model A319–111, –112, –113, –114, –115, –131, –132, and –133 airplanes.

(2) Model A320–211, –212, –214, –216, –231, –232, and –233 airplanes.

(3) Model A321–111, –112, –131, –211, –212, –213, –231, and –232 airplanes.

(d) Subject

Air Transport Association (ATA) of America Code 53, Fuselage.

(e) Reason

This AD was prompted by fatigue testing that determined that fatigue damage could appear on clips, shear webs, and angles at certain rear fuselage sections and certain frames. The FAA is issuing this AD to address fatigue damage on the clips, shear

webs, and angles, which could affect the structural integrity of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Requirements

Except as specified in paragraph (h) of this AD: Comply with all required actions and compliance times specified in, and in accordance with, EASA AD 2020–0153.

(h) Exceptions to EASA AD 2020–0153

The “Remarks” section of EASA AD 2020–0153 does not apply to this AD.

(i) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, Large Aircraft Section, International Validation Branch, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or responsible Flight Standards Office, as appropriate. If sending information directly to the Large Aircraft Section, International Validation Branch, send it to the attention of the person identified in paragraph (j) of this AD. Information may be emailed to: 9-AVS-AIR-730-AMOC@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the responsible Flight Standards Office.

(2) *Contacting the Manufacturer:* For any requirement in this AD to obtain instructions from a manufacturer, the instructions must be accomplished using a method approved by the Manager, Large Aircraft Section, International Validation Branch, FAA; or EASA; or Airbus SAS’s EASA Design Organization Approval (DOA). If approved by the DOA, the approval must include the DOA-authorized signature.

(3) *Required for Compliance (RC):* Except as required by paragraph (i)(2) of this AD, if any service information contains procedures or tests that are identified as RC, those procedures and tests must be done to comply with this AD; any procedures or tests that are not identified as RC are recommended. Those procedures and tests that are not identified as RC may be deviated from using accepted methods in accordance with the operator’s maintenance or inspection program without obtaining approval of an AMOC, provided the procedures and tests identified as RC can be done and the airplane can be put back in an airworthy condition. Any substitutions or changes to procedures or tests identified as RC require approval of an AMOC.

(j) Related Information

For more information about this AD, contact Sanjay Ralhan, Aerospace Engineer, Large Aircraft Section, International Validation Branch, FAA, 2200 South 216th St., Des Moines, WA 98198; telephone and fax 206–231–3223; email Sanjay.Ralhan@faa.gov.

(k) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(i) European Union Aviation Safety Agency (EASA) AD 2020–0153, dated July 10, 2020.

(ii) [Reserved]

(3) For EASA AD 2020–0153, contact the EASA, Konrad-Adenauer-Ufer 3, 50668 Cologne, Germany; telephone +49 221 8999 000; email ADs@easa.europa.eu; internet www.easa.europa.eu. You may find this EASA AD on the EASA website at <https://ad.easa.europa.eu>.

(4) You may view this material at the FAA, Airworthiness Products Section, Operational Safety Branch, 2200 South 216th St., Des Moines, WA. For information on the availability of this material at the FAA, call 206–231–3195. This material may be found in the AD docket on the internet at <https://www.regulations.gov> by searching for and locating Docket No. FAA–2020–0858.

(5) You may view this material that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, email fedreg.legal@nara.gov, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued on December 11, 2020.

Lance T. Gant, Director,
Compliance & Airworthiness Division,
Aircraft Certification Service.

[FR Doc. 2020–28858 Filed 12–28–20; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2020–0781; Product Identifier 2018–CE–045–AD; Amendment 39–21369; AD 2020–26–14]

RIN 2120–AA64

Airworthiness Directives; Mitsubishi Heavy Industries, Ltd. Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: The FAA is superseding Airworthiness Directive (AD) 75–16–20, which applied to all Mitsubishi Heavy Industries, Ltd., Model MU–2B, MU–2B–10, MU–2B–15, MU–2B–20, MU–2B–25, MU–2B–26, MU–2B–30, MU–2B–35, and MU–2B–36 airplanes. AD 75–16–20 required repetitive inspections of the propeller pitch control (PPC) lever for security and proper rigging. This AD requires