

specified in paragraphs (h)(1) and (h)(2) of this AD, as applicable.

(1) For Model 737-600, -700, and -800 series airplanes identified in Boeing Service Bulletin 737-78-1063, Revision 2, dated October 7, 1999: Replace the inner wall blanket insulation, in accordance with the Accomplishment Instructions of Boeing Service Bulletin 737-78-1063, Revision 2, dated October 7, 1999.

(2) For airplanes equipped with engines identified in CFM CFM56-7B Service Bulletin 73-0135, dated March 30, 2007: Install updated full-authority digital electronic control (FADEC) software, in accordance with the Accomplishment Instructions of CFM CFM56-7B Service Bulletin 73-0135, dated March 30, 2007.

Inspection/Measurement

(i) At the applicable time specified in paragraph (j) of this AD: Do the actions specified in paragraphs (i)(1), (i)(2), and (i)(3) of this AD, in accordance with the Accomplishment Instructions of Boeing Service Bulletin 737-78-1088, dated May 12, 2010. If any damage or discrepancy is found, before further flight, do all applicable corrective actions, in accordance with Accomplishment Instructions of Boeing Service Bulletin 737-78-1088, dated May 12, 2010; except as required by paragraph (k) of this AD; and except where the service bulletin refers to "unsatisfactory" findings, this AD assumes those parts or locations are "unserviceable."

(1) Do a detailed inspection for damage of the engine side and inner wall side of the upper and lower insulation blankets.

(2) Measure the electrical conductivity on the aluminum upper compression pads 2 and 3, as applicable.

(3) Inspect for discrepancies of the thrust reverser inner wall (including an ultrasonic inspection for interplay delamination and skin-to-core disbond, a detailed inspection for signs of heat damage as applicable, and a detailed inspection for loose fasteners where the inner wall attaches to the hinge beam and at the fasteners for the compression pads).

Compliance Times for Paragraph (i)

(j) Do the actions specified in paragraph (i) of this AD at the applicable time specified in paragraph (j)(1), (j)(2), (j)(3), (j)(4), or (j)(5) of this AD.

(1) For airplanes with thrust reverser part number (P/N) 315A2295-003 through 315A2295-154 inclusive: Do the actions within 30 months after the effective date of this AD.

(2) For airplanes with thrust reverser P/N 315A2295-155 through 315A2295-174 inclusive: Do the actions within 60 months after the effective date of this AD.

(3) For airplanes with thrust reverser P/N 315A2295-175 through 315A2295-190 inclusive: Do the actions within 72 months after the effective date of this AD.

(4) For airplanes with thrust reverser P/N 315A2295-191 through 315A2295-198 inclusive: Do the actions within 84 months after the effective date of this AD.

(5) For airplanes with thrust reverser P/N 315A2295-199 through 315A2295-202

inclusive: Do the actions within 96 months after the effective date of this AD.

Exception to Boeing Service Bulletin 737-78-1088 Procedures

(k) Where Boeing Service Bulletin 737-78-1088, dated May 12, 2010, specifies to contact Boeing for appropriate action, repair before further flight in accordance with a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

Actions Concurrent With Paragraph (i)

(l) For airplanes identified in Boeing Service Bulletin 737-78-1069, Revision 4, dated June 16, 2005: Before or concurrently with the accomplishment of the requirements of paragraph (i) of this AD, modify the thrust reverser inner wall and insulation blankets, in accordance with the Accomplishment Instructions of Boeing Service Bulletin 737-78-1069, Revision 4, dated June 16, 2005. A modification done before the effective date of this AD is also acceptable if done in accordance with Boeing Service Bulletin 737-78-1069, Revision 1, dated June 13, 2002; Revision 2, dated February 6, 2003; or Revision 3, dated August 5, 2004.

(m) If the actions required by paragraph (i) of this AD are done before the compliance time specified in paragraph (g) of this AD: Before or concurrently with the accomplishment of the actions required by paragraph (i) of this AD, the modification required by paragraph (g) of this AD must be done.

Option to Requirements of Paragraphs (g) and (i)

(n) Accomplishment of all of the actions (including inspections and modification) specified in Boeing Service Bulletin 737-78-1079, Revision 2, dated June 7, 2010, within 24 months after the effective date of this AD, is acceptable for compliance with the requirements of paragraphs (g) and (i) of this AD and is acceptable for compliance with the requirements of this AD, provided applicable repairs are done before further flight, and provided the applicable actions specified in paragraphs (h)(1), (h)(2), and (l) of this AD have been done. Actions done before the effective date of this AD in accordance with Boeing Service Bulletin 737-78-1079, dated August 6, 2007; or Revision 1, dated December 17, 2007; are also acceptable for compliance with the corresponding actions specified in this paragraph.

Alternative Methods of Compliance (AMOCs)

(o)(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Chris Parker, Aerospace Engineer, Propulsion Branch, ANM-140S, FAA, Seattle Aircraft Certification Office (ACO), 1601 Lind Avenue, SW., Renton, Washington 98057-3356; telephone 425-917-6496; fax 425-917-6590. Information may be e-mailed to: 9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your principal maintenance inspector (PMI) or principal avionics inspector (PAI), as appropriate, or lacking a principal inspector, your local Flight Standards District Office. The AMOC approval letter must specifically refer to this AD.

Issued in Renton, Washington on September 15, 2010.

Robert D. Breneman,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2010-24175 Filed 9-24-10; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2010-0942; Directorate Identifier 2010-CE-049-AD]

RIN 2120-AA64

Airworthiness Directives; British Aerospace Regional Aircraft Models Jetstream Series 3101 and Jetstream Model 3201 Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for the products listed above. This proposed AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

As a result of the fatigue-testing programme on the Jetstream fatigue test specimen, it has been identified that failure of the undercarriage jack mounting shaft assembly can occur.

This condition, if not corrected, could lead to a Main Landing Gear (MLG) collapse on the ground or during landing and consequently damage to the aeroplane or injury to the occupants.

The proposed AD would require actions that are intended to address the unsafe condition described in the MCAI.

DATES: We must receive comments on this proposed AD by November 12, 2010.

ADDRESSES: You may send comments by any of the following methods:

• *Federal eRulemaking Portal*: Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

• *Fax*: (202) 493-2251.

• *Mail*: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

• *Hand Delivery*: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone (800) 647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT:

Taylor Martin, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; *telephone*: (816) 329-4138; *fax*: (816) 329-4090; *e-mail*: taylor.martin@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA-2010-0942; Directorate Identifier 2010-CE-049-AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

The European Aviation Safety Agency (EASA), which is the Technical Agent

for the Member States of the European Community, has issued AD No.: 2010-0162, dated August 4, 2010 (referred to after this as “the MCAI”), to correct an unsafe condition for the specified products. The MCAI states:

As a result of the fatigue-testing programme on the Jetstream fatigue test specimen, it has been identified that failure of the undercarriage jack mounting shaft assembly can occur.

This condition, if not corrected, could lead to a Main Landing Gear (MLG) collapse on the ground or during landing and consequently damage to the aeroplane or injury to the occupants.

BAE SYSTEMS have now defined safe life limits for these components.

For the reasons described above, this AD requires the application of safe life limits to these components.

You may obtain further information by examining the MCAI in the AD docket.

Relevant Service Information

BAE Systems has issued British Aerospace Jetstream Series 3100 & 3200 Service Bulletin 05-JA090143, dated April 30, 2009; and British Aerospace Regional Aircraft has issued British Aerospace Jetstream Series 3100 & 3200 Service Bulletin 32-JA990142, dated March 26, 1999. The actions described in this service information are intended to correct the unsafe condition identified in the MCAI.

FAA’s Determination and Requirements of the Proposed AD

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our bilateral agreement with this State of Design Authority, they have notified us of the unsafe condition described in the MCAI and service information referenced above. We are proposing this AD because we evaluated all information and determined the unsafe condition exists and is likely to exist or develop on other products of the same type design.

Differences Between This Proposed AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have proposed different actions in this AD from those in the MCAI in order to follow FAA

policies. Any such differences are highlighted in a NOTE within the proposed AD.

Costs of Compliance

We estimate that this proposed AD will affect 80 products of U.S. registry. We also estimate that it would take about 15 work-hours per product to comply with the basic requirements of this proposed AD. The average labor rate is \$85 per work-hour. Required parts would cost about \$10,000 per product.

Based on these figures, we estimate the cost of the proposed AD on U.S. operators to be \$902,000, or \$11,275 per product.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with

this proposed AD and placed it in the AD docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by adding the following new AD:

British Aerospace Regional Aircraft: Docket No. FAA-2010-0942; Directorate Identifier 2010-CE-049-AD.

Comments Due Date

(a) We must receive comments by November 12, 2010.

Affected ADs

(b) None.

Applicability

(c) This AD applies to British Aerospace Regional Aircraft Models Jetstream Series 3101 and Jetstream Model 3201 airplanes, all serial numbers, certificated in any category.

Subject

(d) Air Transport Association of America (ATA) Code 32: Landing Gear.

Reason

(e) The mandatory continuing airworthiness information (MCAI) states:

As a result of the fatigue-testing programme on the Jetstream fatigue test specimen, it has been identified that failure of the undercarriage jack mounting shaft assembly can occur.

This condition, if not corrected, could lead to a Main Landing Gear (MLG) collapse on the ground or during landing and consequently damage to the aeroplane or injury to the occupants.

BAE SYSTEMS have now defined safe life limits for these components.

For the reasons described above, this AD requires the application of safe life limits to these components.

Actions and Compliance

(f) Unless already done, do the following actions:

(1) Within 30 days after the effective date of this AD, establish the number of landings accumulated since installation of each left and right main landing gear radius rod mounting shaft assemblies following paragraph 2.(A) of BAE Systems British Aerospace Jetstream Series 3100 & 3200

Service Bulletin 05-JA090143, dated April 30, 2009.

(2) Replace the left and right main landing gear radius rod mounting shaft assembly with an airworthy assembly following British Aerospace Jetstream Series 3100 & 3200 Service Bulletin 32-JA990142, dated March 26, 1999, within the following:

(i) *For Model Jetstream Series 3101:* Within 38,220 total landings accumulated on each main landing gear radius rod mounting shaft assembly or within 1,000 landings after the effective date of this AD, whichever occurs later; and

(ii) *For Model Jetstream Model 3201:* Within 31,038 total landings accumulated on each main landing gear radius rod mounting shaft assembly or within 1,000 landings after the effective date of this AD, whichever occurs later.

(3) After replacing each main landing gear radius rod mounting shaft assembly as required by paragraph (f)(2) of this AD, repetitively thereafter replace each assembly with an airworthy assembly at intervals not to exceed the following life limits:

(i) *For Model Jetstream Series 3101:* Within 38,220 total landings; and

(ii) *For Model Jetstream Model 3201:* Within 31,038 total landings.

(4) For operators that do not have landing records, determine the number of landings by multiplying the number of hours time-in-service (TIS) accumulated on each main landing gear radius rod mounting shaft assembly by 0.75. For the purpose of this AD:

(i) 1,000 landings equals 1,333 hours TIS;

(ii) 31,038 landings equals 41,384 hours TIS; and

(iii) 38,220 landings equals 50,960 hours TIS.

(5) Compliance with the life limits set in paragraph (f)(3) of this AD may be done by incorporating these limits into the limitations section of the aircraft maintenance manual. You may do this by inserting a copy of this AD into the limitations section of aircraft maintenance manual.

FAA AD Differences

Note: This AD differs from the MCAI and/or service information as follows: No differences.

Other FAA AD Provisions

(g) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, Standards Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to *Attn:* Taylor Martin, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; *telephone:* (816) 329-4138; *fax:* (816) 329-4090; *e-mail:* taylor.martin@faa.gov. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) *Airworthy Product:* For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective

actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements:* For any reporting requirement in this AD, under the provisions of the Paperwork Reduction Act (44 U.S.C. 3501 *et seq.*), the Office of Management and Budget (OMB) has approved the information collection requirements and has assigned OMB Control Number 2120-0056.

Related Information

(h) Refer to MCAI European Aviation Safety Agency (EASA) AD No.: 2010-0162, dated August 4, 2010; BAE Systems British Aerospace Jetstream Series 3100 & 3200 Service Bulletin 05-JA090143, dated April 30, 2009; and British Aerospace Regional Aircraft British Aerospace Jetstream Series 3100 & 3200 Service Bulletin 32-JA990142, dated March 26, 1999, for related information.

Issued in Kansas City, Missouri, on September 21, 2010.

Patrick R. Mullen,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2010-24180 Filed 9-24-10; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF THE TREASURY

Internal Revenue Service

26 CFR Parts 1, 31, 40, and 301

[REG-153340-09]

RIN 1545-BJ13

Electronic Funds Transfer of Depository Taxes; Hearing Cancellation

AGENCY: Internal Revenue Service (IRS), Treasury.

ACTION: Cancellation of notice of public hearing on proposed rulemaking.

SUMMARY: This document cancels a public hearing on proposed regulation relating to Federal tax deposits (FTDs) by Electronic Funds Transfer (EFT). The proposed regulations affect all taxpayers that currently use FTD coupons.

DATES: The public hearing, originally scheduled for September 27, 2010 at 10 a.m. is cancelled.

FOR FURTHER INFORMATION CONTACT: Regina Johnson of the Publications and Regulations Branch, Legal Processing Division, Associate Chief Counsel (Procedure and Administration) at (202) 622-7180 (not a toll-free number).

SUPPLEMENTARY INFORMATION: A notice of proposed rulemaking and notice of public hearing that appeared in the **Federal Register** on Thursday, August