

Effective Date

(a) This airworthiness directive (AD) becomes effective October 13, 2006.

Affected ADs

(b) This AD supersedes AD 2000-08-01, Amendment 39-11687.

Applicability

(c) This AD applies to Rolls-Royce Deutschland Ltd & Co KG (RRD) Tay 650-15 and Tay 651-54 turbofan engines with stage 1 high pressure turbine (HPT) discs, part number (P/N) JR32013 or P/N JR33838, and stage 1 low pressure turbine (LPT) discs, P/N JR32318A, installed. These engines are installed on, but not limited to, Fokker Model F.28 Mark 0100, and Boeing 727-100 series airplanes modified in accordance with Supplemental Type Certificate (STC) SA8472SW (727-QF).

Unsafe Condition

(d) This AD results from RRD updating their low-cycle-fatigue analysis for stage 1 HPT discs and stage 1 LPT discs and reducing their cyclic life limits. We are issuing this AD to prevent cracks leading to turbine disc failure, which could result in an uncontained engine failure and damage to the airplane.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified unless the actions have already been done.

(f) Information on the referenced Tay 650-15 engine flight plan profiles A, B, C, and D and Tay 651-54 engine datum flight profile, can be found in RRD Tay Engine Manual, Section 70-01-10.

Calculating and Re-Establishing Within 30 Days, the Achieved Cyclic Life of a Stage 1 HPT Disc or Stage 1 LPT Disc Previously Exposed to Different Flight Plan Profiles

(g) If a stage 1 HPT disc or stage 1 LPT disc was previously exposed to flight plan profile(s) different than the currently operated flight plan:

(1) You must calculate and re-establish the achieved cyclic life for that disc, within 30 days after the effective date of this AD.

(2) Use paragraphs 3.A. through 3.D.(2)(c) of Accomplishment Instructions of RRD Alert Service Bulletin (ASB) No. Tay-72-A1676, Revision 1, dated August 16, 2005, to calculate and re-establish the achieved cyclic life.

After an Engine Flight Plan Profile Changeover, Calculating and Re-Establishing Within 30 Days, the Achieved Cyclic Life of Stage 1 HPT Discs and Stage 1 LPT Discs

(h) After an engine has a flight plan profile changeover:

(1) You must calculate and re-establish the achieved cyclic life for the stage 1 HPT disc and stage 1 LPT disc, within 30 days after the flight plan changeover.

(2) Use paragraphs 3.A. through 3.D.(2)(c) of Accomplishment Instructions of RRD ASB No. Tay-72-A1676, Revision 1, dated August 16, 2005, to calculate and re-establish the achieved cyclic life.

Removal of Stage 1 HPT Discs and Stage 1 LPT Discs From Service Tay 650-15 Engine Flight Plan Profile A

(i) Remove from service Tay 650-15 stage 1 HPT discs and stage 1 LPT discs operated under flight plan profile A, before accumulating 23,000 cycles-since-new (CSN), and replace with serviceable parts.

Tay 650-15 Engine Flight Plan Profile B

(j) Remove from service Tay 650-15 stage 1 HPT discs operated under flight plan profile B and replace with serviceable parts:

(1) On or before July 31, 2007, before accumulating 21,000 CSN; and

(2) After July 31, 2007, before accumulating 20,000 CSN.

(k) Remove from service Tay 650-15 stage 1 LPT discs operated under flight plan profile B, before accumulating 21,000 CSN, and replace with serviceable parts.

Tay 650-15 Engine Flight Plan Profile C

(l) Remove from service Tay 650-15 stage 1 HPT discs operated under flight plan profile C and replace with serviceable parts:

(1) After the effective date of this AD, before accumulating 15,800 CSN; and

(2) After July 31, 2007, before accumulating 14,700 CSN.

(m) Remove from service Tay 650-15 stage 1 LPT discs operated under flight plan profile C, before accumulating 18,000 CSN, and replace with serviceable parts.

Tay 650-15 Engine Flight Plan Profile D

(n) Remove from service Tay 650-15 stage 1 HPT discs operated under flight plan profile D and replace with serviceable parts after the effective date of this AD, before accumulating 11,000 CSN.

(o) Remove from service Tay 650-15 stage 1 LPT discs operated under flight plan profile D, before accumulating 14,250 CSN, and replace with serviceable parts.

Tay 651-54 Engine Datum Flight Profile

(p) Remove from service Tay 651-54 stage 1 HPT discs operated under the engine datum flight profile, and replace with serviceable parts after the effective date of this AD, before accumulating 12,600 CSN.

(q) Remove from service Tay 651-54 stage 1 LPT discs before accumulating 20,000 CSN and replace with serviceable parts.

Alternative Methods of Compliance

(r) The Manager, Engine Certification Office, FAA, has the authority to approve alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19.

Related Information

(s) Luftfahrt-Bundesamt airworthiness directive No. D-2005-252R1, dated August 31, 2005, also addresses the subject of this AD.

Material Incorporated by Reference

(t) You must use Rolls-Royce Deutschland Ltd & Co KG Alert Service Bulletin No. Tay-72-A1676, Revision 1, dated August 16, 2005, to perform the actions required by this AD. The Director of the Federal Register approved the incorporation by reference of this service bulletin in accordance with 5

U.S.C. 552(a) and 1 CFR part 51. You can get a copy from Rolls-Royce Deutschland Ltd & Co KG, Eschenweg 11, 15872 Blankenfelde-Mahlow, Germany, telephone 49-0-33-7086-1768; fax 49-0-33-7086-3356. You may review copies at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Burlington, Massachusetts, on August 30, 2006.

Francis A. Favara,

Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. E6-14685 Filed 9-7-06; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2006-24951; Directorate Identifier 2005-NM-184-AD; Amendment 39-14752; AD 2006-18-13]

RIN 2120-AA64

Airworthiness Directives; Gulfstream Model GV and GV-SP Series Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: The FAA is adopting a new airworthiness directive (AD) for certain Gulfstream Model GV and GV-SP series airplanes. This AD requires repairing the force link assembly wire harness. This AD results from a report indicating that the wiring harness outer shield and insulation on the primary conductors may have been inadvertently cut due to an improper method used to remove the wiring outer jacket. We are issuing this AD to prevent the loss of the hardover prevention system (HOPS) in the roll axis due to a short circuit in the wiring harness, which could result in reduced controllability of the airplane.

DATES: This AD becomes effective October 13, 2006.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the AD as of October 13, 2006.

ADDRESSES: You may examine the AD docket on the Internet at <http://dms.dot.gov> or in person at the Docket Management Facility, U.S. Department of Transportation, 400 Seventh Street, SW., Nassif Building, Room PL-401, Washington, DC.

Contact Gulfstream Aerospace Corporation, Technical Publications Dept., P.O. Box 2206, Savannah, Georgia 31402–2206, for service information identified in this AD.

FOR FURTHER INFORMATION CONTACT:

Darby Mirocha, Aerospace Engineer, Systems and Equipment Branch, ACE–119A, FAA, Atlanta Aircraft Certification Office, One Crown Center, 1895 Phoenix Boulevard, suite 450, Atlanta, Georgia 30349; telephone (770) 703–6095; fax (770) 703–6097.

SUPPLEMENTARY INFORMATION:

Examining the Docket

You may examine the airworthiness directive (AD) docket on the Internet at <http://dms.dot.gov> or in person at the

Docket Management Facility office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The Docket Management Facility office (telephone (800) 647–5227) is located on the plaza level of the Nassif Building at the street address stated in the **ADDRESSES** section.

Discussion

The FAA issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to certain Gulfstream Model GV and GV–SP series airplanes. That NPRM was published in the **Federal Register** on June 6, 2006 (71 FR 32487). That NPRM proposed to require repairing the force link assembly wire harness.

Comments

We provided the public the opportunity to participate in the development of this AD. We received no comments on the NPRM or on the determination of the cost to the public.

Conclusion

We have carefully reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

Costs of Compliance

There are about 99 airplanes of the affected design in the worldwide fleet. The following table provides the estimated costs for U.S. operators to comply with this AD.

ESTIMATED COSTS

Work hours	Average labor rate per hour	Parts	Cost per airplane	Number of U.S.-registered airplanes	Fleet cost
3	\$80	The manufacturer states that it will supply required parts to the operators at no cost.	\$240	77	\$18,480

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

(1) Is not a "significant regulatory action" under Executive Order 12866;

(2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and

(3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD docket. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The Federal Aviation Administration (FAA) amends § 39.13 by adding the following new airworthiness directive (AD):

2006–18–13 Gulfstream Aerospace Corporation: Amendment 39–14752.

Docket No. FAA–2006–24951; Directorate Identifier 2005–NM–184–AD.

Effective Date

(a) This AD becomes effective October 13, 2006.

Affected ADs

(b) None.

Applicability

(c) This AD applies to the following Gulfstream Aerospace Corporation airplanes, certificated in any category:

TABLE 1.—APPLICABILITY

Model	Serial Nos.
GV series airplanes ..	674 through 693 inclusive.
GV–SP series airplanes.	5001 through 5072 inclusive.

Unsafe Condition

(d) This AD results from a report indicating that the wiring harness outer shield and insulation on the primary conductors may have been inadvertently cut due to an improper method used to remove the wiring outer jacket. We are issuing this AD to prevent the loss of the hardover prevention system (HOPS) in the roll axis due to a short circuit in the wiring harness, which could result in reduced controllability of the airplane.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Repair

(f) Within 12 months after the effective date of this AD, repair the force link assembly wire harness by doing all actions specified in the Accomplishment Instructions of the applicable service information identified in Table 2 of this AD, except as required by paragraph (g) of this AD.

TABLE 2.—SERVICE INFORMATION

For Model—	Use—
GV–SP series airplanes.	Gulfstream G500 Customer Bulletin 14, dated June 23, 2005.
GV–SP series airplanes.	Gulfstream G550 Customer Bulletin 14, dated June 23, 2005.
GV series airplanes.	Gulfstream GV Customer Bulletin 135, dated June 23, 2005.

Note 1: The Gulfstream customer bulletins identified in Table 2 of this AD include Vought Service Bulletin SB–VAIGV/GVSP–27–PG0098, dated May 9, 2005, as an additional source of service information for the repair.

Exception to Service Bulletin Specifications

(g) During the inspection of the environmental seal around the installed wires required by paragraph (f) of this AD: If any nick or other damage is found, repair before further flight using a method approved by the Manager, Atlanta Aircraft Certification Office (ACO), FAA. For a repair method to be approved by the Manager, Atlanta ACO, as required by this paragraph, the Manager's approval letter must specifically refer to this AD.

Alternative Methods of Compliance (AMOCs)

(h)(1) The Manager, Atlanta ACO, has the authority to approve AMOCs for this AD, if requested in accordance with the procedures found in 14 CFR 39.19.

(2) Before using any AMOC approved in accordance with § 39.19 on any airplane to which the AMOC applies, notify the appropriate principal inspector in the FAA Flight Standards Certificate Holding District Office.

Material Incorporated by Reference

(i) You must use the service information identified in Table 3 of this AD to perform the actions that are required by this AD, unless the AD specifies otherwise. Each Gulfstream customer bulletin listed in Table 3 of this AD includes Vought Aircraft Industries Service Bulletin SB–VAIGV/GVSP–27–PG0098, dated May 9, 2005. The Director of the Federal Register approved the incorporation by reference of these documents in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Contact Gulfstream Aerospace Corporation, Technical Publications Dept., P.O. Box 2206, Savannah, Georgia 31402–2206, for a copy of this service information. You may review copies at the Docket Management Facility, U.S.

Department of Transportation, 400 Seventh Street, SW., Room PL–401, Nassif Building, Washington, DC; on the Internet at <http://dms.dot.gov>; or at the National Archives and Records Administration (NARA). For information on the availability of this material at the NARA, call (202) 741–6030, or go to http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

TABLE 3.—MATERIAL INCORPORATED BY REFERENCE

Customer bulletin	Date
Gulfstream G500 Customer Bulletin 14.	June 23, 2005.
Gulfstream G550 Customer Bulletin 14.	June 23, 2005.
Gulfstream GV Customer Bulletin 135.	June 23, 2005.

Issued in Renton, Washington, on August 28, 2006.

Kalene C. Yanamura,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E6–14688 Filed 9–7–06; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2003–NM–114–AD; Amendment 39–14751; AD 2006–18–12]

RIN 2120–AA64

Airworthiness Directives; Saab Model SAAB–Fairchild SF340A (SAAB/SF340A) and SAAB 340B Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Saab Model SAAB–Fairchild SF340A (SAAB/SF340A) and SAAB 340B airplanes, that requires modification and repetitive inspections of the hot detection system of the tail pipe harness of the engine nacelles. The actions specified by this AD are intended to prevent false warning indications to the flightcrew from the hot detection system due to discrepancies of the harness, which could result in an unnecessary aborted takeoff on the ground or in-flight engine shutdowns. This action is intended to address the identified unsafe condition.

DATES: Effective October 13, 2006.

The incorporation by reference of a certain publication listed in the regulations is approved by the Director

of the Federal Register as of October 13, 2006.

ADDRESSES: The service information referenced in this AD may be obtained from Saab Aircraft AB, SAAB Aircraft Product Support, S–581.88, Linköping, Sweden. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington.

FOR FURTHER INFORMATION CONTACT:

Mike Borfitz, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue, SW., Renton, Washington 98057–3356; telephone (425) 227–2677; fax (425) 227–1149.

SUPPLEMENTARY INFORMATION: A

proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Saab Model SAAB–Fairchild SF340A (SAAB/SF340A) and SAAB 340B airplanes was published as a supplemental notice of proposed rulemaking (NPRM) in the **Federal Register** on June 26, 2006 (71 FR 36252). That action proposed to require modification and repetitive inspections of the hot detection system of the tail pipe harness of the engine nacelles.

Comments

We provided the public the opportunity to participate in the development of this AD. We received no comments on the supplemental NPRM or on the determination of the cost to the public.

Conclusion

We have carefully reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed in the supplemental NPRM.

Cost Impact

We estimate that 280 airplanes of U.S. registry will be affected by this AD.

It will take about 10 work hours per airplane to accomplish the modification, at an average labor rate of \$80 per work hour. Required parts cost will be between \$218 and \$2,253. Based on these figures, the cost impact of the modification on U.S. operators is estimated to be between \$285,040 and \$854,840, or between \$1,018 and \$3,053 per airplane.

It will take about 1 work hour per airplane to accomplish the inspection and application of sealant, at an average labor rate of \$80 per work hour. Based on these figures, the cost impact of this